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| Attitude Data Message test Plan/Report |

CCSDS Record

CCSDS 504.2-Y-1

Yellow Book

FOREWORD

This document records the plans for prototype testing and results of that testing for the Attitude Data Message, CCSDS 504.0-P-2.0, Pink Book. As a record of prototype testing, it is expected that expansion, deletion, or modification of this document will **not** occur. This document is subject to CCSDS document management and change control procedures, which are defined in the *Organization and Processes for the Consultative Committee for Space Data Systems*. Current versions of CCSDS documents are maintained at the CCSDS Web site:

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# Introduction

## PURPOSE

The purpose of this document is to describe the prototype testing conducted on the CCSDS Attitude Data Messages (ADM), CCSDS 504x0p-1.1 (reference [3]). An initial draft of this plan was prepared by the members of the CCSDS Navigation Working Group at the CCSDS Spring 2022 meetings conducted on-line.

## SCOPE

The scope of this document is testing of the Attitude Data Messages version 2. The ADM is part of the technical program of the CCSDS Navigation Working Group. Document 504x0p-1.1 is an update to the existing CCSDS/ISO Standard Attitude Data Messages CCSDS 504.0-B-1 (reference [2]). ADM document CCSDS 504x0p-1.1 completed a joint CCSDS Agency Review; the process is described in reference [1]. In applicable places the prototyping includes results based on modifications to the reference [3] provided via the Review Item Discrepancy (RID) process described in reference [1].

## APPLICABILITY

The ADM describes standard formats for the interagency exchange of data required for spacecraft tracking and navigation (specifically, attitude parameters and attitude ephemeris). There are three distinct message types that make up the Attitude Data Messages. These are:

* Attitude Parameter Message (APM)
* Attitude Ephemeris Message (AEM)
* Attitude Comprehensive Message (ACM)

This document applies to the prototype testing required to advance the ADM version 2 and its three constituent messages from Pink Book to Blue Book status.

In its new revision, the ADM now includes an additional message type, the Attitude Comprehensive Message or ACM. The APM and AEM are still present but have changed. Testing of the APM and AEM will be adapted to the changes made.

## RATIONALE

The CCSDS Procedures Manual states that for a Recommendation to become a Blue Book, the standard must be tested in an operational manner. The following requirements for an implementation exercise were excerpted from reference [1]:

“At least two independent and interoperable prototypes or implementations must have been developed and demonstrated in an operationally relevant environment, either real or simulated.”

This document outlines the Navigation Working Group’s approach to meeting this requirement for the ADM 504x0p-1.1.

## DOCUMENT STRUCTURE

The first sections of this document describe the Test Plan for the prototyping activity; the last sections of the document provide a Test Report of the realized plan. Acronyms are provided in Annex A.

## References

The following documents are referenced in this document. At the time of publication, the editions indicated were valid. All documents are subject to revision, and users of this document are encouraged to investigate the possibility of applying the most recent editions of the documents indicated below. The CCSDS Secretariat maintains a register of currently valid CCSDS documents.

[1] *Procedures Manual for the Consultative Committee for Space Data Systems*. CCSDS A00.0-Y-9. Yellow Book. Issue 9. Washington, D.C.: CCSDS, November 2003.

[2] *Attitude Data Messages*. Recommendation for Space Data System Standards, CCSDS 504.0-B-1. Blue Book. Issue 1. Washington, D.C.: CCSDS, September 2004.

[3] *Attitude Data Messages*. Draft Recommendation for Space Data System Standards, CCSDS 504x0p-1.1 Pink Book. Washington, D.C.: CCSDS, June 2022.

[4] *Orbit Data Messages V3 Test Plan*, CCSDS 502.1-Y-2

# Summary Conclusion/Recommendation

Note : this is the conclusion from the ADM test plan.

We hope to have the same conclusion for the ADM V2 test plan.

The test plan and test reports documented herein substantiate that the organizations participating in the CCSDS Navigation Working Group have successfully conducted prototype testing of the Attitude Parameter Message, Attitude Ephemeris Message and Attitude Comprehensive Message described in the Attitude Data Messages (ADM) 504x0p-1.1 document. During the testing, messages of the various types were produced by several different organizations, and the ability to read/process the messages was demonstrated. Based on the diversity of agencies able to read/write the messages, and the positive test results, the Navigation Working Group recommends that the revised 504x0p-1.1 document be promoted to a Blue Book CCSDS Recommended Standard.

# Attitude Data Messages (ADM) Testing Goals

The test of the ADM will exercise the following three message types that together constitute the Attitude Data Messages:

* Attitude Parameter Message (updated from version 1 (reference [2]))
* Attitude Ephemeris Message (updated from version 1 (reference [2]))
* Attitude Comprehensive Message (new in version 2 (reference [3]))

The tests described in Section 5 of this test plan will be conducted in order to meet the CCSDS requirements described in Section 1. The results of the testing are presented in Section 7.

## APM Overview

The APM is an ASCII file in “keyword=value” format. An XML format is also available. It contains a single state that must be propagated by the recipient. The file is organized into 3 sections: the Header section, Metadata section, and the Data section. The Header section contains identification information (version, creation date, originator). The Metadata section contains information regarding the object to which the state applies, applicable reference frame and time system. The Data section contains the attitude state in quaternion or Euler form, or other quantities as required (angular momentum vector, physical characteristics, spin). There is also an optional section that can contain user defined parameters.

## AEM Overview

The AEM is an ASCII file in a hybrid “keyword=value” format (header and metadata are keyword=value, data lines have a positional field layout). An XML format is also available. The AEM contains attitude states for an object at multiple points in time. The file is organized into 3 sections: the Header section, Metadata section, and the Data section. The Header section contains identification information (version, creation date, originator). The Metadata section contains information regarding the object to which the attitude applies, applicable reference frame, time system, interpolation parameters, and data start/stop times. The Data section contains the attitude state components at each epoch, plus additional derivatives or angular momentum coordinates. The recipient must interpolate to obtain arbitrary states between ephemeris points.

## ACM Overview

The ACM is an ASCII file in a hybrid “keyword=value” format (header and metadata are keyword=value, data lines have a positional field layout). An XML format is also available. The ACM is comprehensive, in allowing users to exchange attitude, uncertainty data, maneuver, and many others. The file is organized into six distinct sections, as shown in the table below (excerpted from Table 5-1 of the draft standard Reference [3]): the Header section, Metadata section, and the Data section with its six subsections as listed.

The Header section contains identification information (version, creation date, originator). The Metadata section contains information regarding the object to which the attitude applies, applicable reference frame, time system, interpolation parameters, and data start/stop times. The Data Section contains attitude and covariance time histories, as well as detailed specificity of physical properties, maneuver data, attitude determination data, and user-defined parameters.

Table ‑ ACM Layout and Ordering Specification

|  |  |  |  |  |
| --- | --- | --- | --- | --- |
| **Section** | | | **Content** | **Status**  **M/O** |
| Header | | | A single header of the message | M |
| Metadata | | | A single Metadata section (data about data) | M |
| Data | attitude data #1 | data description | One or more attitude state time histories (each consisting of one or more attitude states) | O |
| data lines |
| **⁝** | |
| attitude data #n | data description |
| data lines |
| physical properties |  | A single space object  physical characteristics section | O |
| covariance data #1 | data description | One or more covariance time histories (each consisting of one or more covariance matrix diagonals) | O |
| data lines |
| **⁝** | |
| covariance data #n | data description |
| data lines |
| maneuver data #1 |  | One or more maneuver specification sections | O |
| **⁝** | |
| maneuver data #n |  |
| attitude determination data |  | A single attitude determination Data section | O |
| user-defined data |  | A single user-defined Data section | O |

# Test Plan Overview

## Testing philosophy

The prototyping of the ADM is designed to be consistent with the changes made between ADM versions 1 and 2.

The tests described in the next sections are data oriented (not format oriented). The messages that are exchanged can be either XML or KVN messages.

Classically, all prototypes of the APM, AEM, ACM use KVN syntax. The messages are also converted to XML to check the consistency between XML and KVN formats.

## APM

The APM version 1 has already been proved to be useable in an operational context. By analogy, the APM – version 2 is also useable in an operational context if the data present in the message are the same, or if it can be shown that the data in a version 1 message can be converted into data in a version 2 message.

Here is a close examination of the changes between version 1 and 2, and the subsequent need for testing.

Note that explicit subsection START/STOP keywords were added for each subsection to more clearly delineate the logical block boundaries.

| **Differences between versions 1 and 2** | **Specific test needed ?  (and justification)** |
| --- | --- |
| **Header**  MESSAGE\_ID added (optional) | No  Additional information (MESSAGE\_ID) is present, but it is not essential for attitude processing (it is effectively a comment). |
| **MetaData**  No change | No |
| **Quaternion Data**  Direction: in version 2, the direction is implicitly from A to B. | No: the data are the same.  Frame A and B have to be swapped in version 2 if the direction is B2A in version 1. |
| **Euler Data**  Direction: in version 2, the direction is implicitly from A to B.  Rotation sequence: format change  Angle names: format change  Rates are not provided in Euler blocks in version 2. | No: the data are the same.  The data from a version 1 message should be sent as 2 blocks in version 2 if rates are present. Rates should now be provided in a specific “ANGVEL” bloc  Rotation sequence 123 is written XYZ in version 2.  X\_ANGLE is now written ANGLE\_1.  (similar changes for Y and Z components) |
| **Spin Data**  In version 2, the direction is implicitly from A to B.  The version 1 spin block is still present in version 2.  An alternative form has been added. | Test of the additional features needed. |
| **Angular Velocity**  This block has been added. | Test of added keywords needed. |
| **Inertia Block** | No  (no change) |
| **Maneuver block**  Delta\_mass added. | No as delta mass was not needed in version 1. |

The following tests are then proposed:

|  |  |  |  |
| --- | --- | --- | --- |
| **Test #** | **Purpose** | **Agencies, Direction** | **Msg Type** |
| APM#1 | Spin Data test  Test of the new features | X => Y => X | APM |
| APM#2 | Angular momentum test  Test of the new features | X => Y => X | APM |
| APM#3 | Full message, will include test of added angular velocity block.  (in particular for KVN-XML format consistency check) | X => Y => X | APM |

Note: all tests include KVN <-> XML conversion.

## AEM

The AEM version 1 has already been proved to be useable in an operational context. By analogy, the AEM version 2 is also useable in an operational context if the data present in the message are the same, or if it can be shown that the data in a version 1 message can be converted into data in a version 2 message.

Here is a close examination of the changes between version 1 and 2, and the subsequent need for testing.

| **Differences between versions 1 and 2** | **Specific test needed ?  (and justification)** |
| --- | --- |
| **Header**  MESSAGE\_ID added (optional) | No  Additional information (MESSAGE\_ID) is present, but it is not essential for attitude processing (it is effectively a comment). |
| **MetaData**  QUATERNION\_TYPE: removed in version 2 as the order is imposed (Q1, Q2, Q3, QC ⬄ “LAST”).  EULER\_ROT\_SEQ: format change in version 2 (as in APM)  ATTITUDE\_DIR removed in version 2: the direction is A2B implicitly in version 2 (as in APM)  RATE\_FRAME: can be any frame in version 2, and not just REF\_FRAME\_A or REF\_FRAME\_B) | No: the information is the same  QUATERNION\_TYPE: elements should be reordered if QC was “FIRST” in version 1.  EULER\_ROT\_SEQ: value should be changed. Example “XYZ” instead of “123”.  ATTITUDE\_DIR: swap A and B if the value is “B2A” in version 1.  RATE\_FRAME/ANGVEL\_FRAME: possibilities in version 2 include those allowed in version 1 |
| **Data lines**  QUATERNION/RATE: now called QUATERNION/ANGVEL (same information)  EULER\_ANGLE/RATE: now called EULER\_ANGLE/ANGVEL (same information)  EULER\_ANGLE/DERIVATIVE: new in version 2  SPIN/NUTATION\_MOM: new in version 2 | “EULER\_ANGLE/DERIVATIVE” is a new type of line.  “SPIN/NUTATION\_MOM” is a new type of line. |

The following tests are then proposed:

|  |  |  |  |
| --- | --- | --- | --- |
| **Test #** | **Purpose** | **Agencies, Direction** | **Msg Type** |
| AEM#1 | Euler angles / derivative test | X => Y => X | AEM |
| AEM#2 | Spin / nutation + angular momentum | X => Y => X | AEM |
| AEM#3 | Complex message  (containing at least 2 blocks) | X => Y => X | AEM |

Note: all tests include KVN <-> XML conversion.

## ACM

The ACM contains 6 different data blocks that are: Attitude State Time History, Space Object Physical Characteristics, Covariance Time History, Maneuver Specification, Attitude Determination Data, and User-Defined Parameters.

The testing strategy is inspired by what is done for the ODM V3 (OCM), see Reference [4]:

- 1 round-trip test for each data block (only one block in the message).

- 1 test with a full message.

Some of the tests will implement specific features:

- data lines with relative or absolute times

- content with values taken from SANA registry

The objectives of all these tests are to check that :

- the standard contains accurate descriptive information of the data contained in the message, and that the messages that are exchanged are compliant with the description,

- the data present in the message can be used effectively in a (real or simulated) operational context, and that, in particular, no important information is missing.

Here is a summary of the tests that are planned:

| **Test #** | **Purpose** | **Agencies, Direction** | **Msg Type** |
| --- | --- | --- | --- |
| ACM#1 | Attitude states history | X => Y => X | ACM |
| ACM#2 | Physical data | X => Y => X | ACM |
| ACM#3 | Covariance data | X => Y => X | ACM |
| ACM#4 | Maneuver | X => Y => X | ACM |
| ACM#5 | Attitude determination | X => Y => X | ACM |
| ACM#6 | User defined data | X => Y => X | ACM |
| ACM#7 | Full message | X => Y => X | ACM |

Note: All tests include KVN <-> XML conversion.

# Test Plan Details

## TEST CASE APM#1: SPIN DATA Unit Test

### Test description

In this test, X will create an APM message for a spinning object and send it to Y. The message will contain the appropriate header information, as well as reference frame information, and spin data (containing angular momentum vector coordinates). The spin data are the following:

|  |  |
| --- | --- |
| Epoch | 1 jan 2023 0h TAI |
| SPIN\_ALPHA | 10 deg |
| SPIN\_DELTA | 30 deg |
| SPIN\_ANGLE | 0 deg |
| SPIN\_ANGLE\_VEL | 1 deg/s |
| MOMENTUM\_ALPHA | 80 deg |
| MOMENTUM\_DELTA | 10 deg |
| NUTATION\_VEL | 0.5 deg/s |

Y will generate a quaternion at a different epoch (1 jan 2023 12h TAI) using the method described in the standard. Y will send an APM containing the quaternion to X and will also report to X about potential issues. X then will analyze the results obtained by Y by comparing with their own results.

### Expected results

It is anticipated that X and Y will successfully write and read the message and that the round-trip consistency test will yield a match. It is expected that the quaternions computed by X and Y will be the same (to some negligible accuracy). Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE APM#2: Angular momentum Unit Test

### Test description

In this test, X will create an APM message and send it to Y. The message will contain the appropriate header information, as well as quaternion and angular momentum data. The quaternion represents the attitude of the body frame in EME2000. The angular momentum data give the coordinates of the angular momentum vector in the body frame.

The data are the following:

|  |  |
| --- | --- |
| Epoch | 1 jan 2023 0h TAI |
| *Quaternion:* | |
| REF\_FRAME\_A | EME2000 |
| REF\_FRAME\_B | SC\_BODY |
| QC | 0.925417 |
| Q1 | 0.171010 |
| Q2 | -0.030154 |
| Q3 | 0.336824 |
| *Angular momentum:* | |
| ANGVEL\_FRAME | SC\_BODY |
| ANGVEL\_X | 0.0001 deg/sec |
| ANGVEL\_Y | 0.05 deg/sec |
| ANGVEL\_Z | 0.00003 deg/sec |

Y will ingest the message, compute the angular momentum vector in EME2000, and transmit a message containing the computed coordinates to X for comparison and verification.

### EXPECTED RESULTS

It is anticipated that X and Y will successfully write and read the messages and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE APM#3: Full message

### Test description

In this test, X will create an APM message that contains all data blocks and send it to Y. The message will contain the appropriate header and metadata information. In particular, the message will contain the following Euler data describing the orientation of the body frame in EME2000 and the angular velocity in the spacecraft body frame:

|  |  |
| --- | --- |
| Epoch | 1 jan 2023 0h TAI |
| EULER\_ROT\_SEQ | ZXZ |
| ANGLE\_1 | 10 deg |
| ANGLE\_2 | 20 deg |
| ANGLE\_3 | -10 deg |

Y will generate a quaternion describing the orientation of the body frame in EME2000 using the data above and send it to X. Y will also report about potential issues. X will compare their results with Y and report back to Y.

### Expected results

It is anticipated that X and Y will successfully write and read the merged data blocks and that the round-trip consistency test will yield a match. It is expected that the quaternions computed by X and Y will be the same, as will all logical blocks exchanged. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE AEM#1: Euler angle + derivatives unit test

### Test description

In this test, X will create an AEM message containing Euler angles and associated derivatives. The message will contain the appropriate header information, as well as reference frame information and attitude data (Euler angles and derivatives). The data will be the following:

|  |  |
| --- | --- |
| Initial epoch | 1 jan 2023 0h TAI |
| EULER\_ROT\_SEQ | ZXZ |
| ANGLE\_1 | 10 deg |
| ANGLE\_2 | 20 deg |
| ANGLE\_3 | -10 deg |
| ANGLE\_1\_DOT | 1 deg/s |
| ANGLE\_2\_DOT | 1 deg/s |
| ANGLE\_3\_DOT | -1 deg/s |

The AEM will contain the data for 10 different epochs every 30 seconds (TAI).

Y will ingest the message and will generate an AEM containing quaternion and angular velocity data (with QUATERNION/ANGVEL data line type). Y will send it to X and will also report about potential issues. X will compare results with Y and report back to Y.

5.3.2 EXPECTED RESULTS

It is anticipated that the quaternion and angular velocity ephemeris determined by Y and sent to X will be consistent with the reference data computed by X using the same inputs. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test

## TEST CASE AEM#2: Spin / nutation + angular momentum unit test

### Test description

In this test, X will create an AEM message for a spinning object and send it to Y. The message will contain the appropriate header and metadata information, as well as reference frame information, and spin data (containing angular momentum vector coordinates). The spin data at initial epoch are the following:

|  |  |
| --- | --- |
| Epoch | 1 jan 2023 0h TAI |
| SPIN\_ALPHA | 10 deg |
| SPIN\_DELTA | 30 deg |
| SPIN\_ANGLE | 0 deg |
| SPIN\_ANGLE\_VEL | 1 deg/s |
| MOMENTUM\_ALPHA | 80 deg |
| MOMENTUM\_DELTA | 10 deg |
| NUTATION\_VEL | 0.5 deg/s |

The AEM will contain the data for 10 different epochs every 30 seconds (TAI). At each epoch the spin\_angle will be incremented by 1 deg.

Y will ingest the message, and transmit the message back to X for comparison and verification.

### Expected results

It is anticipated that X and Y will successfully write and read the spacecraft attitude data and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE AEM#3: complex message

### Test description

In this test, X will create an AEM message containing all data blocks. The message will contain the appropriate header and metadata information, as well as reference frame information, and the appropriate data. Y will ingest the message. The message will in particular contain the following data describing the attitude of the body frame in EME2000:

|  |  |  |  |  |
| --- | --- | --- | --- | --- |
| Time offset (seconds) | QC | Q1 | Q2 | Q3 |
| 0 | 0.925417 | 0.171010 | -0.030154 | 0.336824 |
| 30 | 0.851651 | 0.397131 | 0.144544 | 0.309976 |
| 60 | 0.719846 | 0.413176 | 0.492404 | 0.262003 |
| 90 | 0.538986 | 0.142244 | 0.806707 | 0.196175 |
| 120 | 0.321394 | -0.321394 | 0.883022 | 0.116978 |
| 150 | 0.081900 | -0.763129 | 0.640342 | 0.029809 |
| 180 | -0.163176 | -0.969846 | 0.171010 | -0.059391 |
| 210 | -0.397131 | -0.851651 | -0.309976 | -0.144544 |
| 240 | -0.604023 | -0.492404 | -0.586824 | -0.219846 |
| 270 | -0.769751 | -0.099601 | -0.564863 | -0.280166 |
| 300 | -0.883022 | 0.116978 | -0.321394 | -0.321394 |

Using the data at the initial epoch, Y will convert the coordinates of some vector given in the inertial frame into coordinates in the body frame. The coordinates in the inertial frame are [1; 1; 1]. Y will send the results to X and will also report to X about potential issues.

X will compare their results with Y and report back to Y.

### Expected results

It is anticipated that X and Y will successfully write and read the spacecraft attitude data and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#1: ACM attitude DATA UNIT TEST

### Test description

For this test, X will send an ACM describing a sequence of spacecraft attitude states as a function of relative time to Y. The attitude states will consist of a DCM (matrix from the inertial frame to the body frame). X will send the message to Y, Y will compute a quaternion from the DCM and send it back to X (the format does not matter). Y will also report about potential issues. X will compare results with Y and report back to Y.

Epoch: 2023-01-01T00:00:00 UTC

Time offset: 0 sec

DCM = [ -0.3333 0.9020 -0.2745

-0.6667 -0.0196 0.7451

0.6667 0.4314 0.6078]

Time offset: 1 sec

DCM = [-0.3347 0.9017 -0.2736

-0.6655 -0.0206 0.7461

0.6672 0.4318 0.6070]

### Expected results

It is anticipated that X and Y will successfully write and read the attitude state time history and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#2: ACM PHYSICAL CHARACTERISTICS UNIT TEST

### TEST DESCRIPTION

For this test, X will send an ACM describing space object physical characteristics to Y. In this simple case, an inertia matrix and wet mass will be shared. Y will ingest the message, and transmit the message back to Y for comparison and verification.

Inertia Matrix in the Spacecraft body frame:

[770.8 -74.5 -80.2

-74.5 1277.4 -25.4

-80.2 -25.4 1497.5] kg.m2

Wet mass: 1900 kg

### EXPECTED RESULTS

It is anticipated that X and Y will successfully write and read the space object physical characteristics and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#3: Covariance data UNIT TEST

### TEST DESCRIPTION

For this test, X will send an ACM describing a sequence of spacecraft attitude covariances as a function of absolute time to Y. In this simple case, two covariance blocks will be included: (1) covariance containing quaternion errors; and (2) covariance containing angle and gyro bias errors. Y will ingest the message, and transmit the message back to X for comparison and verification.

1)

Epoch = 2022-09-27T00:00:00 UTC

Time = 2022-09-27T00:09:00

Covariance = 2.70e-10 1.48e-10 5.18e-12 1.80e-10

Time = 2022-09-27T00:10:00

Covariance = 2.72e-10 1.51e-10 5.19e-12 1.83e-10

2)

Epoch = 2022-09-27T00:00:00 UTC

Time = 2022-09-27T00:09:00

Covariance = 8.50-e-9 deg2 1.99e-8 deg2 5.34e-8 deg2 1.66e-13 (deg/sec)2 2.88e-13 (deg/sec)2 5.36e-13 (deg/sec)2

Time = 2022-09-27T00:10:00

Covariance = 8.11e-9 deg2 1.86e-8 deg2 4.86e-8 deg2 1.41e-13(deg/sec)2 2.76e-13(deg/sec)2 4.96e-13(deg/sec)2

Note that units for Covariance are specified in Annex B of 504x0p-1.1.

### EXPECTED RESULTS

It is anticipated that X and Y will successfully write and read the sequence of spacecraft attitude and gyro bias covariances as a function of absolute time and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#4: Maneuver data UNIT TEST

### TEST DESCRIPTION

For this test, X will send an ACM describing space object attitude maneuvers to Y. In this simple case, a “target momentum” maneuver will be shared. Y will ingest the message, and transmit the message back to X for comparison and verification.

Time = 2022-11-02T00:00:00 UTC

Target momentum = [0 10 0] Nms

### EXPECTED RESULTS

It is anticipated that X and Y will successfully write and read the space object maneuvers and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#5: Attitude determination data UNIT TEST

### TEST DESCRIPTION

For this test, X will send an ACM describing attitude determination characteristics for the LRO spacecraft’s onboard Extended Kalman Filter (EKF) to Y. Y will ingest the message, and transmit the message back to X for comparison and verification.

|  |  |
| --- | --- |
| AD\_METHOD | EKF |
| ATTITUDE\_SOURCE | OBC |
| NUMBER\_STATES | 7 |
| ATTITUDE\_STATES | QUATERNION |
| REF\_FRAME\_A | J2000 |
| REF\_FRAME\_B | SC\_BODY\_1 |
| RATE\_STATES | GYRO\_BIAS |
| SIGMA\_U | 1.9e-8 deg/s\*\*1.5 |
| SIGMA\_V | 8.3e-5 deg/s\*\*0.5 |
| NUMBER\_SENSORS\_USED | 2 |
| SENSORS\_USED\_1 | AST |
| NUMBER\_SENSOR\_NOISE\_COVARIANCE\_1 | 3 |
| SENSOR\_NOISE\_STDDEV\_1 | 0.003 deg 0.003 deg 0.010 deg |
| SENSORS\_USED\_2 | AST |
| NUMBER\_SENSOR\_NOISE\_COVARIANCE\_1 | 3 |
| SENSOR\_NOISE\_STDDEV\_1 | 0.003 deg 0.003 deg 0.010 deg |

### EXPECTED RESULTS

It is anticipated that X and Y will successfully write and read the attitude determination data and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#6: User-DEFINED parameters UNIT TEST

### TEST DESCRIPTION

For this test, X will send an ACM describing user-defined parameters to Y. In this simple case, additional satellite characteristics will be shared (thruster direction in body frame for 2 thrusters). Y will ingest the message, and transmit the message back to X for comparison and verification.

Thruster 1 direction = [-0.97 0.003 -0.2431]

Thruster 2 direction = [-0.96 -0.001 -0.28]

### EXPECTED RESULT

It is anticipated that X and Y will successfully write and read the user-defined parameters and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## TEST CASE ACM#7: Full message

### TEST DESCRIPTION

For this test, X will send an ACM containing the merged data blocks of all of the previous unit test content to Y. Y will ingest the message, and transmit the message back to X for comparison and verification.

### EXPECTED RESULTS

It is anticipated that X and Y will successfully write and read the merged data blocks and that the round-trip consistency test will yield a match. Assuming that these criteria are met, the test will be considered successful. In the event of discrepancies, troubleshooting will be conducted by the participants in the test.

## Message formats

The messages are typically exchanged in KVN format. All messages will be separately converted from KVN to XML for validation against the schemas.

Note:   
- The messages that are exchanged can also be in XML format.

# Test Report Overview

Engineers at X and Y will prepare test data sheets as applicable, and send them to the Navigation Working Group via email.

The Test Report Details will be consolidated in Section 7 of this document. A summarization of the test process and the recommendation of the Navigation Working Group may be found in Section 2 of the report. The report will be submitted to the CCSDS Engineering Steering Group (CESG) and CCSDS Management Council (CMC), along with results of the Agency Reviews. At that time, a formal request will be submitted to the CMC for progression of the ADM to CCSDS Blue Book status.

The next page contains a format for the test data sheets that will be used to report the results of individual tests. The form includes sections for the producer of the message and the consumer of the message (producing agency, producing test engineer, consuming agency, and consuming test engineer).

**SAMPLE**



**Attitude Data Messages P-1.1 Prototype Test Data Sheet**

|  |  |  |
| --- | --- | --- |
| 1 | Report Date: |  |
| 2 | Program Under Test: | Attitude Data Messages P1.1 (ADM) Prototype |
| 3 | Test Case Number: |  |
| 4 | Agencies Participating in this Test Case: |  |
| 5 | Agency Responsible for Producing Test Message |  |
| 6 | Producing Test Engineer: |  |
| 7 | Agency Responsible for Consuming Test Message |  |
| 8 | Consuming Test Engineer: |  |
| 9 | Spacecraft: |  |
| 10 | Results (Pass, Partial Pass, Fail): |  |
| 11 | Variances from Expected Result: |  |
| 12 | Comments: |  |

# Test Report Details

## Test Case APM#1

## Test Case APM#2

## Test Case APM#3

## Test Case AEM#1

## Test Case AEM#2

## Test Case AEM#3

## Test Case ACM#1

## Test Case ACM#2

## Test Case ACM#3

## Test Case ACM#4

## Test Case ACM#5

## Test Case ACM#6

## Test Case ACM#7

1. ABBREVIATIONS AND ACRONYMS

**(INFORMATIVE)**

ASCII American Standard Code for Information Interchange

CCSDS Consultative Committee for Space Data Systems

CNES Centre National d’Etudes Spatiales

DLR/GSOC Deutsches Zentrum für Luft und Raumfahrt (German Aerospace Center)/German Space Operations Center

ESA/ESOC European Space Agency/European Space Operations Center

ESA/ESAC European Space Agency/European Space Astronomy Center

JAXA Japan Aerospace Exploration Agency

KVN Keyword = Value Notation

NASA/GSFC National Aeronautics and Space Administration/Goddard Space Flight Center

NASA/JPL National Aeronautics and Space Administration/Jet Propulsion Laboratory

NASA/JSC National Aeronautics and Space Administration/Johnson Space Flight Center

NavWG CCSDS Navigation Working Group

OCM Orbit Comprehensive Message

SANA Space Assigned Numbers Authority

XML Extensible Markup Language

XSLT Extensible Stylesheet Language Transformations