

Recommendation for Space Data System Standards

**ORBIT DATA
MESSAGES**

PROPOSED STANDARD

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FOREWORD

This document is a Recommended Standard for Orbit Data Messages (ODMs) and has been prepared by the Consultative Committee for Space Data Systems (CCSDS). The set of orbit data messages described in this Recommended Standard is the baseline concept for trajectory representation in data interchange applications that are cross-supported between Agencies of the CCSDS.

This Recommended Standard establishes a common framework and provides a common basis for the interchange of orbit and orbit-relevant data. It allows implementing organizations within each Agency to proceed coherently with the development of compatible derived standards for the flight and ground systems that are within their cognizance. Derived Agency standards may implement only a subset of the optional features allowed by the Recommended Standard and may incorporate features not addressed by this Recommended Standard.

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- Space and Upper Atmosphere Research Commission (SUPARCO)/Pakistan.
- Swedish Space Corporation (SSC)/Sweden.
- United States Geological Survey (USGS)/USA.

DOCUMENT CONTROL

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CCSDS 502.0.R-2, Cor. 1	Technical Corrigendum 1		Corrects/clarifies text; updates references [2] and [5] to current issues in 0.
CCSDS 502.0-P2.41	Orbit Comprehensive Message SANA Registry for ODM Keyword values		Added Orbit Comprehensive Message (OCM) and transitioned ODM keyword values to SANA registry

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1 INTRODUCTION

1.1 PURPOSE AND SCOPE

This Orbit Data Messages (ODM) Recommended Standard specifies four standard message formats for use in transferring spacecraft orbit information between space agencies and commercial or governmental spacecraft operators: the Orbit Parameter Message (OPM), the Orbit Mean-Elements Message (OMM), the Orbit Ephemeris Message (OEM), and the Orbit Comprehensive Message (OCM). Such exchanges are used for:

- a) pre-flight planning for tracking or navigation support;
- b) scheduling tracking support;
- c) carrying out tracking operations (sometimes called metric predicts);
- d) performing orbit comparisons;
- e) carrying out navigation operations such as orbit propagation and orbit reconstruction;
- f) assessing mutual physical and electromagnetic interference among satellites orbiting the same celestial body (primarily Earth, Moon, and Mars at present);
- g) performing orbit conjunction (collision avoidance) studies; and
- h) developing and executing collaborative maneuvers to mitigate interference or enhance mutual operations.

This Recommended Standard includes sets of requirements and criteria that the message formats have been designed to meet. For exchanges where these requirements do not capture the needs of the participating agencies and satellite operators, another mechanism may be selected.

The Orbit Data Messages (ODM) standard is an international standard published under the auspices of CCSDS and International Standards Organization (ISO) Technical Committee 20, Subcommittee 13, developed jointly and in concert with the ISO TC20/SC14. As such, this CCSDS standard is also properly labeled as ISO 26900.

The recommended Orbit Data Message format is ASCII (reference [4]).

This ODM document describes both ‘Keyword = Value Notation’ (or KVN) as well as Extensible Markup Language (XML, reference [5]) formatted messages. Selection of KVN or XML format should be mutually agreed between message exchange partners.

NOTE – As currently specified, an OPM, OMM, or OEM file is to represent orbit data for a single spacecraft and the OCM is to represent orbit data for either a single spacecraft or single parent spacecraft of a parent/child spacecraft deployment scenario. It is possible that the architecture may support multiple spacecraft per file; this could be considered in the future.

1.2 APPLICABILITY

The Orbit Data Message family of standardized orbit messages is applicable to all phases of the spacecraft and launch vehicle life cycle. The rationale behind the design of each particular orbit data message is described in ANNEX J and may help the application engineer to select a suitable message. Definition of the orbit accuracy underlying a particular orbit message is outside of the scope of this Recommended Standard and should be mutually agreed between message exchange partners participants (or specified via COMMENT sections in the message itself). Applicability information specific to each orbit data message format appears in Sections 3, 4, 5, and 6 as well as in J2.4.

This Recommended Standard is applicable only to the message format and content, but not to its transmission. The transmission of the message between agencies and operators is outside the scope of this document and should be mutually agreed between message exchange partners.

Description of the message formats based on the use of Extensible Markup Language (XML) is detailed in Section 8.

1.3 RATIONALE

This update to version 2 of the Orbit Data Messages adds a fourth message type (OCM) based on collaboration of the CCSDS Navigation Working Group and the ISO Technical Committee 20, Subcommittee 14, Working Group 3 (ISO TC20/SC14/WG3). A full list of the changes in this document is located in ANNEX L.

1.4 DOCUMENT STRUCTURE

Section 2 provides a brief overview of the CCSDS-recommended Orbit Data Message types, the Orbit Parameter Message (OPM), Orbit Mean-Elements Message (OMM), Orbit Ephemeris Message (OEM), and the Orbit Comprehensive Message (OCM).

Section 3 provides details about the structure and content of the OPM.

Section 4 provides details about the structure and content of the OMM.

Section 5 provides details about the structure and content of the OEM.

Section 6 provides details about the structure and content of the OCM.

Section 7 discusses the syntax considerations of the set of Orbit Data Messages (OPM, OMM, OEM, and OCM).

Section 8 provides details on the XML instantiations of the OPM, OMM, OEM, and OCM.

Following the principal content of the document, there are a number of annexes, both normative and informative, to guide the ODM user.

1.5 CONVENTIONS AND DEFINITIONS

1.5.1 NOTATION

1.5.1.1 Unit Notations

The following conventions for unit notations apply throughout this Recommended Standard, with message-specific guidance provided in Section 7-5. Units are drawn from the International System of Units (SI); units are either SI base units, SI derived units, or units outside the SI that are accepted for use with the SI [1]. Except as noted, the units used within this document are as follows:

- d: days, 86400 SI seconds;
- kg: kilograms;
- km: kilometers;
- m: meters;
- n/a: (units are not applicable)
- %: percent.
- s: SI seconds;
- SFU: Solar Flux Units, equivalent to $10^{-22} \text{ W}/(\text{m}^2 \cdot \text{Hz})$
- W: watts;

1.5.1.2 General

The following notational conventions are used in this document:

- a) multiplication of units is denoted with a single asterisk '*' (e.g., 'kg*s');
- b) exponents of units are denoted with a double asterisk '**' (e.g., $\text{m}^2 = \text{m}^{**2}$);
- c) square roots of units are denoted by the same exponent notation of a double asterisk '**' (e.g., $\sqrt{\text{km}} = \text{km}^{**0.5}$);
- d) division of units is denoted with a single forward slash '/' (e.g., m/s).

1.5.2 NOMENCLATURE

The following conventions apply for the normative specifications in this Manual:

- a) the words 'shall' and 'must' imply a binding and verifiable specification;
- b) the word 'should' implies an optional, but desirable, specification;

- c) the word ‘may’ implies an optional specification;
- d) the words ‘is’, ‘are’, and ‘will’ imply statements of fact.

NOTE – These conventions do not imply constraints on diction in text that is clearly informative in nature.

1.5.3 DEFINITIONS

For the purposes of this document, the following definitions apply:

- a) the word ‘agencies’ may also be construed as meaning ‘satellite operators’ or ‘satellite service providers’;
- b) the word ‘participant’ denotes an entity that has the ability to acquire or broadcast navigation messages and/or radio frequencies, for example, a spacecraft, a tracking station, a tracking instrument, or an agency/operator;
- c) the notation ‘n/a’ signifies ‘not applicable’;
- d) depending on context, the term ‘ODM’ may be used to refer to this document, or may be used to refer collectively to the OPM, OMM, OEM, and OCM messages.
- e) An 'observation' is a unique measurement set of a satellite’s state from a single sensor configuration at a single time (e.g. azimuth from a single sensor at a single time).
- f) A 'sensor track' is a set of observations within a specified number of minutes for the same object, observed by the same sensor configuration, where each observation is within a specified number of minutes (which is dependent on the orbit regime of the object) of the other observations in the track (e.g. a set of 10 two-way transponder range measurements from the same sensor using the same transponder on the satellite), where the number of minutes could alternately be defined as the time between start and stop of the measurement “session” or signal modulation that enables metric tracking.

1.6 REFERENCES

The following documents contain provisions which, through reference in this text, constitute provisions of this Recommended Standard. At the time of publication, the editions indicated were valid. All documents are subject to revision, and users of this Recommended Standard are encouraged to investigate the possibility of applying the most recent editions of the documents indicated below. The CCSDS Secretariat maintains a register of currently valid CCSDS Recommended Standards.

- [1] *The International System of Units (SI)*. 8th ed. Sèvres, France: BIPM, 2006.
- [2] *Time Code Formats*. Recommendation for Space Data System Standards, CCSDS 301.0-B-4. Blue Book. Issue 4. Washington, D.C.: CCSDS, November 2010.

- [3] *United Nations Office of Outer Space Affairs satellite designator/index, searchable at* <<http://www.unoosa.org/oosa/osoindex> >
- [4] *Information Technology—8-Bit Single-Byte Coded Graphic Character Sets—Part 1: Latin Alphabet No. 1.* International Standard, ISO/IEC 8859-1:1998. Geneva: ISO, 1998.
- [5] *XML Specification for Navigation Data Messages.* Recommendation for Space Data System Standards, CCSDS 505.0-B-1. Blue Book. Issue 1. Washington, D.C.: CCSDS, December 2010.
- [6] Paul V. Biron and Ashok Malhotra, eds. *XML Schema Part 2: Datatypes.* 2nd Edition. W3C Recommendation. N.p.: W3C, October 2004. <<http://www.w3.org/TR/2001/REC-xmlschema-2-20010502/>>
- [7] *IEEE Standard for Binary Floating-Point Arithmetic.* IEEE Std 754-1985. New York: IEEE, 1985.
- [8] Henry S. Thompson, et al. eds. *XML Schema Part 1: Structures.* 2nd ed. W3C Recommendation. N.p.: W3C, October 2004.
- [9] CCSDS 503.0-B-1, Tracking Data Message, November 2007.
- [10] CCSDS 504.0-B-1, Attitude Data Message, May 2008.
- [11] SANA Navigation Working Group Registry:
https://sanaregistry.org/r/navigation_standard_normative_annexes

2 OVERVIEW

2.1 ORBIT PARAMETER MESSAGE (OPM)

An OPM specifies the position and velocity of a single object at a specified epoch. Optionally, osculating Keplerian elements may be provided. Note that a sequence of OPMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H. This message is suited to exchanges that (1) involve automated interaction and/or human interaction, and (2) do not require high-fidelity dynamic modeling.

The OPM requires the use of a propagation technique to determine the position and velocity at times different from the specified epoch, leading to a higher level of effort for software implementation than for the OEM.

The OPM also contains an optional 6x6 position/velocity covariance matrix which reflects the uncertainty of the orbit state and may be used in the propagation process to estimate future uncertainties.

The OPM allows for modeling of any number of maneuvers (as both finite and instantaneous events) and simple modeling of solar radiation pressure and atmospheric drag.

Though primarily intended for use by computers, the attributes of the OPM also make it suitable for applications such as exchanges by email, FAX or voice, or applications where the message is to be frequently interpreted by humans.

2.2 ORBIT MEAN-ELEMENTS MESSAGE (OMM)

An OMM specifies the orbital characteristics of a single object at a specified epoch, expressed in mean Keplerian elements. Note that a sequence of OMMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H. The OMM is suited to exchanges that (1) involve automated interaction and/or human interaction, and (2) do not require high-fidelity dynamic modeling. Such exchanges may be inter-agency exchanges, or ad hoc exchanges among satellite operators when interface control documents have not been negotiated. Ad hoc interactions usually involve more than one satellite, each satellite controlled and operated by a different operating authority.

The OMM includes keywords and values that can be used to generate canonical NORAD Two Line Element Sets (TLEs) to accommodate the needs of heritage users (see ANNEX N, Reference [N-3]).

The OMM also contains an optional covariance matrix which reflects the uncertainty of the mean Keplerian elements. This information may be used to determine contact parameters that encompass uncertainties in predicted future states of orbiting objects of interest.

This message is suited for directing antennas and planning contacts with satellites. It is not recommended for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, or propagating precisely the orbits of active satellites, inactive man-made objects, and near-Earth debris fragments. It is not suitable for numerical integration of the governing equations.

Though primarily intended for use by computers, the attributes of the OMM also make it suitable for applications such as exchanges by email, FAX, or voice, or applications where the message is to be frequently interpreted by humans.

2.3 ORBIT EPHEMERIS MESSAGE (OEM)

An OEM specifies the position and velocity of a single object at multiple epochs contained within a specified time range. Note that a sequence of OEMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H. The OEM is suited to exchanges that (1) involve automated interaction (e.g., computer-to-computer communication where frequent, fast automated time interpretation and processing is required), and (2) require higher fidelity or higher precision dynamic modeling than is possible with the OPM.

The OEM allows for dynamic modeling of any number of gravitational and non-gravitational accelerations. The OEM requires the use of an interpolation technique to interpret the position and velocity at times different from the tabular epochs.

The OEM also contains an optional covariance matrix which reflects the uncertainty of the orbit solution used to generate states in the ephemeris.

2.4 ORBIT COMPREHENSIVE MESSAGE (OCM)

An OCM specifies position and velocity of either a single object or an en masse parent/child deployment scenario stemming from a single object. Note that a sequence of OCMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H.

The OCM aggregates and extends OPM, OEM, and OMM content in a single comprehensive hybrid message (file) and includes the following additional capabilities:

- Optional Earth Orientation (UT1 and UTC) at a nearby (relevant) reference epoch;
- Optional Leap second specification
- Optional area cross-sections for drag, SRP perturbations modeling.
- Optional spacecraft dimensions and orientation information for collision probability estimation
- Optional orbit states (specified using one or more of Cartesian and orbit elements and reference frames) for a single or parent object at either a single epoch or as a time history (ephemeris);

- Optional covariance matrix of selectable/arbitrary order for a single or parent object at either a single epoch or as a time history (ephemeris) which reflects the uncertainty of the orbit solution or simulation used to obtain the nominal states in the orbit state(s);
- Optional covariance content options (e.g. Cartesian 3x3, 6x6, 7x7, or any combination of order, reference frame and orbit elements)
- Optional maneuver specification (impulsive or finite burn);
- Optional perturbations model specification;
- Optional orbit determination data and metrics;

The OCM simultaneously emphasizes flexibility and message conciseness by offering extensive optional content while minimizing mandatory content. The OCM is well-suited for exchanges that (1) involve automated interaction (e.g., computer-to-computer communication where frequent, fast automated time interpretation and processing is required), and (2) involve regular orbit data transfer for numerous objects (e.g. 200,000) using minimal network bandwidth, disk storage, and quantity of files. The OCM allows the user, in a single message/file, to either embed high-fidelity orbit propagation into an ephemeris time history (akin to the OEM ephemeris), or specify orbital states which can be propagated with supplied perturbations model parameters (akin to OPM content), or both.

2.5 EXCHANGE OF MULTIPLE MESSAGES

For a given object, multiple OPM, OMM, or OEM messages may be provided in a message exchange session to achieve ephemeris fidelity requirements, whereas a single, self-contained OCM may be sufficient. If ephemeris information for multiple objects is to be exchanged, then multiple OPM, OMM, OEM, or OCM files must be used, with the exception that the OCM supports parent/child deployment scenario specifications in a single message.

2.6 DEFINITIONS

Definitions of time systems, reference frames, planetary models, maneuvers, and other fundamental topics related to the interpretation and processing of state vectors and spacecraft ephemerides are provided in ANNEX N, Reference [N-1].

3 ORBIT PARAMETER MESSAGE (OPM)

3.1 GENERAL

3.1.1 Orbit information may be exchanged between two participants by sending a state vector (see ANNEX N, Reference [N-1]) for a specified epoch using an Orbit Parameter Message (OPM). The message recipient must have an orbit propagator available that is able to propagate the OPM state vector to compute the orbit at other desired epochs. For this propagation, additional ancillary information (spacecraft properties such as mass, area, and maneuver planning data, if applicable) may be included with the message.

3.1.2 Osculating Keplerian elements and the Gravitational Coefficient may be included in the OPM in addition to the Cartesian state to aid the message recipient in performing consistency checks. If any Keplerian element is included, the entire set of elements must be provided.

3.1.3 If participants wish to exchange mean element information, then the Orbit Mean-Elements Message (OMM) or Orbit Comprehensive Message (OCM) should be the selected message type. (see Sections 4 and 6.)

3.1.4 The use of the OPM is best applicable under the following conditions:

- a) an orbit propagator consistent with the models used to develop the orbit data should be available at the receiver's site;
- b) the receiver's modeling of gravitational forces, solar radiation pressure, atmospheric drag, and thrust phases (see ANNEX N, Reference [N-1]) should fulfill accuracy requirements established between the exchange partners.

3.1.5 The OPM shall be a plain text file consisting of orbit data for a single object.

NOTE: A sequence of OPMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H.

3.1.6 The OPM file-naming scheme should be mutually agreed between message exchange partners.

3.1.7 The method of exchanging OPMs should be mutually agreed between message exchange partners.

NOTE 1: Detailed syntax rules for the OPM are specified in Section 7.

NOTE 2: Example OPMs and associated supplementary (non-normative) information are provided in ANNEX D.

3.2 OPM CONTENT/STRUCTURE

3.2.1 GENERAL

The OPM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) data; and
- d) optional comments (explanatory information).

3.2.2 OPM HEADER

3.2.2.1 Table 3-1 specifies for each header item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

3.2.2.2 Only those keywords shown in table 3-1 shall be used in an OPM header.

Table 3-1: OPM Header

Keyword	Description	Examples of Values	M/O/C
CCSDS_OPM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (allowed in the OPM Header only immediately after the OPM version number). (see Section 7.8 for formatting rules.)	This is a comment	O
CREATION_DATE	File creation date/time in UTC. (For format specification, see Section 7.5.12)	2001-11-06T11:17:33 2002-204T15:56:23Z	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in ANNEX B, Section B1.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	OPM 201113719185 ABC-12_34	O

3.2.3 OPM METADATA

Table 3-2 specifies for each metadata item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

3.2.3.1 Only those keywords shown in Table 3-2 shall be used in OPM metadata.

NOTE – For some keywords (OBJECT_NAME, OBJECT_ID) there are no definitive lists of authorized values maintained by a control authority; the references listed in 1.6 are the best known sources for authorized values to date. For the TIME_SYSTEM keyword, see ANNEX B, Section B3, for guidance and a link to the approved set of values.

Table 3-2: OPM Metadata

Keyword	Description	Examples of Values	M/O/C
COMMENT	Comments (allowed at the beginning of the OPM Metadata). (see Section 7.8 for formatting rules.)	This is a comment	O
OBJECT_NAME	Spacecraft name for which the orbit state is provided. While there is no CCSDS-based restriction on the value for this keyword, it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [3], which include Object name and international designator of the participant). Where OBJECT_NAME is not known or cannot be disclosed, the value should be set to UNKNOWN.	EUTELSAT W1 MARS PATHFINDER STS 106 NEAR UNKNOWN	M
OBJECT_ID	Object identifier of the object for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but values should be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [3]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one letter for the identification of the part brought into space by the launch. In cases where the asset is not listed in reference 3, the UN Office of Outer Space Affairs designator index format is not used, or the content cannot be disclosed, the value should be set to UNKNOWN.	2000-052A 1996-068A 2000-053A 1996-008A UNKNOWN	M
CENTER_NAME	Origin of the OPM reference frame , which shall be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter. Natural bodies shall be selected from the accepted set of values indicated in ANNEX B, Section B2.	EARTH EARTH BARYCENTER MOON SOLAR SYSTEM BARYCENTER SUN JUPITER BARYCENTER STS 106 EROS	M

Keyword	Description	Examples of Values	M/O/C
REF_FRAME	Reference frame in which the state vector and optional Keplerian element data are given. Use of values other than those in 3.2.3.2 should be documented in an ICD.	ICRF1 ITRF2000 J2000 TEME	M
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (see Section 7.5.12 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	C
TIME_SYSTEM	Time system used for state vector, maneuver, and covariance data (also see Table 3-3). Select from the accepted set of values indicated in ANNEX B, Section B3.	UTC, TAI, TT, GPS, TDB, TCB	M

3.2.3.2 Values for the REF_FRAME keyword should be selected from the following set:

REF_FRAME Value	Meaning
EME2000	Earth Mean Equator and Equinox of J2000
GCRF	Geocentric Celestial Reference Frame
GRC	Greenwich Rotating Coordinates
ICRF	International Celestial Reference Frame
ITRF2000	International Terrestrial Reference Frame 2000
ITRF-93	International Terrestrial Reference Frame 1993
ITRF-97	International Terrestrial Reference Frame 1997
MCI	Mars Centered Inertial
TDR	True of Date, Rotating
TEME	True Equator Mean Equinox (only used in OMMs)
TOD	True of Date

3.2.4 OPM DATA

3.2.4.1 Table 3-3 provides an overview of the six logical blocks in the OPM Data section (State Vector, Osculating Keplerian Elements, Spacecraft Parameters, Position/Velocity

Covariance Matrix, Maneuver Parameters, and User-Defined Parameters), and specifies for each data item:

- a) the keyword to be used;
- b) a short description of the item;
- c) the units to be used;
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

3.2.4.2 Only those keywords shown in Table 3-3 shall be used in OPM data.

NOTE – Requirements relating to the keywords in Table 3-3 appear after the table.

Table 3-3: OPM Data

Keyword	Description	Units	M/O/C
State Vector Components in the Specified Coordinate System			
COMMENT	(see Section 7.8 for formatting rules.)		O
EPOCH	Epoch of state vector & optional Keplerian elements. (see Section 7.5.12 for formatting rules.)		M
X	Position vector X-component	km	M
Y	Position vector Y-component	km	M
Z	Position vector Z-component	km	M
X_DOT	Velocity vector X-component	km/s	M
Y_DOT	Velocity vector Y-component	km/s	M
Z_DOT	Velocity vector Z-component	km/s	M
Osculating Keplerian Elements in the Specified Reference Frame (none or all parameters of this block must be given.)			
COMMENT	(see Section 7.8 for formatting rules.)		O
SEMI_MAJOR_AXIS	Semi-major axis	km	C
ECCENTRICITY	Eccentricity		C
INCLINATION	Inclination	deg	C
RA_OF_ASC_NODE	Right ascension of ascending node	deg	C
ARG_OF_PERICENTER	Argument of pericenter	deg	C
TRUE_ANOMALY or MEAN_ANOMALY	True anomaly or mean anomaly	deg	C
GM	Gravitational Coefficient (Gravitational Constant x Central Mass)	km**3/s**2	C
Spacecraft Parameters (if maneuver is specified, then all parameters of this block must be given)			
COMMENT	(see Section 7.8 for formatting rules.)		O
MASS	Spacecraft mass	kg	C
SOLAR_RAD_AREA	Solar Radiation Pressure Area (A _R)	m**2	C
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient (C _R)		C
DRAG_AREA	Drag Area (A _D)	m**2	C
DRAG_COEFF	Drag Coefficient (C _D)		C
Position/Velocity Covariance Matrix (6x6 Lower Triangular Form. None or all parameters of the matrix must be given. COV_REF_FRAME may be omitted if it is the same as REF_FRAME.)			
COMMENT	(see Section 7.8 for formatting rules.)		O
COV_REF_FRAME	Reference frame in which the covariance data are given. Select from the accepted set of values indicated in 3.2.4.11.		C
CX_X	Covariance matrix [1,1]	km**2	C
CY_X	Covariance matrix [2,1]	km**2	C
CY_Y	Covariance matrix [2,2]	km**2	C
CZ_X	Covariance matrix [3,1]	km**2	C
CZ_Y	Covariance matrix [3,2]	km**2	C
CZ_Z	Covariance matrix [3,3]	km**2	C
CX_DOT_X	Covariance matrix [4,1]	km**2/s	C
CX_DOT_Y	Covariance matrix [4,2]	km**2/s	C
CX_DOT_Z	Covariance matrix [4,3]	km**2/s	C
CX_DOT_X_DOT	Covariance matrix [4,4]	km**2/s**2	C
CY_DOT_X	Covariance matrix [5,1]	km**2/s	C
CY_DOT_Y	Covariance matrix [5,2]	km**2/s	C
CY_DOT_Z	Covariance matrix [5,3]	km**2/s	C
CY_DOT_X_DOT	Covariance matrix [5,4]	km**2/s**2	C

Keyword	Description	Units	M/O/C
CY_DOT_Y_DOT	Covariance matrix [5,5]	km**2/s**2	C
CZ_DOT_X	Covariance matrix [6,1]	km**2/s	C
CZ_DOT_Y	Covariance matrix [6,2]	km**2/s	C
CZ_DOT_Z	Covariance matrix [6,3]	km**2/s	C
CZ_DOT_X_DOT	Covariance matrix [6,4]	km**2/s**2	C
CZ_DOT_Y_DOT	Covariance matrix [6,5]	km**2/s**2	C
CZ_DOT_Z_DOT	Covariance matrix [6,6]	km**2/s**2	C
Maneuver Parameters (Repeat for each maneuver)			
COMMENT	(see Section 7.8 for formatting rules.)		O
MAN_EPOCH_IGNITION	Epoch of ignition. (see Section 7.5.12 for formatting rules.)		O
MAN_DURATION	Maneuver duration (If = 0, impulsive maneuver)	s	O
MAN_DELTA_MASS	Mass change during maneuver (value is < 0)	kg	O
MAN_REF_FRAME	Reference frame in which the velocity increment vector data are given. Select from the accepted set of values indicated in 3.2.4.11.		O
MAN_DV_1	1 st component of the velocity increment	km/s	O
MAN_DV_2	2 nd component of the velocity increment	km/s	O
MAN_DV_3	3 rd component of the velocity increment	km/s	O
User-Defined Parameters (all parameters in this section must be described in an ICD).			
USER_DEFINED_x	User-defined parameter, where 'x' is replaced by a variable length user specified character string. Any number of user-defined parameters may be included, if necessary to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED_EARTH_MODEL = WGS-84		O

3.2.4.3 All values except Maneuver Parameters in the OPM data are ‘at epoch’, i.e., the value of the parameter at the time specified in the EPOCH keyword.

3.2.4.4 Table 3-3 is broken into six logical blocks, each of which has a descriptive heading. These descriptive headings shall not be included in an OPM, unless they appear in a properly formatted COMMENT statement.

3.2.4.5 If the solar radiation coefficient, C_R , is set to zero, no solar radiation pressure shall be considered.

3.2.4.6 If the atmospheric drag coefficient, C_D , is set to zero, no atmospheric drag shall be considered.

3.2.4.7 Parameters for thrust phases may be optionally given for the computation of the trajectory during or after maneuver execution (see ANNEX N, Reference [N-1] for the simplified modeling of such maneuvers). For impulsive maneuvers, MAN_DURATION must be set to zero. MAN_DELTA_MASS may be used for both finite and impulsive maneuvers; the value must be a negative number.

3.2.4.8 Multiple sets of maneuver parameters may appear. For each maneuver, all the maneuver parameters shall be repeated in the order shown in Table 3-3.

3.2.4.9 If the OPM contains a maneuver definition, then the Conditional elements of the Spacecraft Parameters section (designated with a “C”) must be included.

3.2.4.10 Values in the covariance matrix shall be expressed in the applicable reference frame (COV_REF_FRAME keyword), and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row left to right. Variance and covariance values shall be expressed in standard double precision as related in 7.5. This logical block of the OPM may be useful for risk assessment and establishing maneuver and mission margins. The intent is to provide causal connections between output orbit data and both physical hypotheses and measurement uncertainties. These causal relationships guide operators' corrective actions and mitigations.

3.2.4.11 Values for the MAN_REF_FRAME and COV_REF_FRAME keyword may be selected from the following set:

Reference Frame Value	Meaning
RSW	Another name for 'Radial, Transverse, Normal'
RTN	Radial, Transverse, Normal
TNW	A local orbital coordinate frame that has the x-axis along the velocity vector, W along the orbital angular momentum vector, and N completes the right handed system.

3.2.4.12 A section of User-Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User-Defined Parameters, if included, should be used as sparingly as possible; their use is not encouraged.

4 ORBIT MEAN-ELEMENTS MESSAGE (OMM)

4.1 GENERAL

4.1.1 Orbit information may be exchanged between two participants by sending an orbital state based on mean Keplerian elements (see ANNEX N, Reference [N-1]) for a specified epoch using an Orbit Mean-Elements Message (OMM). The message recipient must use appropriate orbit propagator algorithms in order to correctly propagate the OMM state to compute the orbit at other desired epochs.

4.1.2 The OMM is intended to allow replication of the data content of an existing TLE in a CCSDS standard format, but the message can also accommodate other implementations of mean elements. All essential fields of the 'de facto standard' TLE are included in the OMM in a style that is consistent with that of the other ODMs (i.e., the OPM and OEM). From the fields in the OMM, it is possible to generate a TLE (see ANNEX N, Reference [N-2]). Programs that convert OMMs to TLEs must be aware of the structural requirements of the TLE, including the checksum algorithm and the formatting requirements for the values in the TLE. The checksum and formatting requirements of the TLE do not apply to the values in an OMM.

4.1.3 If participants wish to exchange osculating element information, then the Orbit Parameter Message (OPM) or the Orbit Comprehensive Message (OCM) should be the selected message type. (see Sections 3 and 6.)

4.1.4 The use of the OMM is best applicable under the following conditions:

- a) an orbit propagator consistent with the models used to develop the orbit data should be run at the receiver's site;
- b) the receiver's modeling of gravitational forces, solar radiation pressure, atmospheric drag, etc. (see ANNEX N, Reference [N-1]), should fulfill accuracy requirements established between the exchange partners.

4.1.5 The OMM shall be a plain text file consisting of orbit data for a single object.

NOTE: A sequence of OMMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H.

4.1.6 The OMM file-naming scheme should be mutually agreed between message exchange partners.

4.1.7 The method of exchanging OMMs should be mutually agreed between message exchange partners.

NOTE 1: Detailed syntax rules for the OMM are specified in Section 7.

NOTE 2: Example OMMs and associated supplementary (non-normative) information are provided in ANNEX E.

4.2 OMM CONTENT/STRUCTURE

4.2.1 GENERAL

The OMM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) data; and
- d) optional comments (explanatory information).

4.2.2 OMM HEADER

4.2.2.1 Table 4-1 specifies for each header item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

4.2.2.2 Only those keywords shown in Table 4-1 shall be used in an OMM header.

Table 4-1: OMM Header

Keyword	Description	Examples of Values	M/O/C
CCSDS_OMM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (allowed in the OMM Header only immediately after the OMM version number). (see Section 7.8 for formatting rules.)	This is a comment	O
CREATION_DATE	File creation date/time in UTC. (For format specification, see Section 7.5.12.)	2001-11-06T11:17:33 2002-204T15:56:23Z	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in ANNEX B, Section B1.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M

MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	OMM 201113719185 ABC-12_34	O
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4.2.3 OMM METADATA

4.2.3.1 Table 4-2 specifies for each metadata item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

4.2.3.2 Only those keywords shown in Table 4-2 shall be used in OMM metadata.

NOTE – For some keywords (OBJECT_NAME and OBJECT_ID there are no definitive lists of authorized values maintained by a control authority; the references listed in 1.6 are the best known sources for authorized values to date.

Table 4-2: OMM Metadata

Keyword	Description	Examples of Values	M/O/C
COMMENT	Comments (allowed at the beginning of the OMM Metadata). (see Section 7.8 for formatting rules.)	This is a comment	O
OBJECT_NAME	Spacecraft name for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference 3), which include Object name and international designator of the participant. In cases where OBJECT_NAME is not known or cannot be disclosed, the value should be set to UNKNOWN.	TelKom 2 Spaceway 2 INMARSAT 4-F2 UNKNOWN	M

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Keyword	Description	Examples of Values	M/O/C
OBJECT_ID	Object identifier of the object for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but values should be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference 3). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch. In cases where the asset is not listed in reference 3, the UN Office of Outer Space Affairs designator index format is not used, or the content cannot be disclosed, the value should be set to UNKNOWN.	2005-046A 2005-046B 2003-022A UNKNOWN	M
CENTER_NAME	Origin of the OMM reference frame , which shall be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter. Natural bodies shall be selected from the accepted set of values indicated in ANNEX B, Section B2.	EARTH MARS MOON	M
REF_FRAME	Name of the reference frame in which the Keplerian element data are given. Select from the accepted set of values indicated in 3.2.3.2. Note: NORAD Two Line Element Sets and corresponding SGP orbit propagator ephemeris outputs are explicitly defined to be in the True Equator Mean Equinox of Date (TEME of Date) reference frame. Therefore, TEME of date shall be used for OMMs based on NORAD Two Line Element sets, rather than the almost imperceptibly different TEME of Epoch (see ANNEX N, Reference [N-2] or [N-3] for further details).	TEME J2000	M
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (see Section 7.5.12 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	C
TIME_SYSTEM	Time system used for the orbit state and covariance matrix. Select from the accepted set of values indicated in ANNEX B, Section B3.	UTC	M
MEAN_ELEMENT_THEORY	Description of the Mean Element Theory. Indicates the proper method to employ to propagate the state.	SGP4 DSST USM	M

4.2.4 OMM DATA

4.2.4.1 Table 4-3 provides an overview of the five logical blocks in the OMM Data section (Mean Keplerian Elements, Spacecraft Parameters, TLE Related Parameters, Position/Velocity Covariance Matrix, and User-Defined Parameters), and specifies for each data item:

- a) the keyword to be used;
- b) a short description of the item;
- c) the units to be used;
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

4.2.4.2 Only those keywords shown in Table 4-3 shall be used in OMM data.

NOTE – Requirements relating to the keywords in Table 4-3 appear after the table.

Table 4-3: OMM Data

Keyword	Description	Units	M/O/C
Mean Keplerian Elements in the Specified Reference Frame			
COMMENT	(see Section 7.8 for formatting rules.)		O
EPOCH	Epoch of Mean Keplerian elements. (see Section 7.5.12 for formatting rules.)		M
SEMI_MAJOR_AXIS or MEAN_MOTION	Semi-major axis in kilometers (preferred), or, if MEAN_ELEMENT_THEORY = SGP/SGP4, the Keplerian Mean motion in revolutions per day	km rev/day	M
ECCENTRICITY	Eccentricity		M
INCLINATION	Inclination	deg	M
RA_OF_ASC_NODE	Right ascension of ascending node	deg	M
ARG_OF_PERICENTER	Argument of pericenter	deg	M
MEAN_ANOMALY	Mean anomaly	deg	M
GM	Gravitational Coefficient (Gravitational Constant x Central Mass)	km**3/s**2	O
Spacecraft Parameters			
COMMENT	(see Section 7.8 for formatting rules.)		O
MASS	Spacecraft Mass	kg	O
SOLAR_RAD_AREA	Solar Radiation Pressure Area (A _R)	m**2	O
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient (C _R)		O
DRAG_AREA	Drag Area (A _D)	m**2	O
DRAG_COEFF	Drag Coefficient (C _D)		O
TLE Related Parameters (This section is only required if MEAN_ELEMENT_THEORY=SGP/SGP4)			
COMMENT	(see Section 7.8 for formatting rules.)		O

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Keyword	Description	Units	M/O/C
EPHEMERIS_TYPE	Default value = 0. (see Section 4.2.4.7.)		O
CLASSIFICATION_TYPE	Default value = U. (see Section 4.2.4.7.)		O
NORAD_CAT_ID	NORAD Catalog Number ('Satellite Number') an integer of up to nine digits. This keyword is only required if MEAN_ELEMENT_THEORY=SGP/SGP4.		O
ELEMENT_SET_NO	Element set number for this satellite. Normally incremented sequentially, but may be out of sync if it is generated from a backup source. Used to distinguish different TLEs, and therefore only meaningful if TLE-based data is being exchanged (i.e., MEAN_ELEMENT_THEORY = SGP/SGP4).		O
REV_AT_EPOCH	Revolution Number		O
BSTAR	SGP/SGP4 drag-like coefficient (in units 1/[Earth radii]). Only required if MEAN_ELEMENT_THEORY=SGP/SGP4	1/ER	O
MEAN_MOTION_DOT	First Time Derivative of the Mean Motion (only required if MEAN_ELEMENT_THEORY = SGP)	rev/day**2	O
MEAN_MOTION_DDOT	Second Time Derivative of Mean Motion (only required if MEAN_ELEMENT_THEORY = SGP)	rev/day**3	O
Position/Velocity Covariance Matrix (6x6 Lower Triangular Form. None or all parameters of the matrix must be given. COV_REF_FRAME may be omitted if it is the same as the REF_FRAME.)			
COMMENT	(see Section 7.8 for formatting rules.)		O
COV_REF_FRAME	Reference frame in which the covariance data are given. Select from the accepted set of values indicated 3.2.4.11.		C
CX_X	Covariance matrix [1,1]	km**2	C
CY_X	Covariance matrix [2,1]	km**2	C
CY_Y	Covariance matrix [2,2]	km**2	C
CZ_X	Covariance matrix [3,1]	km**2	C
CZ_Y	Covariance matrix [3,2]	km**2	C
CZ_Z	Covariance matrix [3,3]	km**2	C
CX_DOT_X	Covariance matrix [4,1]	km**2/s	C
CX_DOT_Y	Covariance matrix [4,2]	km**2/s	C
CX_DOT_Z	Covariance matrix [4,3]	km**2/s	C
CX_DOT_X_DOT	Covariance matrix [4,4]	km**2/s**2	C
CY_DOT_X	Covariance matrix [5,1]	km**2/s	C
CY_DOT_Y	Covariance matrix [5,2]	km**2/s	C
CY_DOT_Z	Covariance matrix [5,3]	km**2/s	C
CY_DOT_X_DOT	Covariance matrix [5,4]	km**2/s**2	C
CY_DOT_Y_DOT	Covariance matrix [5,5]	km**2/s**2	C
CZ_DOT_X	Covariance matrix [6,1]	km**2/s	C
CZ_DOT_Y	Covariance matrix [6,2]	km**2/s	C
CZ_DOT_Z	Covariance matrix [6,3]	km**2/s	C
CZ_DOT_X_DOT	Covariance matrix [6,4]	km**2/s**2	C
CZ_DOT_Y_DOT	Covariance matrix [6,5]	km**2/s**2	C
CZ_DOT_Z_DOT	Covariance matrix [6,6]	km**2/s**2	C
User-Defined Parameters (all parameters in this section must be described in an ICD).			
USER_DEFINED_x	User-defined parameter, where 'x' is replaced by a variable length user specified character string. Any number of user-defined parameters may be included, if necessary to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED_EARTH_MODEL = WGS-84		O

4.2.4.3 All values in the OMM are ‘at epoch’, i.e., the value of the parameter at the time specified in the EPOCH keyword.

4.2.4.4 Table 4-3 is broken into five logical blocks, each of which has a descriptive heading. These descriptive headings shall not be included in an OMM, unless they appear in a properly formatted COMMENT statement.

4.2.4.5 Values in the covariance matrix shall be expressed in the applicable reference frame (COV_REF_FRAME keyword if used, or REF_FRAME keyword if not), and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row left to right. Variance and covariance values shall be expressed in standard double precision as related in Section 7.5. This logical block of the OMM may be useful for risk assessment and establishing maneuver and mission margins.

4.2.4.6 For operations in Earth orbit with a TLE-based OMM, some special conventions must be observed, as follows:

- The value associated with the CENTER_NAME keyword shall be ‘EARTH’.
- The value associated with the REF_FRAME keyword shall be ‘TEME’.
- The value associated with the TIME_SYSTEM keyword shall be ‘UTC’.
- The format of the OBJECT_NAME and OBJECT_ID keywords shall be that of the UN Office of Outer Space Affairs designator index (reference 3).
- The MEAN_MOTION keyword must be used instead of SEMI_MAJOR_AXIS.

4.2.4.7 For those who wish to use the OMM to represent a TLE, there are a number of considerations that apply with respect to precision of angle representation, use of certain fields by the propagator, reference frame, etc. Some sources suggest the coding for the EPHEMERIS_TYPE keyword: 1=SGP, 2=SGP4, 3=SDP4, 4=SGP8, 5=SDP8. Some sources suggest the following coding for the CLASSIFICATION_TYPE keyword: U=unclassified, S=secret. (For further information see ANNEX N, Reference [N-2] and [N-3])

4.2.4.8 Maneuvers are not accommodated in the OMM. Users of the OMM who wish to model maneuvers may use several OMM files to describe the orbit at applicable epochs.

4.2.4.9 NORAD Two Line Element Sets are implicitly in a True Equator Mean Equinox (TEME) of Date reference frame, which is ill defined in international standard or convention. TEME may be used only for OMMs based on NORAD Two Line Element sets, and in no other circumstances. There are subtle differences between TEME of Epoch and TEME of Date (see reference [N-2] and [N-3]). The effect is very small relative to TLE accuracy. The preferred option is TEME of Date. Users should specify in the ICD if their assumption is TEME of Epoch.

4.2.4.10 A section of User-Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User-Defined Parameters, if included, should be used as sparingly as possible; their use is not encouraged.

5 ORBIT EPHEMERIS MESSAGE (OEM)

5.1 GENERAL

5.1.1 Orbit information may be exchanged between two participants by sending an ephemeris in the form of a series of state vectors (Cartesian vectors providing position and velocity, and optionally accelerations) using an Orbit Ephemeris Message (OEM). The message recipient must have a means of interpolating across these state vectors to obtain the state at an arbitrary time contained within the span of the ephemeris.

5.1.2 The OEM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and representing the orbits of active satellites, inactive man-made objects, near-Earth debris fragments, etc. The OEM reflects the dynamic modeling of any users' approach to conservative and non-conservative phenomena.

5.1.3 The OEM shall be a plain text file consisting of orbit data for a single object.

NOTE: A sequence of OEMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H.

5.1.4 The OEM file-naming scheme should be mutually agreed between message exchange partners.

5.1.5 The method of exchanging OEMs should be mutually agreed between message exchange partners.

NOTE 1: Detailed syntax rules for the OEM are specified in Section 7.

NOTE 2: Example OEMs and associated supplementary (non-normative) information are provided in ANNEX F.

5.2 OEM CONTENT/STRUCTURE

5.2.1 GENERAL

5.2.1.1 The OEM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) ephemeris data;
- d) optional covariance matrix data; and
- e) optional comments (explanatory information).

5.2.1.2 OEM files must have a set of minimum required sections; some may be repeated. Table 5-1 outlines the contents of an OEM.

Table 5-1: OEM File Layout Specifications

Required Sections	Header Metadata Ephemeris Data (Appropriate comments should also be included, although they are not required.)
Allowable Repetitions of Sections	Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) ...etc. (Appropriate comments should also be included.)

5.2.2 OEM HEADER

5.2.2.1 The OEM header assignments are shown in Table 5-2, which specifies for each item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). **An ‘M’ denotes mandatory keywords that must be included in this section if that particular data section is included.** Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

5.2.2.2 Only those keywords shown in Table 5-2 shall be used in an OEM header.

Table 5-2: OEM Header

Keyword	Description	Examples of Values	M/O/C
CCSDS_OEM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (allowed in the OEM Header only immediately after the OEM version number). (see Section 7.8 for formatting rules.)	COMMENT This is a comment	O
CREATION_DATE	File creation date and time in UTC. (For format specification, see Section 7.5.12.)	2001-11-06T11:17:33 2002-204T15:56:23	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in ANNEX B, Section B1.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	OEM 201113719185 ABC-12_34	O

5.2.3 OEM METADATA

5.2.3.1 The OEM metadata assignments are shown in Table 5-3, which specifies for each item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is Mandatory (M), Optional (O), or Conditional (C). Conditional indicates that the item is mandatory if specified conditions are met (e.g., providing all covariance matrix elements if any are provided).

5.2.3.2 Only those keywords shown in Table 5-3 shall be used in OEM metadata.

NOTE – For some keywords (OBJECT_NAME and OBJECT_ID) there are no definitive lists of authorized values maintained by a control authority; the references listed in 1.6 are the best known sources for authorized values to date. For the TIME_SYSTEM keyword, see ANNEX B, Section B3, for guidance and a link to the approved set of values.

5.2.3.3 A single metadata group shall precede each ephemeris data block. Multiple occurrences of a metadata group followed by an ephemeris data block may be used. Before each metadata group the string 'META_START' shall appear on a separate line and after each metadata group (and before the associated ephemeris data block) the string 'META_STOP' shall appear on a separate line.

Table 5-3: OEM Metadata

Keyword	Description	Examples of Values	M/O/C
META_START	The OEM message contains metadata, ephemeris data, and covariance data; this keyword is used to delineate the start of a metadata block within the message (metadata are provided in a block, surrounded by 'META_START' and 'META_STOP' markers to facilitate file parsing). This keyword must appear on a line by itself.	n/a	M
COMMENT	Comments allowed only immediately after the META_START keyword. (see Section 7.8 for formatting rules.)	COMMENT This is a comment.	O
OBJECT_NAME	The name of the object for which the ephemeris is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference 3), which include Object name and international designator of the participant. In cases where OBJECT_NAME is not known or cannot be disclosed, the value should be set to UNKNOWN.	EUTELSAT W1 MARS PATHFINDER STS 106 NEAR UNKNOWN	M
OBJECT_ID	Object identifier of the object for which the ephemeris is provided. There is no CCSDS-based restriction on the value for this keyword, but values should be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference 3). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch. In cases where the asset is not listed in reference 3, the UN Office of Outer Space Affairs designator index format is not used, or the content cannot be disclosed, the value should be set to UNKNOWN.	2000-052A 1996-068A 2000-053A 1996-008A UNKNOWN	M

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Keyword	Description	Examples of Values	M/O/C
CENTER_NAME	<p>Origin of the OEM reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, or another reference frame center (such as a spacecraft, formation flying reference “chief” spacecraft, etc.).</p> <p>Natural bodies shall be selected from the accepted set of values indicated in ANNEX B, Section B2.</p> <p>For spacecraft, it is recommended to use either the OBJECT_ID or international designator of the participant as catalogued in the UN Office of Outer Space Affairs designator index (reference 3).</p>	<p>EARTH EARTH BARYCENTER MOON SOLAR SYSTEM BARYCENTER SUN JUPITER BARYCENTER STS 106 EROS</p>	M
REF_FRAME	Reference frame in which the ephemeris data are given. Select from the accepted set of values indicated in 3.2.3.2.	<p>ICRF3 ITRF1993 ITRF1997 ITRF2000 ITRFyyyy (template for future versions) TOD (True Equator and Equinox of Date) J2000 (Earth Mean Equator and Equinox of J2000) TDR (true of date rotating) GRC (Greenwich rotating coordinate frame, another name for TDR)</p>	M
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (see Section 7.5.12 for formatting rules.)	<p>2001-11-06T11:17:33 2002-204T15:56:23Z</p>	C
TIME_SYSTEM	Time system used for metadata, ephemeris and covariance data (also see Table 3-3). Select from the accepted set of values indicated in ANNEX B, Section B3.	UTC, TAI, TT, GPS, TDB, TCB	M
START_TIME	Start of TOTAL time span covered by ephemeris data and covariance data immediately following this metadata block. (For format specification, see Section 7.5.12.)	<p>1996-12-18T14:28:15.1172 1996-277T07:22:54</p>	M

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Keyword	Description	Examples of Values	M/O/C
USEABLE_START_TIME	<p>Start time of USEABLE time span covered by ephemeris data immediately following this metadata block. For format specification, see Section 7.5.12.</p> <p>This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes prior to the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p>	<p>1996-12- 18T14:28:15.1172 1996-277T07:22:54</p>	O
USEABLE_STOP_TIME	<p>Stop time of USEABLE time span covered by ephemeris data immediately following this metadata block. For format specification, see Section 7.5.12.</p> <p>This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes following the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p>	<p>1996-12- 18T14:28:15.1172 1996-277T07:22:54</p>	O
STOP_TIME	<p>End of TOTAL time span covered by ephemeris data and covariance data immediately following this metadata block. (For format specification, see Section 7.5.12.)</p>	<p>1996-12- 18T14:28:15.1172 1996-277T07:22:54</p>	M
INTERPOLATION	<p>This keyword may be used to specify the recommended interpolation method for ephemeris data in the immediately following set of ephemeris lines. PROPAGATE indicates that orbit propagation is the preferred method to obtain states at intermediate times, via either a midpoint-switching or endpoint switching approach.</p>	<p>HERMITE LINEAR LAGRANGE PROPAGATE</p>	O
INTERPOLATION_DEGREE	<p>Recommended interpolation degree for ephemeris data in the immediately following set of ephemeris lines. Must be an integer value. This keyword must be used if the 'INTERPOLATION' keyword is used and set to anything other than PROPAGATE.</p>	<p>5 1</p>	C
META_STOP	<p>The OEM message contains metadata, ephemeris data, and covariance data; this keyword is used to delineate the end of a metadata block within the message (metadata are provided in a block, surrounded by 'META_START' and 'META_STOP' markers to facilitate file parsing). This keyword must appear on a line by itself.</p>	n/a	M

5.2.4 OEM DATA: EPHEMERIS DATA LINES

5.2.4.1 Each set of ephemeris data, including the time tag, must be provided on a single line. The order in which data items are given shall be fixed: **Epoch, X, Y, Z, X_DOT, Y_DOT, Z_DOT, X_DDOT, Y_DDOT, Z_DDOT**.

5.2.4.2 The position and velocity terms shall be mandatory; acceleration terms may be provided.

5.2.4.3 At least one space character must be used to separate the items in each ephemeris data line.

5.2.4.4 Repeated time tags may occur in consecutive ephemeris data blocks if the STOP_TIME of the first ephemeris data block is greater than the START_TIME of the second ephemeris data block. Although the USEABLE_STOP_TIME and USEABLE_START_TIME of the consecutive ephemeris data blocks must not overlap (except for a possibly shared endpoint), the STOP_TIME of the first ephemeris data block may be greater than the START_TIME of the second ephemeris data block if the extra data is required for interpolation purposes.

5.2.4.5 The TIME_SYSTEM value must remain fixed within an OEM.

5.2.4.6 The occurrence of a second (or greater) metadata block after some ephemeris data indicates that interpolation using succeeding ephemeris data with ephemeris data occurring prior to that metadata block shall not be done. This method may be used for proper modeling of propulsive maneuvers or any other source of a discontinuity such as eclipse entry or exit.

5.2.4.7 Details about interpolation method should be specified using the INTERPOLATION and INTERPOLATION_DEGREE keywords within the OEM. All data blocks must contain a sufficient number of ephemeris data records to allow the recommended interpolation method to be carried out consistently throughout the OEM.

5.2.5 OEM DATA: COVARIANCE MATRIX LINES

5.2.5.1 A single covariance matrix data section may optionally follow each ephemeris data block.

5.2.5.2 If present, the covariance matrix data lines in the OEM are separated from the ephemeris data by means of two mandatory keywords as specified in Table 5-4: 'COVARIANCE_START' and 'COVARIANCE_STOP'. The COVARIANCE_START keyword must appear before the first line of the covariance matrix data. The COVARIANCE_STOP keyword must appear after the last line of covariance data. Each of these keywords shall appear on a line by itself with no time tags or values.

5.2.5.3 The epoch of the navigation solution related to the covariance matrix must be provided via the 'EPOCH' keyword. The reference frame of the covariance matrix, if different

from that of the states in the ephemeris, must be provided via the ‘COV_REF_FRAME’ keyword.

5.2.5.4 Each row of the 6x6 lower triangular covariance matrix must be provided on a single line. The order in which data items are given shall be fixed. The elements in each row of covariance data shall be defined by the order in the ephemeris data line (i.e., X, Y, Z, X_DOT, Y_DOT, Z_DOT). The six rows of the covariance matrix contain from one to six numbers depending on what row of the matrix is being represented (first row has one element, second row has two, continuing in this fashion until the sixth row has six elements).

5.2.5.5 At least one space character must be used to separate the items in each covariance matrix data line.

5.2.5.6 Multiple covariance matrices may appear in the covariance matrix section; they may appear with any desired frequency (one for each navigation solution that makes up the overall ephemeris is recommended). The OEM may also contain propagated covariances, not just individual covariances associated with navigation solutions.

5.2.5.7 If there are multiple covariance matrices in the data section, they must be ordered by increasing time tag.

Table 5-4: OEM Covariance Keywords

Keyword	Description	Examples of Values	M/O/C
COVARIANCE_START	This keyword is used to delineate the start of a covariance data block within the message.	n/a	M
COV_REF_FRAME	Reference frame in which the covariance data are given. Select from the accepted set of values indicated in 3.2.3.2 or 3.2.4.11.	ICRF3 ITRF1993 ITRF1997 ITRF2000 ITRFyyyy (template for future versions) TOD (True Equator and Equinox of Date) J2000 (Earth Mean Equator and Equinox of J2000) TDR (true of date rotating) GRC (Greenwich rotating coordinate frame, another name for TDR)	C
COVARIANCE_STOP	This keyword is used to delineate the end of a covariance data block within the message.	n/a	M

6 ORBIT COMPREHENSIVE MESSAGE (OCM)

6.1 GENERAL DESCRIPTION

6.1.1 Comprehensive orbit information may be exchanged between two participants by sending orbit data/content for one or more epochs using an Orbit Comprehensive Message (OCM). The OCM aggregates and extends OMM, OPM, and OEM content in a single hybrid message. The OCM simultaneously emphasizes flexibility and message conciseness by offering extensive optional standardized content while minimizing mandatory content.

6.1.2 The OCM shall be a plain text file consisting of orbit data for a single space object, or in the case of a parent/child satellite deployment scenario, a single parent object.

NOTE: A sequence of OCMs for either a single object or for multiple objects can be aggregated into a single Navigation Data Message (NDM) XML file as described in Section 8.12 and shown in ANNEX H.

6.1.3 Orbit information may be exchanged between two or more participants by sending an ephemeris in the form of one or more time series of orbital states (selectable as orbital elements and/or Cartesian vectors providing position and optionally velocity and accelerations) using an Orbit Comprehensive Message (OCM). If orbital states are desired at arbitrary time(s) contained within the span of the provided orbit or covariance time histories, the message recipient must use a suitable interpolation method. For times outside of supplied orbit or covariance state time spans or if the step size between time points is too large to support interpolation (ANNEX N, Reference [N-6]), optional perturbations parameters should be included in this message to allow the message recipient to use a suitably-compatible orbit and covariance propagator.

6.1.4 The OCM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and representing the orbits of active satellites, inactive man-made objects, near-Earth debris fragments, etc. The OCM reflects the dynamic modeling of any users' approach to conservative and non-conservative phenomena.

6.1.5 The OCM file-naming scheme should be mutually agreed between message exchange partners.

6.1.6 The method of exchanging OCMs should be mutually agreed between message exchange partners.

NOTE 1: Detailed syntax rules for the OCM are specified in Section 7.

NOTE 2: Example OCMs and associated supplementary (non-normative) information are provided in ANNEX G.

6.2 OCM STRUCTURE AND OVERARCHING REQUIREMENTS

6.2.1 GENERAL STRUCTURE

6.2.1.1 The OCM shall be represented as the combination of the following **mandatory (M)** and **optional (O)** elements, ordered as listed in Table 6-1.

6.2.1.2 Within the tables of each OCM section, each keyword is labeled as being Mandatory (M), Optional (O), or Conditional (C). **An ‘M’ denotes mandatory keywords that must be included in this section if that particular data section is included.** Keywords that have a pre-defined (default) value are listed as ‘O’ (optional), because if the keyword is not provided, then that default value as defined in the corresponding table shall be used in OCM processing. **A ‘C’ denotes keywords that are mandatory if this particular data block is included and certain conditions are met**, as specified in the keyword description.

NOTE: One can observe in Table 6-1 that the OCM fully supports what might be considered as a “degenerate” case, where the message is constructed without any data blocks. This was an intentional choice, given that the many metadata elements the OCM can accommodate are very useful (e.g., to convey phonebook information, link disparate messages together, and convey timing source information).

Table 6-1: OCM File Layout and Ordering Specification

Section		Content	Status M/O	
OCM Header		A single header of the message	M	
OCM Metadata		A single metadata section (data about data)	M	
Data	orbit data 1	data description data lines	One or more orbit state time histories (consisting of one or more orbit states)	
	⋮			
	orbit data n	data description data lines		
	physical properties		A single space object physical characteristics section	O
	covariance data 1	data description data lines	One or more covariance time histories (each consisting of one or more covariance matrices)	
	⋮			
	covariance data n	data description data lines		
	maneuver data 1	data description data lines	One or more maneuver specifications for either impulsive or finite burns or acceleration profiles	
	⋮			
	maneuver data n	data description data lines		
perturbation parameters		A single perturbations parameters section (required if an orbit determination section is provided)	C	
orbit determination		A single orbit determination data section	O	
user-defined parameters		A single user-defined parameters section containing data and supplemental comments (explanatory information)	O	

6.2.1 GENERAL REQUIREMENTS

The following requirements apply to all OCM sections and content:

6.2.1.1 The order of occurrence of OCM keywords shall be fixed as listed in the keyword value tables in the OCM section descriptions.

6.2.1.2 If the message creator does not have a value for a mandatory keyword, a value of “UNKNOWN” can be used.

6.2.1.3 All time-tags may be specified by either a (signed) double precision relative time (e.g., 20157.26) measured in SI seconds with respect to the specified epoch time (EPOCH_TZERO) or as an absolute time (e.g., 2018-11-13T11:13:20.5Z in CCSDS Time String A or B format, as specified in Section 7.5.10).

6.2.1.4 Duplicate time tags shall not be used.

6.2.1.5 Within an OCM data block, all time-tags must adhere to either relative time, or absolute time, for the entirety of that data block. Relative and absolute time shall not be used within the same data block.

6.2.1.6 Time tags of information within ordered sequences of OCM sections may be separated by uniform or non-uniform step size(s).

6.2.1.7 Time tags of one OCM section may or may not match those of another OCM section.

6.2.2 OCM HEADER

6.2.2.1 Table 6-2 specifies the keywords for each header item.

6.2.2.2 Only those keywords shown in Table 6-2 shall be used in an OCM header.

6.2.2.3 The order of occurrence of these OCM header keywords shall be fixed as shown in Table 6-2.

Table 6-2: OCM Header

Keyword	Description	Examples of Values	M/O/C
CCSDS_OCM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	3.0	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Header only immediately after the OCM version number). (see Section 7.8 for formatting rules).	This is a comment	O
CREATION_DATE	File creation date/time in UTC. (For format specification, see Section 7.5.12).	2001-11-06T11:17:33 2002-204T15:56:23Z	M
ORIGINATOR	Creating agency or operator. Select from the accepted set of values indicated in ANNEX B, Section B1.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT, USAF, INMARSAT	M
MESSAGE_ID	Free text field containing an ID that uniquely identifies a message from this message originator. The format and content of the message identifier value are at the discretion of the originator.	OCM 201113719185 ABC-12_34	O

6.2.3 OCM METADATA

6.2.3.1 Table 6-3 specifies the metadata keywords. Only those keywords shown in Table 6-3 shall be used in OCM metadata.

6.2.3.2 The metadata section must begin with keyword META_START and end with keyword META_STOP.

6.2.3.3 At most, only one metadata section shall appear in the entire scope of an OCM.

6.2.3.4 The order of occurrence of these OCM metadata keywords shall be fixed as shown in Table 6-3.

NOTE 1: For some keywords (OBJECT_NAME, OBJECT_DESIGNATOR) there are no definitive lists of authorized values maintained by a control authority; the references listed in Section 1.6 are the best known sources for authorized values to date.

NOTE 2 – While specification of OBJECT_NAME, OBJECT_DESIGNATOR, and INTERNATIONAL_DESIGNATOR are individually optional, it is recommended that at least one of these three keywords be supplied.

NOTE 3 – Metadata fields which are relied upon by the subsequent optional OCM message subtypes (orbit state time histories, maneuver data, etc.) are designated as such in the right-hand column of Table 6-3.

Table 6-3: OCM Metadata

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
META_START	Start of the metadata section.				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Metadata section; see Section 7.8 for comment formatting rules).			This is a comment	O
CLASSIFICATION	User-defined free-text message classification/caveats of this OCM. It is recommended that selected values be pre-coordinated between exchanging entities by mutual agreement.			SBU "Operator-proprietary data; secondary distribution not permitted"	O
OBJECT_NAME	Free text field containing the name of the object. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from either the UN Office of Outer Space Affairs designator index (reference 3, which include Object name and international designator of the participant), the spacecraft operator, or a State Actor or commercial SSA provider maintaining the "CATALOG_NAME" space catalog. If the object name is not known (uncorrelated object), "UNKNOWN" may be used (or this keyword omitted).			SPOT-7 ENVISAT IRIDIUM NEXT-8 INTELSAT G-15 UNKNOWN	O
INTERNATIONAL_DESIGNATOR	Free text field containing an international designator for the object as assigned by the UN Committee on Space Research (COSPAR) and the US National Space Science Data Center (NSSDC). Such designator values have the following COSPAR format: YYYY-NNN{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch. In cases where the object has no international designator, the value UNKNOWN may be used. NOTE: The international designator was typically specified by "OBJECT_ID" in the OPM, OMM, and OEM.			2000-052A 1996-068A 2000-053A 1996-008A UNKNOWN	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
CATALOG_NAME	Free text field containing the satellite catalog source (or source agency or operator, value to be drawn from the abbreviated "Organizations" name column of the SANA registry at https://www.sanaregistry.org/r/organizations) from which OBJECT_DESIGNATOR was obtained.		CSPOC	CSPOC RFSA ESA COMSPOC	O
OBJECT_DESIGNATOR	Free text field specification of the unique satellite identification designator for the object, as reflected in the catalog whose name is "CATALOG_NAME". If the ID is not known (uncorrelated object), "UNKNOWN" may be used (or this keyword omitted).			22444 18SPCS 18571 2147483648_04ae[...]d 84c UNKNOWN	O
ALTERNATE_NAMES	Free text comma-delimited field containing alternate name(s) of this space object, including assigned names used by spacecraft operator, State Actors, commercial SSA providers and/or media.			SV08, IN8	O
ORIGINATOR_POC	Free text field containing originator or programmatic Point-of-Contact (PoC) for OCM.			Mr. Rodgers	O
ORIGINATOR_POSITION	Free text field containing contact position of the originator PoC.			Flight Dynamics Mission Design Lead	O
ORIGINATOR_PHONE	Free text field containing originator PoC phone number.			+12345678901	O
ORIGINATOR_ADDRESS	Free text field containing originator PoC address information for OCM creator (suggest email, website, or physical address, etc.).			JOHN.DOE@ SOMEWHERE.ORG	O
TECH_ORG	Free text field containing the creating agency or operator (value should be drawn from the abbreviated "Organizations" name column of the SANA registry at https://www.sanaregistry.org/r/organizations).			NASA	O
TECH_POC	Free text field containing technical Point-of-Contact (PoC) for OCM.			Maxwell Smart	O
TECH_POSITION	Free text field containing contact position of the technical PoC.			Flight Dynamics Mission Design Lead	O
TECH_PHONE	Free text field containing technical PoC phone number.			+49615130312	O
TECH_ADDRESS	Free text field containing technical PoC address information for OCM creator (suggest email, website, or physical address, etc.).			JOHN.DOE@ SOMEWHERE.ORG	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
PREVIOUS_MESSAGE_ID	Free text field containing an ID that uniquely identifies the previous message from this message originator for this particular space object. The format and content of the message identifier value are at the discretion of the originator. NOTE: One may provide the previous message ID without supplying the PREVIOUS_MESSAGE_EPOCH, and vice versa.			OCM 201113719184 ABC-12_33	O
NEXT_MESSAGE_ID	Free text field containing an ID that uniquely identifies the next message from this message originator for this particular space object. The format and content of the message identifier value are at the discretion of the originator. NOTE: One may provide the next message ID without supplying the NEXT_MESSAGE_EPOCH, and vice versa.			OCM 201113719186 ABC-12_35	O
ADM_MSG_LINK	Free text field containing a unique identifier of Attitude Data Message(s) that are linked (relevant) to this Orbit Data Message.			ADM_MSG_35132.txt ADM_ID_0572	O
CDM_MSG_LINK	Free text field containing a unique identifier of Conjunction Data Message(s) that are linked (relevant) to this Orbit Data Message.			CDM_MSG_35132.txt CDM_ID_8257	O
PRM_MSG_LINK	Free text field containing a unique identifier of Pointing Request Message(s) that are linked (relevant) to this Orbit Data Message.			PRM_MSG_35132.txt PRM_ID_6897	O
RDM_MSG_LINK	Free text field containing a unique identifier of Reentry Data Message(s) that are linked (relevant) to this Orbit Data Message.			RDM_MSG_35132.txt RDM_ID_1839	O
TDM_MSG_LINK	An free-text string containing a comma-separated list of file name(s) and/or associated identification number(s) of Tracking Data Message (TDM) [9] observations upon which this OD is based.			TDM_MSG_37.txt TDM_835, TDM_836	O
OPERATOR	Free text field containing the operator of the space object.			INTELSAT	O
OWNER	Free text field containing the owner of the space object.			SIRIUS	O
COUNTRY	Free text field containing the name of the country, country code, or country abbreviation where the space object owner is based.			US SPAIN	O
CONSTELLATION	Free text field containing the name of the constellation to which this space object belongs.			SPIRE	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OBJECT_TYPE	Specification of the type of object. Select from the accepted set of values indicated in ANNEX B, Section B11.			PAYLOAD ROCKET BODY DEBRIS UNKNOWN OTHER	O
TIME_SYSTEM	Time system for all absolute time stamps in this OCM including EPOCH_TZERO. Select from the accepted set of values indicated in ANNEX B, Section B3. This field is used by all OCM data blocks. If the spacecraft clock (SCLK) timescale is selected, then "EPOCH_TZERO" shall be interpreted as the spacecraft clock epoch and both SCLK_OFFSET_AT_EPOCH and SCLK_SEC_PER_SI_SEC shall be supplied.		UTC	UTC	O
EPOCH_TZERO	Default epoch to which <u>all</u> relative times are referenced in data blocks (for format specification, see Section 7.5.10.). The time scale of EPOCH_TZERO is controlled via the "TIME_SYSTEM" keyword, with the exception that for the spacecraft clock (SCLK) timescale, EPOCH_TZERO shall be interpreted as being in the UTC timescale. This field is used by all OCM data blocks.			2001-11-06T11:17:33	M
OPS_STATUS	Specification of the operational status of the space object. Select from the accepted set of values indicated in ANNEX B, Section B12.			OPERATIONAL	O
ORBIT_CATEGORY	Specification of the type of orbit. Select from the accepted set of values indicated in ANNEX B, Section B14.			GEO LEO	O
OCM_DATA_ELEMENTS	Comma-delimited list of elements of information data blocks included in this message. The order shall be the same as the order of the data blocks in the message. Values shall be confined to the following list: ORB, PHYS, COV, MAN, PERT, OD, USER. If the OCM contains multiple ORB, COV or MAN data blocks (as allowed by Table 6.1), the corresponding ORB, COV or MAN entry shall be duplicated to match.			ORB, ORB, PHYS, COV, MAN, MAN, PERT, OD, USER	O
SCLK_OFFSET_AT_EPOCH	Defines the number of spacecraft clock counts existing at EPOCH_TZERO. This is only used if the spacecraft clock (SCLK) timescale is employed by the user.	s	0.0	-5000.0	C
SCLK_SEC_PER_SI_SEC	Defines the current number of clock seconds occurring during one SI second. Note that this clock rate may vary with time and is the current approximate value. This is only used if the spacecraft clock (SCLK) timescale is employed by the user.	s	1.0	2.5 [s]	C

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
PREVIOUS_MESSAGE_EPOCH	Creation epoch of the previous message from this originator for this particular space object. For format specification, see Section 7.5.10. NOTE: One may provide the previous message epoch without supplying the PREVIOUS_MESSAGE_ID, and vice versa.			2001-11-06T11:17:33	O
NEXT_MESSAGE_EPOCH	Anticipated (or actual) epoch of the next message from this originator for this particular space object. For format specification, see Section 7.5.10. NOTE: One may provide the next message epoch without supplying the NEXT_MESSAGE_ID, and vice versa.			2001-11-07T11:17:33	O
START_TIME	Time of the earliest data contained in the OCM, specified as either a relative or absolute time tag.			0.0 2001-11-06T00:00:00	O
STOP_TIME	Time of the latest data contained in the OCM, specified as either a relative or absolute time tag.			86400.0 2001-11-08T00:00:00	O
TIME_SPAN	Span of time that the OCM covers, measured in days. TIME_SPAN is defined as (STOP_TIME-START_TIME), measured in days, irrespective of whether START_TIME or STOP_TIME are actually provided by the message creator.	d		20.0	O
TAIMUTC_AT_TZERO	Difference (TAI – UTC) in seconds (i.e. total number of leap seconds elapsed since 1958) as modeled by the message originator at epoch “EPOCH_TZERO”.	s		36	O
UT1MUTC_AT_TZERO	Difference (UT1 – UTC) in seconds, as modeled by the originator at epoch “EPOCH_TZERO”.	s		0.357	O
EOP_SOURCE	Free text field specifying the source and version of the message originator’s Earth Orientation Parameters (EOP) used in the creation of this message, including leap seconds, TAI-UT1, etc.			CELESTRAK_20201028	O
INTERP_METHOD_EOP	Free text field specifying the method used to select or interpolate sequential EOP data.		LINEAR	PRECEDING_VALUE NEAREST_NEIGHBOR LINEAR LAGRANGE_ORDER_5	O
CELESTIAL_SOURCE	Free text field specifying the source and version of the message originator’s celestial body (e.g., Sun/Earth/Planetary) ephemeris data used in the creation of this message.			JPL_DE_FILES	O
META_STOP	End of the metadata section.				M

6.2.4 OCM DATA: ORBIT STATE TIME HISTORY

6.2.4.1 Table 6-4 provides an overview of the OCM orbit state time history (“ephemeris”) section. Only those keywords shown in Table 6-4 shall be used in the OCM orbit state time history data specification.

6.2.4.2 Each orbit state time history data block must begin with keyword ORB_START and end with keyword ORB_STOP.

6.2.4.3 Multiple orbit state data blocks shall appear in an OCM only if they are delimited by separate ORB_START and ORB_STOP keywords

6.2.4.4 Each orbit state time history data block should differ from all others in at least one of the following respects:

- 1) the selected element set (ORB_TYPE)
- 2) the orbit basis, i.e., the orbit determination, navigation solution, or simulation (ORB_BASIS_ID)
- 3) the reference frame is unique (ORB_REF_FRAME)
- 4) the orbit center is unique (CENTER_NAME)
- 5) the data interval timespan is unique (i.e., has no overlap with any other data interval(s))

6.2.4.5 Where multiple orbit state time history data blocks are provided for the same ORB_BASIS and ORB_BASIS_ID, the top-most depiction shall be adopted as the true or master depiction.

6.2.4.6 Each orbit state time history shall be time-ordered to be monotonically increasing.

6.2.4.7 Positionally-discontinuous orbit states (i.e., separated by a gap in the orbit state time history data across which interpolation should not be performed) shall be represented by separate orbit state time history data blocks.

6.2.4.8 Velocity-discontinuous orbit states (i.e., by introduction of an impulsive maneuver) shall be represented by separate orbit state time history data blocks.

6.2.4.9 All orbit state time history blocks must contain a sufficient number of data records to allow the recommended interpolation method to be carried out consistently throughout the time span.

6.2.4.10 If the user includes orbit states at key mission events or times, it may be useful to provide times, names, and significance for such mission events in the descriptive comment line(s) immediately following the ORB_START keyword.

6.2.4.11 Each line of orbit ephemeris data shall be provided in fixed order beginning with an absolute or relative time tag, followed by the corresponding orbit state elements as defined by ORB_TYPE (see SANA registry [11] and ANNEX B, Section B7).

6.2.4.12 At least one space character must be used to separate the items in each orbit ephemeris data line.

6.2.4.13 The number of significant figures and time steps suitable for interpolation of an orbit ephemeris time history should be chosen according to best practice to avoid positional and interpolation loss of precision [ANNEX N, Reference N-6].

6.2.4.14 If an orbit state time history section is included in the message, a corresponding perturbations section should be included as well to specify the perturbations incorporated in these orbit states.

6.2.4.15 The CENTER_NAME shall be used to specify the origin of the reference frame that the orbit time history is specified in. The specified center may either be a natural, gravitationally-attracting body such as is provided in ANNEX B Section B2, or it may be a non-gravitationally-attracting origin to allow relative positional time histories. If a non-gravitationally-attracting origin is selected, however, then the specified ORB_TYPE shall be confined to Cartesian or spherical coordinates, where the reference frame may be rotating or inertially fixed.

Table 6-4: OCM Data: Orbit State Time History

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ORB_START	Start of an orbit state vector or time history section.			n/a	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the Orbit State Time History section only immediately after the ORB_START key word; see Section 7.8 for comment formatting rules).			This is a comment	O
ORB_ID	Alphanumeric free-text string containing the identification number for this orbit state time history block.			ORB_20160402_XYZ	O
ORB_PREV_ID	Alphanumeric free-text string containing the identification number for the previous orbit time history, contained either within this message, or presented in a previous OCM. Note: if this message is not part of a sequence of orbit time histories or if this orbit time history is the first in a sequence of orbit time histories, then ORB_PREV_ID should be excluded from this message.			ORB20160305A	O
ORB_NEXT_ID	Alphanumeric free-text string containing the identification number for the next orbit time history, contained either within this message, or presented in a future OCM. Note: if this message is not part of a sequence of orbit time histories or if this orbit time history is the last in a sequence of orbit time histories, then ORB_NEXT_ID should be excluded from this message.			ORB20160305C	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ORB_BASIS	<p>Basis of this Orbit State time history data, this is free text field with the following suggested values:</p> <ol style="list-style-type: none"> “PREDICTED”. “DETERMINED_OD” when estimated from observation-based orbit determination reconstruction and/or calibration. For definitive OD performed onboard whose solutions have been telemetered to the ground for inclusion in an OCM, the ORB_BASIS will be considered to be DETERMINED_OD. “DETERMINED_TLM” when read directly from telemetry, e.g., based on inertial navigation systems or GNSS data. “SIMULATED” for generic simulations, future mission design studies, and optimization studies. “OTHER” for other bases of this data. 		PREDICTED	PREDICTED	O
ORB_BASIS_ID	<p>Alphanumeric free-text string containing the identification number for the orbit determination, navigation solution, or simulation upon which this orbit state time history block is based. Where a matching orbit determination block accompanies this orbit state time history, the ORB_BASIS_ID should match the corresponding OD_ID.</p>			OD_5910	O
INTERPOLATION	<p>This keyword may be used to specify the recommended interpolation method for ephemeris data in the immediately following set of ephemeris lines. PROPAGATE indicates that orbit propagation is the preferred method to obtain states at intermediate times, via either a midpoint-switching or endpoint switching approach.</p>		HERMITE	HERMITE LINEAR LAGRANGE PROPAGATE	O
INTERPOLATION_DEGREE	<p>Recommended interpolation degree for ephemeris data in the immediately following set of ephemeris lines. Must be an integer value. This keyword must be used if the ‘INTERPOLATION’ keyword is used and set to anything other than PROPAGATE.</p>		3	5 1	C
ORB_AVERAGING	<p>Specifies whether the provided orbit state/elements are based upon either osculating or mean elements, and if so, which mean element definition is employed. The intent of this field is to allow the user to correctly interpret how to use the provided orbit elements and know how to use them operationally.</p> <p>Values should be selected from the accepted set indicated in ANNEX B, Section B12. If an alternate single- or double-averaging formulation is used other than provided, the user may name as mutually agreed by message exchange participants.</p>		OSCULATING	OSCULATING BROUWER KOZAI (other...)	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
CENTER_NAME	<p>Origin of the orbit reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, or another reference frame center (such as a spacecraft, formation flying reference “chief” spacecraft, etc.).</p> <p>Natural bodies shall be selected from the accepted set of values indicated in ANNEX B, Section B2.</p> <p>For spacecraft, it is recommended to use either the Object name or international designator of the participant as catalogued in the UN Office of Outer Space Affairs designator index (reference 3). Alternately, the OBJECT_DESIGNATOR may be used.</p> <p>For other reference frame origins, this field is a free text descriptor which may draw upon other naming conventions and sources.</p>		EARTH	EARTH MOON SOLAR SYSTEM BARYCENTER SUN ISS EROS EARTH_SUN _L2 EGLIN	O
ORB_REF_FRAME	Reference frame of the orbit state time history. Select from the accepted set of values indicated in ANNEX B, Section B4.		ICRF3	J2000	O
ORB_FRAME_EPOCH	Epoch of the orbit data reference frame, if not intrinsic to the definition of the reference frame. See 7.5.12 for formatting rules.		EPOCH_TZER O	2001-11- 06T11:17:33 2002- 204T15:56:2 3Z	C
USEABLE_START_TIME	<p>Start time of USEABLE time span covered by ephemeris data immediately following this metadata block. For format specification, see Section 7.5.12.</p> <p>This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes prior to the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p>		(defaults to all data being valid)	1996-12- 18T14:28 :15.1172 1996- 277T07:2 2:54	O
USEABLE_STOP_TIME	<p>Stop time of USEABLE time span covered by ephemeris data immediately following this metadata block. For format specification, see Section 7.5.12.</p> <p>This optional keyword allows the message creator to introduce fictitious (but numerically smooth) data nodes following the actual data time history to support interpolation methods requiring more than two nodes (e.g., pure higher-order Lagrange interpolation methods). The use of this keyword and introduction of fictitious node points are optional and may not be necessary.</p>		(defaults to all data being valid)	1996-12- 18T14:28 :15.1172 1996- 277T07:2 2:54	O
ORB_TYPE	Specifies the orbit element set type; selected per ANNEX B, Section B7.		CARTPV	n/a	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ORB_UNITS	A comma-delimited set of SI unit designations for each element of the orbit state time history following the orbit state time tag solely for informational purposes, provided as an alphanumeric free-text string enclosed in square brackets. When provided, each orbit state element shall have a corresponding units entry, with non-dimensional values (such as orbit eccentricity) denoted by "n/a". Note that the listing of units via the ORB_UNITS keyword does not override the mandatory units specified for the selected ORB_TYPE , as discussed in ANNEX B, Section B7.			[km,km,km,k m/s,km/s,k m/s] [km,n/a,deg, deg, deg, deg]	O
... < Insert orbit state time history here >	Orbit time history line(s) shall be formatted as specified in 6.2.4.11, containing time and orbit elements as selected by ORB_TYPE in the SANA Orbital Elements registry (ANNEX B, Section B7). Units are as specified in this registry).				M
ORB_STOP	End of an orbit state vector or time history section.			n/a	M

6.2.5 OCM DATA: SPACE OBJECT PHYSICAL CHARACTERISTICS

6.2.5.1 Table 6-5 gives an overview of the OCM space object physical characteristics section. Only those keywords shown in Table 6-5 shall be used in OCM space object physical characteristics data.

6.2.5.2 At most, only one space object physical characteristics section shall appear in an OCM.

6.2.5.3 The space object physical characteristics data section in the OCM shall be indicated by two keywords: PHYS_START and PHYS_STOP.

6.2.5.4 The Space Object Optimally-Encompassing Box (OEB) parameters are defined in further detail in Informative Annex C1.

6.2.5.5 Modeling of cross-sectional area and the contributions of relevant parameters (DRAG_CONST_AREA, SRP_CONST_AREA, AREA_ALONG_OEB_MAX, AREA_ALONG_OEB_INT, and AREA_ALONG_OEB_MIN) to total cross-sectional area is provided in Informative Annex C1.

Table 6-5: OCM Data: Space Object Physical Characteristics

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
PHYS_START	Start of a Space Object Physical Characteristics specification				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Space Object Physical Characteristics only immediately after the PHYS_START keyword; see Section 7.8 for comment formatting rules).			This is a comment	O
MANUFACTURER	Free text field containing the satellite manufacturer name.			BOEING	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
BUS_MODEL	Free text field containing the satellite manufacturer's spacecraft bus model name.			702	O
DOCKED_WITH	Free text field containing a comma-separated list of other space objects that this object is docked to.			ISS	O
DRAG_CONST_AREA	Attitude-independent drag cross-sectional area (A_D) facing the relative wind vector, not already incorporated into the attitude-dependent "AREA_ALONG_OEB" parameters.	m**2		2.5	O
DRAG_COEFF_NOM	Nominal drag Coefficient ($C_{D_{Nom}}$). If the atmospheric drag coefficient, C_D , is set to zero, no atmospheric drag shall be considered.			2.2	O
DRAG_UNCERTAINTY	Drag coefficient one sigma (1σ) percent uncertainty, where the actual range of drag coefficients to within 1σ shall be obtained from $\left[1.0 \pm \frac{DRAGUNCERTAINTY}{100.0}\right] (C_{D_{Nom}})$. This factor is intended to allow operators to supply the nominal ballistic coefficient components while accommodating ballistic coefficient uncertainties.	%	0.0	10.0	O
INITIAL_WET_MASS	Space object total mass at beginning of life.	kg		500	O
WET_MASS	Space object total mass (i.e., "wet mass") at the current reference epoch "EPOCH_TZERO".	kg		472.3	O
DRY_MASS	Space object dry mass (without propellant).	kg		300	O
OEB_PARENT_FRAME	Parent reference frame which maps to the Optimally Enclosing Box (OEB) frame via the quaternion-based transformation defined in ANNEX C, Section C1. Select from the accepted set of values indicated in ANNEX B, Sections B4 and B5.		RSW_ROT ATING	ITRF1997	O
OEB_PARENT_FRAME_EPOCH	Epoch of the OEB parent frame, if not intrinsic to the definition of the reference frame. (see Section 7.5.12 for formatting rules.).		EPOCH_T ZERO	2001-11- 06T11:17:33 2002-204T15:56:23Z	C
OEB_Q1	$q_1 = e_1 * \sin(\phi/2)$, where per ANNEX N, Reference [N-1], ϕ = Euler rotation angle and e_1 = 1st component of Euler rotation axis for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally-Encompassing Box (OEB) (defined in ANNEX C, Section C1). A value of "-999" denotes a tumbling space object.			-0.575131822	O
OEB_Q2	$q_2 = e_2 * \sin(\phi/2)$, where per ANNEX N, Reference [N-1], ϕ = Euler rotation angle and e_2 = 2nd component of Euler rotation axis for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally-Encompassing Box (defined in ANNEX C, Section C1). A value of "-999" denotes a tumbling space object.			-0.280510532	O
OEB_Q3	$q_3 = e_3 * \sin(\phi/2)$, where per ANNEX N, Reference [N-1], ϕ = Euler rotation angle and e_3 = 3rd component of Euler rotation axis for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally-Encompassing Box (defined in ANNEX C, Section C1). A value of "-999" denotes a tumbling space object.			-0.195634856	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OEB_QC	$q_c = \cos(\phi/2)$, where per ANNEX N, Reference [N-1], ϕ = Euler axis/angle rotation angle for the rotation that maps from the OEB_PARENT_FRAME (defined above) to the frame aligned with the Optimally-Encompassing Box (ANNEX C, Section C1). q_c shall be made non-negative by convention. A value of “-999” denotes a tumbling space object.			0.743144825	O
OEB_MAX	Maximum physical dimension (along \hat{X}_{OEB}) of the Optimally-Encompassing Box (OEB) in meters,	m		1	O
OEB_INT	Intermediate physical dimension (along \hat{Y}_{OEB}) of Optimally-Encompassing Box (OEB) normal to OEB_MAX direction.	m		0.5	O
OEB_MIN	Minimum physical dimension (along \hat{Z}_{OEB}) of Optimally-Encompassing Box (OEB) in direction normal to both OEB_MAX and OEB_INT directions.	m		0.3	O
AREA_ALONG_OEB_MAX	Attitude-dependent cross-sectional area of space object (not already included in DRAG_CONST_AREA and SRP_CONST_AREA) when viewed along max OEB (\hat{X}_{OEB}) direction as defined in ANNEX C.	m**2		0.15	O
AREA_ALONG_OEB_INT	Attitude-dependent cross-sectional area of space object (not already included in DRAG_CONST_AREA and SRP_CONST_AREA) when viewed along intermediate OEB (\hat{Y}_{OEB}) direction as defined in ANNEX C.	m**2		0.3	O
AREA_ALONG_OEB_MIN	Attitude-dependent cross-sectional area of space object (not already included in DRAG_CONST_AREA and SRP_CONST_AREA) when viewed along minimum OEB (\hat{Z}_{OEB}) direction as defined in ANNEX C.	m**2		0.5	O
AREA_MIN_FOR_PC	Minimum cross-sectional area for collision probability estimation purposes.	m**2		1.0	O
AREA_MAX_FOR_PC	Maximum cross-sectional area for collision probability estimation purposes.	m**2		1.0	O
AREA_TYP_FOR_PC	Typical (50 th percentile) cross-sectional area sampled over all space object orientations for collision probability estimation purposes.	m**2		1.0	O
RCS	Typical (50 th percentile) effective Radar Cross Section of the space object sampled over all possible viewing angles.	m**2		1.25	O
RCS_MIN	Minimum Radar Cross Section observed for this object.	m**2		1.1	O
RCS_MAX	Maximum Radar Cross Section observed for this object .	m**2		2.5	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
SRP_CONST_AREA	Attitude-independent solar radiation pressure cross-sectional area (A_R) facing the Sun, not already incorporated into the attitude-dependent "AREA_ALONG_OEB" parameters (computed from $\{ \text{AREA_ALONG_OEB_MAX} \cos(\theta_1) + \text{AREA_ALONG_OEB_INT} \cos(\theta_2) + \text{AREA_ALONG_OEB_MIN} \cos(\theta_3) \}$ Where θ_i represents the angle between the normal to each MAX/MED/MIN face and the direction to the Sun.	m**2		1.0	O
SOLAR_RAD_COEFF	Nominal Solar Radiation Pressure Coefficient (C_{RNom}). Note that if the solar radiation coefficient, C_R , is set to zero, no solar radiation pressure shall be considered.			1.7	O
SOLAR_RAD_UNCERTAINTY	Solar Radiation Pressure one sigma (1σ) percent uncertainty, where the actual range of SRP coefficients to within 1σ shall be obtained from $\left[1.0 \pm \frac{SRP_UNCERTAINTY}{100.0} \right] (C_{RNom})$. This factor is intended to allow operators to supply the nominal ballistic coefficient components while accommodating ballistic coefficient uncertainties.	%		1.0	O
VM_ABSOLUTE	Typical (50 th percentile) absolute Visual Magnitude of the space object sampled over all possible viewing angles and "normalized" as discussed in Informative Annex C2 to a 1 AU Sun-to-target distance, a phase angle of 0° and a 40,000 km target-to-sensor distance (equivalent of GEO satellite tracked at 15.6° above local horizon).			15.0	O
VM_APPARENT_MIN	Minimum apparent Visual Magnitude observed for this space object.			19.0	O
VM_APPARENT	Typical (50 th percentile) apparent Visual Magnitude observed for this space object.			15.0	O
VM_APPARENT_MAX	Maximum apparent Visual Magnitude observed for this space object (NOTE: The "MAX" value represents the brightest observation, which associates with a lower Vmag) .			16.0	O
REFLECTIVITY	Typical (50 th percentile) coefficient of reflectivity of the space object over all possible viewing angles, ranging from -1.0 to +1.0.			0.7	O
ATT_CONTROL_MODE	Free-text specification of primary mode of attitude control for the space object. Suggested examples include: <ul style="list-style-type: none"> - THREE_AXIS - SPIN - DUAL_SPIN - TUMBLING - GRAVITY_GRADIENT 			SPIN	O

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ATT_ACTUATOR_TYPE	Free-text specification of type of actuator for attitude control. Suggested examples include: <ul style="list-style-type: none"> - ATT_THRUSTERS - ACTIVE_MAG_TORQUE - PASSIVE_MAG_TORQUE - REACTION_WHEELS - MOMENTUM_WHEELS - CONTROL_MOMENT_GYROSCOPE - NONE - OTHER 			ATT_THRUSTERS	O
ATT_KNOWLEDGE	Accuracy of attitude knowledge.	deg		0.3	O
ATT_CONTROL	Accuracy of attitude control system (ACS) to maintain attitude, assuming attitude knowledge was perfect (i.e., deadbands).	deg		2.0	O
ATT_POINTING	Overall accuracy of spacecraft to maintain attitude, including attitude knowledge errors and ACS operation.	deg		2.3	O
AVG_MANEUVER_FREQ	Average maneuver frequency, measured in the number of orbit- or attitude-adjust maneuvers per year.	#/yr		20.0	O
MAX_THRUST	Maximum composite thrust the spacecraft can accomplish in any single body-fixed direction.	N		1.0	O
DV_BOL	Total ΔV capability of the spacecraft at beginning of life.	km/s		1.0	O
DV_REMAINING	Total ΔV remaining for the spacecraft.	km/s		0.2	O
IXX	Moment of Inertia about the X-axis of the space object's primary body frame (e.g. SC_Body_1) – see ANNEX N, Reference [N-1].	kg*m ^{**2}		1000.0	O
IYY	Moment of Inertia about the Y-axis.	kg*m ^{**2}		800.0	O
IZZ	Moment of Inertia about the Z-axis.	kg*m ^{**2}		400.0	O
IXY	Inertia Cross Product of the X & Y axes.	kg*m ^{**2}		20.0	O
IXZ	Inertia Cross Product of the X & Z axes.	kg*m ^{**2}		40.0	O
IYZ	Inertia Cross Product of the Y & Z axes.	kg*m ^{**2}		60.0	O
PHYS_STOP	End of the Space Object Physical Characteristics specification.				M

6.2.6 OCM DATA: COVARIANCE TIME HISTORY

6.2.6.1 Table 6-6 provides an overview of the OCM covariance time history section. Only those keywords shown in Table 6-6 shall be used in the OCM covariance time history data specification.

6.2.6.2 Each covariance time history data block must begin with keyword COV_START and end with keyword COV_STOP.

6.2.6.3 Multiple orbit state covariance data blocks shall appear in an OCM only if they are delimited by separate COV_START and COV_STOP keywords

6.2.6.4 Each covariance time history data block should differ from all others in at least one of the following respects:

- 1) the selected element set (COV_TYPE)
- 2) the orbit basis (COV_BASIS)
- 3) the orbit determination, navigation solution, or simulation (COV_BASIS_ID)
- 4) the reference frame is unique (COV_REF_FRAME)
- 5) the data interval timespan is unique (i.e., has no overlap with any other data interval(s))

6.2.6.5 Where multiple covariance time history data blocks are provided for the same COV_BASIS and COV_BASIS_ID, the top-most depiction shall be adopted as the true or master depiction.

6.2.6.6 Each covariance time history shall be time-ordered to be monotonically increasing.

6.2.6.7 Discontinuous covariance time segments shall be represented by separate covariance time history data blocks.

6.2.6.8 If the user includes orbit state covariances at key mission events or times, it may be useful to provide times, names, and significance for such mission events in descriptive comment line(s) immediately following the COV_START keyword.

6.2.6.9 Values in the orbit state covariance matrix shall be expressed in the applicable reference frame specified via the ‘COV_REF_FRAME’ keyword.

6.2.6.10 If a covariance time history section is included in the message, a corresponding perturbations section should be included as well to specify the perturbations incorporated in these covariances.

6.2.6.10.1 Each covariance time history line shall begin with either a relative or absolute time tag corresponding to the specified covariance matrix.

6.2.6.11 For all covariance representations, the covariance time tag and covariance matrix elements (or dispersions and eigenvectors, in the case of COV_TYPE= SIG3EIGVEC3 or TSIG3EIGVEC3) shall all be presented on a single line.

6.2.6.11.1 The composition of the covariance matrix shall be commensurate with the specified COV_TYPE value.

6.2.6.11.2 Directly following the time tag specification on the same line as the time tag, all elements of the “NxN” Lower Triangular Matrix covariance shall be presented in rowwise fashion, beginning with element [1,1], followed by [2,1], [2,2], [3,1], [3,2] and so on, until all LTM entries (there are $\sum_{i=1}^N i$ entries in total) as shown and ordered in Figure 6-1.

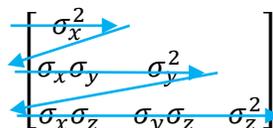


Figure 6-1: Covariance LTM ordering following time tag.

6.2.6.11.3 At least one space character must be used to separate the items in each covariance matrix data line as related in 7.4.1.6.

6.2.6.12 Variance and covariance values shall be expressed in floating point or scientific notation as related in 7.5. The number of significant figures and time steps suitable for interpolation of a covariance time history should be chosen according to best practice to avoid covariance interpolation loss of precision ANNEX N, References [N-6] and [N-13]).

NOTE: It is strongly recommended that covariance matrix time history interpolation be done by either (1) using orbit-dynamics-aware numerical methods as provided in [N-9] or (2) eigenvalue/vector decomposition, linear (or higher-order) interpolation of neighboring eigenvalues; Euler axis/angle rotation of eigenvectors at intermediate time(s) of interest; and re-composition of attained eigenvalues and eigenvectors into covariances at time(s) of interest (see ANNEX C, Section C5 and ANNEX N, References [N-10, N-11, and N-12]). Direct interpolation of covariance matrix components or failure to incorporate a sufficient number of significant figures on the interpolated covariance elements can produce invalid (non-positive-semidefinite) covariances.

Table 6-6: OCM Data: Covariance Time History

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
COV_START	Start of a covariance time history section.			n/a	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM covariance time history section only immediately after the COV_START key word; see Section 7.8 for comment formatting rules).			This is a comment	O
COV_ID	Alphanumeric free-text string containing the identification number for this covariance time history block.			COV_20160402_XYZ	O
COV_PREV_ID	Alphanumeric free-text string containing the identification number for the previous covariance time history, contained either within this message, or presented in a previous OCM. Note: if this message is not part of a sequence of covariance time histories or if this covariance time history is the first in a sequence of covariance time histories, then COV_PREV_ID should be excluded from this message.			COV_20160305A	O
COV_NEXT_ID	Alphanumeric free-text string containing the identification number for the next covariance time history, contained either within this message, or presented in a future OCM. Note: if this message is not part of a sequence of covariance time histories or if this covariance time history is the last in a sequence of covariance time histories, then COV_NEXT_ID should be excluded from this message.			COV_20160305C	O
COV_BASIS	Basis of this covariance time history data. This is free text field with the following suggested values: <ol style="list-style-type: none"> 1. "PREDICTED" 2. "DETERMINED_OD" when estimated from observation-based orbit determination reconstruction and/or calibration. For definitive OD performed onboard whose solutions have been telemetered to the ground for inclusion in an OCM, the COV_BASIS shall be considered to be DETERMINED_OD. 3. EMPIRICAL (for empirically-determined such as overlap analyses) 4. SIMULATED for simulation-based (including Monte Carlo) estimations, future mission design studies, and optimization studies. 5. "OTHER" for other bases of this data 		PREDICTED	PREDICTED EMPIRICAL DETERMINED_OD SIMULATED OTHER	O
COV_BASIS_ID	Alphanumeric free-text string containing the identification number for the orbit determination, navigation solution, or simulation upon which this covariance time history block is based. Where a matching orbit determination block accompanies this covariance time history, the COV_BASIS_ID should match the corresponding OD_ID.			OD_5910	O
COV_REF_FRAME	Reference frame of the covariance time history. Select from the accepted set of values indicated in ANNEX B, Section B4 and B5.		TNW_INERTIAL	J2000	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
COV_FRAME_EPOCH	Epoch of the covariance data reference frame, if not intrinsic to the definition of the reference frame. See 7.5.12 for formatting rules.		EPOCH_TZERO	2001-11-06T11:17:33 2002-204T15:56:23Z	C
COV_SCALE_MIN	Minimum scale factor to apply to this covariance data to achieve realism.		1	0.5	O
COV_SCALE_MAX	Maximum scale factor to apply to this covariance data to achieve realism.		1	5.0	O
COV_CONFIDENCE	A measure of the confidence in the covariance errors matching reality, as characterized via a Wald test, a Chi-squared test, the log of likelihood, or a numerical representation per mutual agreement. This is a free text field.	%		50%	O
COV_TYPE	Indicates covariance composition. Select from ANNEX B, Sections B7 and B8.		CARTP CARTPV ADBARV	n/a	O
COV_UNITS	A comma-delimited set of SI unit designations for each element of the covariance time history following the covariance time tag solely for informational purposes , provided as an alphanumeric free-text string enclosed in square brackets. When provided, these units designations shall correspond to the units of the standard deviations (or square roots) of each of the covariance matrix diagonal elements (or variances), respectively , and all diagonal elements shall have a corresponding units entry, with non-dimensional values (such as dispersion in orbit eccentricity) denoted by "n/a". Note that the listing of units via the COV_UNITS keyword does not override the mandatory units specified for the selected COV_TYPE , as discussed in ANNEX B, Section B7 and B8.			[km,km,km,km/s,km/s, km/s]	O
...< Insert covariance data here>	Covariance time history line(s) shall be formatted as specified in 6.2.6.10.1 through 6.2.6.12 containing time and covariance elements as selected by COV_TYPE in the SANA Orbital Elements or the Additional Covariance Representations registry (ANNEX B, Section B7 and B8). Units are as specified in these registries).				M
COV_STOP	End of a covariance time history section.			n/a	M

6.2.7 OCM DATA: MANEUVER SPECIFICATION

6.2.7.1 Table 6-7 provides an overview of the OCM maneuver specification section. Only those keywords shown in Table 6-7 shall be used in the OCM maneuver time history data specification.

6.2.7.2 The order of occurrence of these OCM Maneuver Specification keywords shall be fixed as shown in Table 6-7.

6.2.7.3 Maneuver data in the OCM shall be indicated by two keywords: MAN_START and MAN_STOP.

6.2.7.4 Multiple maneuver data blocks shall appear in an OCM only when delimited by separate MAN_START and MAN_STOP keywords.

6.2.7.5 The time intervals of multiple maneuver data blocks may be separated in time, abut, or overlap.

NOTE: This is done to accommodate multiple maneuver reference frames, multiple thrusters in simultaneous operation, deployments during thrusting phases, multiple basis definitions (MAN_BASIS), etc.

6.2.7.6 Each maneuver data block shall be assigned a maneuver device ID (MAN_DEVICE_ID) value, which specifies the unique thruster (or other propulsive device) used in this maneuver sequence time history data block.

- Except for the special values “ALL” and “DEPLOY”, MAN_DEVICE_ID is a free text field which allows the user to identify which specific thruster or other propulsive device performed this maneuver time history.
- A MAN_DEVICE_ID value of “ALL” shall be used to indicate that this maneuver sequence represents an aggregation of thrust, acceleration, and/or velocity increments imparted by any/all thrusters utilized in the maneuver which are not attributed to a single specific propulsive device.
- A MAN_DEVICE_ID value of “DEPLOY” shall be used to indicate that this maneuver data block represents ONLY maneuvers caused by a series of one or more deployments from this host vehicle.

6.2.7.7 Multiple maneuver data blocks may invoke the same maneuver device ID (MAN_DEVICE_ID) value.

6.2.7.8 All specified maneuver constituents having a common MAN_ID, MAN_BASIS and MAN_REF_FRAME shall be added together to obtain the total composite maneuver description.

6.2.7.9 Each maneuver data block should differ from all other maneuver data blocks in at least one of the following respects:

- 1) the maneuver device ID (MAN_DEVICE_ID) is unique
- 2) the maneuver device ID is the same, but the “ON” time intervals are unique and do not overlap with any other data interval(s) for this maneuver device ID (e.g., during multiple interleaved duty cycle “ON” firings).
- 3) the maneuver basis (MAN_BASIS) is unique
- 4) the reference frame is unique (MAN_REF_FRAME)
- 5) the maneuver is based upon a unique orbit determination, navigation solution, or simulation (e.g., MAN_BASIS_ID)
- 6) the data interval timespan is unique (i.e., has no overlap)

6.2.7.10 In the event that the only difference between multiple maneuver time history data blocks is the selected maneuver composition (MAN_COMPOSITION) or reference frame (MAN_REF_FRAME), the top-most depiction (i.e., the time history occurring first in the OCM) shall be adopted as the official depiction, and those subsequent data blocks shall be treated as containing informational derivative depictions.

6.2.7.11 The **MAN_COMPOSITION** keyword shall specify the individual maneuver time history elements to follow the maneuver time tag.

NOTE 1: Specification of a time history of acceleration parameters ACC_X, ACC_Y, and ACC_Z allows the OCM originator to portray and share the net effects of maneuvers without the OCM recipient needing to do complex finite burn modeling.

NOTE 2: Specification of ΔV parameters allows modeling of impulsive (instantaneous velocity change) maneuvers.

NOTE 3: Specification of thrust parameters provides a finite burn capability. In the case of low-thrust and/or long-duration burns, sequential low-thrust interval maneuver lines can be used to reflect the evolution of the low-thrust maneuver thrust parameters.

6.2.7.12 The **MAN_COMPOSITION** keyword shall contain a comma-separated list of values taken from either Table 6-8 or Table 6-9 (i.e., values unique to tables Table 6-8 and Table 6-9 shall not be combined within a single maneuver data block).

6.2.7.13 The values contained in the **MAN_COMPOSITION** keyword shall appear in the order fixed in either Table 6-8 or Table 6-9.

6.2.7.13.1 Maneuver time history lines shall be confined to only one spacecraft object.

6.2.7.14 At least one of the time tag formats (TIME_ABSOLUTE and/or TIME_RELATIVE) shall be selected in the beginning of the **MAN_COMPOSITION** specification sequence.

6.2.7.15 Within a single maneuver time history line, acceleration, impulsive ΔV , and thrust parameters shall not be additive, but rather shall be interpreted as alternate representations of the same underlying propulsive phenomenology.

6.2.7.16 Time tag(s) on each maneuver line shall be considered to be at the start of the maneuver, whereas impulsive ΔV entries in the propulsive representation (Table 6-8) shall be interpreted as occurring at a time tag of $T_{\text{start}} + \frac{1}{2} (\text{MAN_DURA})$.

NOTE: While one could artificially make T_{start} and the impulsive maneuver time be the same value by setting MAN_DURA equal to zero, the actual duration of the maneuver is typically nonzero and should be provided if/when known to facilitate improved modeling and maneuver reconstruction.

6.2.7.16.1 When invoked, interpolation of acceleration (ACC_INTERP=ON) and/or thrust vectors (THR_INTERP=ON) shall be done using a suitable interpolation scheme such as the Euler axis/angle formulation discussed in Informative Annex C5.

6.2.7.16.2 Thrust and acceleration levels for any propulsive device shall be presumed to be “OFF” until explicitly turned “ON” by setting one or more thrust or acceleration components to a non-zero value.

6.2.7.16.3 Thrust and acceleration shall be set back to “OFF” after the maneuver duration has elapsed [$T_{\text{start}} + \text{MAN_DURA}$]. Thrusters may also be turned “OFF” by setting all thrust and acceleration components to zero.

6.2.7.16.4 Thruster duty cycles shall be specified using the thrust maneuver specification (DC_TYPE \neq CONTINUOUS) based on either a reference direction or reference time.

NOTE: This duty cycle specification imposes cut-outs of non-thrust periods onto the thrust (finite burn) parameters to reflect the periods of duty cycle inactivity.

6.2.7.16.5 If the value of the DC_TYPE keyword is **not CONTINUOUS**, then the following duty cycle parameters shall be present: DC_WIN_OPEN, DC_WIN_CLOSE, DC_EXEC_START, DC_EXEC_STOP, DC_REF_TIME, DC_TIME_PULSE_DURATION, and DC_TIME_PULSE_PERIOD.

6.2.7.16.6 If the value of the DC_TYPE keyword is **TIME_AND_ANGLE**, then the following duty cycle parameters shall also be present: DC_REF_DIR, DC_BODY_FRAME, DC_BODY_TRIGGER, DC_PA_START_ANGLE, and DC_PA_STOP_ANGLE

6.2.7.16.7 DC_MIN_CYCLES and DC_MAX_CYCLES may be specified to constrain the number of duty cycles performed in either TIME or TIME_AND_ANGLE mode. These parameters may override the duty cycle maneuver stop time (DC_EXEC_STOP).

NOTE 1: Relationships between such duty cycle parameters are described in Informative Annex C3.

NOTE 2: The effects of using a pulse width modulation thruster controller can be accommodated by applying a reduced constant thrust level or by invoking the duty cycle parameters, or a combination thereof (being careful to avoid double-booking of thruster degradations).

Table 6-7: OCM Data: Maneuver Specification

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_START	Start of a maneuver data block.				M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Maneuver Specification only immediately after the MAN_START key word; see Section 7.8 for comment formatting rules).			This is a comment	O
MAN_ID	Alphanumeric free-text string containing the unique maneuver identification number for this maneuver. All supplied maneuver “constituents” within the same MAN_BASIS and MAN_REF_FRAME categories shall be added together to represent the total composite maneuver description.			E_W_20160305B STAGE2	M
MAN_PREV_ID	Alphanumeric free-text string containing the identification number of the previous maneuver for this MAN_BASIS, , contained either within this message, or presented in a previous OCM. Note: if this message is not part of a sequence of maneuver messages or if this maneuver is the first in a sequence of maneuvers, then MAN_PREV_ID should be excluded from this message.			E_W_20160305A	O
MAN_NEXT_ID	Alphanumeric free-text string containing the identification number of the next maneuver for this MAN_BASIS, contained either within this message, or presented in a future OCM. Note: if this message is not part of a sequence of maneuver messages or if this maneuver is the last in a sequence of maneuvers, then MAN_NEXT_ID should be excluded from this message.			E_W_20160305C	O
MAN_BASIS	Basis of this maneuver time history data, which shall be selected from one of the following values: <ol style="list-style-type: none"> 1. “CANDIDATE” for a proposed operational or a hypothetical (i.e., mission design and optimization studies) future maneuver 2. “PLANNED” for a currently planned future maneuver 3. “ANTICIPATED” for a non-cooperative future maneuver that is anticipated (i.e. likely) to occur (e.g., based upon patterns-of-life analysis). 4. “DETERMINED_TLM” when 5. “DETERMINED_OD” when a past maneuver is estimated from observation-based orbit determination reconstruction and/or calibration. 6. “SIMULATED” for generic maneuver simulations, future mission design studies, and optimization studies. 7. “OTHER” for other bases of this data. 		PLANNE D	DETERMINED_TLM CANDIDATE	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_BASIS_ID	Alphanumeric free-text string containing the identification number for the orbit determination, navigation solution, or simulation upon which this maneuver time history block is based. Where a matching orbit determination block accompanies this maneuver time history, the MAN_BASIS_ID should match the corresponding OD_ID.			OD_20181122A	O
MAN_DEVICE_ID	Alphanumeric free-text string containing the maneuver device IDentifier used for this maneuver. "ALL" indicates that this maneuver represents the summed acceleration, velocity increment, or thrust imparted by any/all thrusters utilized in the maneuver.			THR_02 DEPLOYMENT ALL	M
MAN_PREV_EPOCH	Identifies the completion time of the previous maneuver for this MAN_BASIS.			50.0 2001-11- 06T11:17:33 2002- 204T15:56:23Z	O
MAN_NEXT_EPOCH	Identifies the start time of the next maneuver for this MAN_BASIS.			50.0 2001-11- 06T11:17:33 2002- 204T15:56:23Z	O
MAN_PURPOSE	The user can specify the intention(s) of the maneuver. Multiple maneuver purposes can be provided as a comma-delimited list. While there is no CCSDS-based restriction on the value for this free-text keyword, it is suggested to use: Aerobraking (AEROBRAKE), Attitude adjust (ATTITUDE) Collision avoidance (COLA) Deployment (DEPLOY) Disposal (DISPOSAL) Gravity assist flyby (GRAV_ASSIST_FROM_XXXX, where XXXX=body center name, e.g. SANA Registry [11]) Inclination adjustment (INCLINATION) Launch & Early Orbit (LEOP) Maneuver cleanup (MNVR_CLEANUP) Mass adjust (MASS_ADJUST) Momentum desaturation (MOM_DESAT) Orbit adjust (ORBIT) Orbit trim (TRIM) Other (OTHER) Period adjustment (PERIOD) Pointing Request Message (PRM_ID_XXXX) Relocation (RELOCATION) Science objective (SCI_OBJ) Spin rate adjust (SPIN_RATE) Station-keeping (SK) Trajectory correction (TRAJ_CORR)			DISPOSAL	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
MAN_PRED_SOURCE	For future maneuvers, specifies the source of the orbit and/or attitude state(s) upon which the maneuver is based. While there is no CCSDS-based restriction on the value for this free-text keyword, it is suggested to consider using ORB_ID and OD_ID keywords as described in Tables 6-4 and 6-11 respectively, or a combination thereof.			OD_5	O
MAN_REF_FRAME	Reference frame in which all maneuver vector direction data is provided in this maneuver data block. Select from the accepted set of values indicated in ANNEX B, Section B4. The reference frame must be the same for all data elements within a given maneuver time history block.		TNW_IN ERTIAL	J2000	O
MAN_FRAME_EPOCH	Epoch of the maneuver data reference frame, if not intrinsic to the definition of the reference frame. See 7.5.12 for formatting rules.			2001-11- 06T11:17:33 2002- 204T15:56:23Z	C
GRAV_ASSIST_NAME	Origin of maneuver gravitational assist body, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter. See ANNEX B, Section B2, for acceptable GRAV_ASSIST_NAME values (and the procedure to propose new values).		EARTH	EARTH MOON EROS JUPITER	O
DC_TYPE	Duty cycle type to use for this maneuver time history section: <ul style="list-style-type: none"> CONTINUOUS denotes full/continuous thrust <default>; TIME denotes a time-based duty cycle driven by time past a reference time and the duty cycle ON and OFF durations; TIME_AND_ANGLE denotes a duty cycle driven by the phasing/clocking of a space object body frame "trigger" direction past a reference direction. 		CONTIN UOUS	CONTINUOUS TIME TIME_AND_ANG LE	O
DC_WIN_OPEN	Start time of the duty cycle-based maneuver window that occurs on or prior to the actual maneuver execution start time. For example, this may identify the time at which the satellite is first placed into a special duty-cycle-based maneuver mode. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".			50.0 2001-11- 06T11:17:33 2002- 204T15:56:23Z	C
DC_WIN_CLOSE	End time of the duty cycle-based maneuver window that occurs on or after the actual maneuver execution end time. For example, this may identify the time at which the satellite is taken out of a special duty-cycle-based maneuver mode. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".			100.0 2001-11- 07T51:17:33 2002- 204T15:58:03Z	C
DC_MIN_CYCLES	Minimum number of "ON" duty cycles (may override DC_EXEC_STOP). This value is optional even if DC_TYPE ≠ "CONTINUOUS".		0	5	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_MAX_CYCLES	Maximum number of "ON" duty cycles (may override DC_EXEC_STOP). This value is optional even if DC_TYPE ≠ "CONTINUOUS".		0	200	O
DC_EXEC_START	Start time of the initial duty cycle-based maneuver sequence execution. DC_EXEC_START is defined to occur on or prior to the first maneuver "ON" portion within the duty cycle sequence. DC_EXEC_START must occur coincident with or after DC_WIN_OPEN. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".			50.0 2001-11-06T11:17:33 2002-204T15:56:23Z	C
DC_EXEC_STOP	End time of the final duty cycle-based maneuver sequence execution. DC_EXEC_STOP typically occurs on or after the end of the final maneuver "ON" portion within the duty cycle sequence. DC_EXEC_STOP occurs prior to or coincident with DC_WIN_CLOSE. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".			100.0 2001-11-07T51:17:33 2002-204T15:58:03Z	C
DC_REF_TIME	Reference time for the THRUST duty cycle, specified as either time in seconds (relative to EPOCH_TZERO), or as an absolute "<epoch>" (see Section 7.5.12 for formatting rules). Note: Depending upon EPOCH_TZERO, DC_REF_TIME relative times may be negative. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".			8000.0 2001-11-06T11:17:33	C
DC_TIME_PULSE_DURATION	Thruster pulse "ON" duration, initiated at first satisfaction of the burn "ON" time constraint or upon completion of the previous DC_TIME_PULSE_PERIOD cycle. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".	s		10.0	C
DC_TIME_PULSE_PERIOD	Elapsed time between the start of one pulse and the start of the next. Must be greater than or equal to DC_TIME_PULSE_DURATION. This keyword shall be set if DC_TYPE ≠ "CONTINUOUS".	s		200.0	C

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_REF_DIR	<p>For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE), specifies the reference unit vector direction in the “MAN_REF_FRAME” reference frame at which, when mapped into the space object’s spin plane (normal to the spin axis) the duty cycle is triggered (see DC_PA_START_ANGLE for phasing).</p> <p>This (tripartite, or three-element vector) value shall be provided if DC_TYPE = “TIME_AND_ANGLE”.</p> <p>This reference direction does <u>NOT</u> represent the duty cycle midpoint.</p>			1.0 0.0 0.0	C
DC_BODY_FRAME	<p>For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE), specifies the body reference frame in which DC_BODY_TRIGGER will be specified. Select from the accepted set of values indicated in ANNEX B, Section B6. This keyword shall be set if DC_TYPE = “TIME_AND_ANGLE”.</p>			SC_BODY_1 SENSOR_3	C
DC_BODY_TRIGGER	<p>For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE), specifies the body frame reference unit vector direction in the “DC_BODY_FRAME” reference frame at which, when its projection onto the spin plane crosses the corresponding projection of DC_REF_DIR onto the spin plane, this angle-based duty cycle is initiated (see DC_PA_START_ANGLE for phasing).</p> <p>This tripartite value shall be provided if DC_TYPE = “TIME_AND_ANGLE”.</p>			0.707 0.0 0.707	C
DC_PA_START_ANGLE	<p>For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE), specifies the phase angle offset of thruster pulse start, measured with respect to the occurrence of a DC_BODY_TRIGGER crossing of the DC_REF_DIR direction when both are projected into the spin plane (normal to the body spin axis). This phase angle offset can be positive or negative to allow the duty cycle to begin prior to the next crossing of the DC_REF_DIR. As this angular direction is to be used in a modulo sense, there is no requirement for the magnitude of the phase angle offset to be less than 360.</p> <p>This keyword shall be set if DC_TYPE = “TIME_AND_ANGLE”.</p>	deg		25.0	C

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
DC_PA_STOP_ANGLE	For phase angle thruster duty cycles (DC_TYPE=TIME_AND_ANGLE), specifies the phase angle of thruster pulse stop, measured with respect to the DC_BODY_TRIGGER crossing of the DC_REF_DIR direction when both are projected into the spin plane. This phase angle offset can be positive or negative to allow the duty cycle to end after to the next crossing of the DC_REF_DIR. As this angular direction is to be used in a modulo sense, there is no requirement for the magnitude of the phase angle offset to be less than 360. This keyword shall be set if DC_TYPE = "TIME_AND_ANGLE".	deg		35.0	C
MAN_COMPOSITION	The comma-delimited ordered set of maneuver elements of information contained on every maneuver time history line, with values selected from Table 6-8 or Table 6-9. Note that this composition must include one or both of TIME_ABSOLUTE and TIME_RELATIVE.			TIME_RELATIVE, THR_X, THR_Y, THR_Z, THR_ISP, THR_EFFIC, DELTA_MASS, DV_X, DV_Y, DV_Z, DV_SIGMA	M
MAN_UNITS	A comma-delimited set of SI unit designations for each and every element of the maneuver time history following the maneuver time tag(s) solely for informational purposes, provided as an alphanumeric free-text string enclosed in square brackets. All elements comprising the MAN_COMPOSITION must have a corresponding units entry; percentages shall be denoted by "%" and control switches, non-dimensional values and text strings shall be labelled as "n/a". When provided, each maneuver data element selected via MAN_COMPOSITION shall have a corresponding units entry, with percentages denoted by "%" and control switches, non-dimensional values and text strings denoted by "n/a". Note that the listing of units via the MAN_UNITS keyword does not override the mandatory units for the selected MAN_COMPOSITION, as specified in Table 6-8 or Table 6-9.			[N,N,N,s,%kg,m/s,m/s,m/s,%,n/a]	O
... < Insert maneuver data here >	<Maneuver time history data, content, and units shall be provided as specified by MAN_COMPOSITION.>				M
MAN_STOP	End maneuver data block.				M

Table 6-8: OCM Data: Selectable propulsive (i.e., non-deployment) maneuver fields in the maneuver time history data

Keyword	Description	Units	Examples of Values
TIME_ABSOLUTE	Absolute epoch time as formatted in Section 7.5.10.	n/a	2018-11-13T11:13:20.5Z
TIME_RELATIVE	Relative epoch time measured in seconds with respect to the epoch time specified via the EPOCH_TZERO keyword.	s	20157.26
MAN_DURA	The maneuver duration associated with this impulsive ΔV , thrust, and/or acceleration-imparted event.	s	200.0
DELTA_MASS	Mass change (where a negative number denotes a mass decrement/loss to the host) associated with this time interval. For “thrust” specification, this mass change shall include the mass change prescribed by the rocket equation.	kg	-5.0
ACC_X	Acceleration component A_x in the selected maneuver frame.	km/s** 2	0.000734092785
ACC_Y	Acceleration component A_y in the selected maneuver frame.	km/s** 2	0.000189779834
ACC_Z	Acceleration component A_z in the selected maneuver frame.	km/s** 2	0.0000794872502
ACC_INTERP	Acceleration vector Euler axis/angle interpolation mode between current and next acceleration line.	n/a	OFF ON
ACC_SIGMA	One-sigma percent error on acceleration magnitude.	%	1.0
DV_X	Velocity increment ΔV_x in the selected maneuver reference frame. The actual ΔV should be impulsively applied at a time of <time tag> + $\frac{1}{2}$ (MAN_DURA).	km/s	0.025
DV_Y	Velocity increment ΔV_y in the selected maneuver reference frame. The actual ΔV should be impulsively applied at a time of <time tag> + $\frac{1}{2}$ (MAN_DURA).	km/s	0.0015
DV_Z	Velocity increment ΔV_z in the selected maneuver reference frame. The actual ΔV should be impulsively applied at a time of <time tag> + $\frac{1}{2}$ (MAN_DURA).	km/s	0.00029
DV_SIGMA	One-sigma percent error on ΔV magnitude.	%	2.0
THR_X	Thrust component T_x measured in the selected maneuver reference frame.	N	1.0
THR_Y	Thrust component T_y measured in the selected maneuver reference frame.	N	2.0
THR_Z	Thrust component T_z measured in the selected maneuver reference frame.	N	3.0
THR_EFFIC	Thrust efficiency “ η ,” typically ranging between 0.0 and 1.0.	n/a	0.95
THR_INTERP	Thrust vector Euler axis/angle interpolation mode between current and next thrust line; values shall be selected as either “OFF” or “ON”.	n/a	OFF ON
THR_ISP	Thrust specific impulse.	s	330.0
THR_SIGMA	One-sigma percent error on thrust magnitude.	%	2.0

Table 6-9: OCM Data: Selectable deployment fields in the maneuver time history data

Keyword	Description	Units	Examples of Values
TIME_ABSOLUTE	Absolute epoch time as formatted in Section 7.5.10.	s	2018-11-13T11:13:20.5Z
TIME_RELATIVE	Relative epoch time measured in seconds with respect to the epoch time specified via the EPOCH_TZERO keyword.	s	20157.26
DEPLOY_ID	Free-text identifier of the resulting “child” object deployed from this host during TIME_SPAN. Setting DEPLOY_ID to zero “0” indicates that a deployment did not occur at this time tag.	n/a	CubeSat_001
DEPLOY_DV_X	Velocity increment ΔV_x of the deployed “child” object measured in the selected maneuver reference frame, applied instantaneously at the time tag of deployment.	km/s	0.0001
DEPLOY_DV_Y	Velocity increment ΔV_y of the deployed “child” object measured in the selected maneuver reference frame, applied instantaneously at the time tag of deployment.	km/s	0.00003
DEPLOY_DV_Z	Velocity increment ΔV_z of the deployed “child” object measured in the selected maneuver reference frame, applied instantaneously at the time tag of deployment.	km/s	0.00002
DEPLOY_MASS	Decrement in host mass as a result of deployment (shall be ≤ 0.0).	kg	-1.0
DEPLOY_DV_SIGMA	One-sigma percent error on deployment ΔV magnitude.	%	5.0
DEPLOY_DV_RATIO	Ratio of child-to-host ΔV vectors, such that: $\overline{\Delta v}_{host} = \text{DEPLOY_DV_RATIO} \times \overline{\Delta v}_{child}$ <p>NOTE: As an opposite ΔV is typically imparted to the host during deployment, this number is typically less than or equal to zero. This ratio allows the user to specify how much ΔV is imparted to the host vehicle. This is usually not -1.0 (i.e. an equal-and-opposite imparted velocity), to account for the mass fraction between the child and the host as well as any rotational torque acted on the host as a result of deployment direction centerline offsets as compared to the host’s center of gravity.</p>	n/a	-0.05
DEPLOY_DV_CDA	Typical (50 th percentile) product of drag coefficient (Cd) times cross-sectional area for the deployed “child” object.	m**2	0.022

6.2.8 OCM DATA: PERTURBATIONS SPECIFICATION

6.2.8.1 Table 6-10 provides an overview of the OCM Perturbations Specification section. Only those keywords shown in Table 6-10 shall be used in OCM perturbations specification.

6.2.8.2 Only one OCM Perturbations Specification section shall appear in an OCM.

6.2.8.3 The OCM Perturbations Specification section shall be delimited by two keywords: PERT_START and PERT_STOP.

Table 6-10: OCM Data: Perturbations Specification

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
PERT_START	Start of the perturbations specification	n/a			M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Perturbations Specification only immediately after the PERT_START key word; see Section 7.8 for comment formatting rules).	n/a		This is a comment	O
ATMOSPHERIC_MODEL	Name of atmosphere model, which shall be selected from the accepted set of values indicated in ANNEX B, Section B9.	n/a		MSISE90 NRLMSIS00 J70 J71 JROBERTS DTM JB2008	O
GRAVITY_MODEL	The gravity model (selected from the accepted set of gravity model names indicated in ANNEX B, Section B10), followed by the degree (D) and order (O) of the applied spherical harmonic coefficients used in the simulation. NOTE: Specifying a zero value for “order” (e.g., 2D 00) denotes zonals (J2 ... JD)	n/a		EGM-96: 36D 36O WGS-84: 8D 0O GGM-01: 36D 36O TEG-4: 36D 36O	O
EQUATORIAL_RADIUS	Oblate spheroid equatorial radius of the central body.	km		6378.137	O
GM	Gravitational coefficient of attracting body (Gravitational Constant x Central Mass).	km**3 /s**2		3.986004e5	O
N_BODY_PERTURBATIONS	One OR MORE (N-body) gravitational perturbations bodies used. Values, listed serially in comma-delimited fashion, denote a natural solar or extra-solar system body (stars, planets, asteroids, comets, and natural satellites). Note that only those entries (or those procedurally added to the CENTER_NAME content as specified in ANNEX B) that are denoted as an “Attracting Body” in ANNEX B, Section B2 are acceptable values.	n/a		MOON, SUN, JUPITER	O
CENTRAL_BODY_ROTATION	Central body angular rotation rate, measured about the major principal axis of the inertia tensor of the central body, relating inertial and central-body-fixed reference frames. Note that the rotation axis may be slightly offset from the inertial frame Z-axis definition.	deg/s		4.17807421629e-3	O
OBLATE_FLATTENING	Central body’s oblate spheroid oblateness for the polar-symmetric oblate central body model (e.g., for the Earth, it is approximately 1.0/298.257223563).	n/a		0.00335281066475	O
OCEAN_TIDES_MODEL	Name of ocean tides model (optionally specify order or constituent effects, diurnal, semi-diurnal, etc.).	n/a		DIURNAL SEMI-DIURNAL	O
SOLID_TIDES_MODEL	Name of solid tides model (optionally specify order or constituent effects, diurnal, semi-diurnal, etc.).	n/a		DIURNAL SEMI-DIURNAL	O
REDUCTION_THEORY	Specification of the reduction theory used for precession and nutation modeling. This is a free text field, so if the examples on the right are insufficient, others may be used.			IAU1976/FK5 IAU2010 IERS1996	O
ALBEDO_MODEL	Name of the albedo model.			STK	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
ALBEDO_GRID_SIZE	Number of grid points used in the albedo model.			100	O
SHADOW_MODEL	Shadow model used for Solar Radiation Pressure; dual cone uses both umbra/penumbra regions. Selected option should be one of "NONE", "CYLINDRICAL", "CONE", or "DUAL CONE".			NONE CYLINDRICAL DUAL CONE	O
SHADOW_BODIES	Comma-separated list of planetary bodies for which SRP shadowing is modeled, selected from ANNEX B for CENTER_NAME values.		EARTH	EARTH MOON	O
SRP_MODEL	Name of SRP model. This is a free text field, so if the examples on the right are insufficient, others may be used.			GPS_ROCK BOX_WING CANNONBALL COD ...	O
SW_DATA_SOURCE	Free text field specifying the source and version of the Space Weather data used in the creation of this message. Multiple space weather sources can be specified in a comma-delimited fashion.			CELESTRAK	O
SW_DATA_EPOCH	Epoch of the Space Weather data.			2001-11-08T00:00:00	O
SW_INTERP_METHOD	Free text field specifying the method used to select or interpolate any and all sequential space weather data (K_p , a_p , Dst, $F_{10.7}$, $M_{10.7}$, $S_{10.7}$, $Y_{10.7}$, etc.). While not constrained to specific entries, it is anticipated that the utilized method would match methods detailed in numerical analysis textbooks.			PRECEDING_VALUE NEAREST_NEIGHBOR LINEAR LAGRANGE_ORDER_5	O
FIXED_GEOMAG_KP	A fixed (time invariant) value of the planetary 3-hour-range geomagnetic index K_p used to override the normal time-varying 3-hourly K_p values (e.g., obtained from SW_DATA_SOURCE).	nT		3.2	O
FIXED_GEOMAG_AP	A fixed (time invariant) value of the 3-hourly (equivalent range) geomagnetic index a_p used to override the normal time-varying 3-hourly a_p values (e.g., obtained from SW_DATA_SOURCE).	nT		21	O
FIXED_GEOMAG_DST	A fixed (time invariant) value of the planetary 1-hour-range geomagnetic index Dst used to override the normal time-varying daily Dst values (e.g., obtained from SW_DATA_SOURCE).	nT		-20	O
FIXED_F10P7	A fixed (time invariant) value of the solar flux units (SFU) daily proxy $F_{10.7}$ used to override the normal time-varying daily $F_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_F10P7_MEAN	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy $F_{10.7}$ used to override the normal time-varying averaged $F_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		132.0	O
FIXED_M10P7	A fixed (time invariant) value of the solar flux daily proxy $M_{10.7}$ used to override the normal time-varying daily $M_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_M10P7_MEAN	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy $M_{10.7}$ used to override the normal time-varying averaged $M_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
FIXED_S10P7	A fixed (time invariant) value of the solar flux daily proxy $S_{10.7}$ used to override the normal time-varying daily $S_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_S10P7_MEAN	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy $S_{10.7}$ used to override the normal time-varying averaged $S_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_Y10P7	A fixed (time invariant) value of the solar flux daily proxy $Y_{10.7}$ used to override the normal time-varying daily $Y_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
FIXED_Y10P7_MEAN	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy $Y_{10.7}$ used to override the normal time-varying averaged $Y_{10.7}$ values (e.g., obtained from SW_DATA_SOURCE).	SFU		120.0	O
PERT_STOP	End of the perturbations specification.				M

6.2.9 OCM DATA: ORBIT DETERMINATION DATA

6.2.9.1 Table 6-11 provides an overview of the OCM orbit determination data section. Only those keywords shown in Table 6-11 shall be used in OCM orbit determination data specification.

6.2.9.2 At most, only one Orbit Determination Data section shall appear in an OCM.

6.2.9.3 Orbit determination data in the OCM shall be indicated by two keywords: OD_START and OD_STOP.

6.2.9.4 The values of the DAYS_SINCE_FIRST_OBS, and DAYS_SINCE_LAST_OBS keywords shall be specified as relative time, in days, to the value of the OD_EPOCH keyword.

6.2.9.5 If an orbit determination parameters section is included in the message, a corresponding perturbations section shall be included as well to specify the perturbations incorporated in the orbit determination.

6.2.9.6 The Orbit Determination parameters section shall reflect the orbit determination settings used to generate all OCM orbit, covariance and maneuver time history data sections that are based upon “determined” orbit solutions and contain the ORB_BASIS_ID, COV_BASIS_ID and MAN_BASIS_ID keyword values matching the OD_ID value of the OD block, respectively.

Table 6-11: OCM Data: Orbit Determination Data

Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OD_START	Start of an orbit determination data section			n/a	M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided in the OCM Orbit Determination Data section only immediately after the OD_START key word; see Section 7.8 for comment formatting rules).			This is a comment	O
OD_ID	Identification number for this orbit determination.			OD_20160402	M
OD_PREV_ID	Optional identification number for the previous orbit determination. Note: if this orbit determination is the first one, then OD_PREV_ID should be excluded from this message.			OD_20160401	O
OD_METHOD	Type of orbit determination method used to produce the orbit estimate. While this is a free-text field, it is suggested that it be comprised of the method, followed by a colon delimiter and the actual OD tool used to estimate the orbit (e.g., BAHN, ODIN, ODTK. NOTE: Commonly-used methods include Batch Weighted Least Squares (BWLS), Extended Kalman Filter (EKF), Sequential Filter (SF), Square Root Information Filter (SRIF), Sequential Simultaneous Estimation Method (SSEM)).			BWLS: BAHN BWLS: ODIN SF: ODTK	M

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
OD_EPOCH	Relative or absolute time tag of the orbit determination solved-for state in the selected OCM time system specified by the TIME_SYSTEM keyword.			2001-11-06T11:17:33 27854.239	M
DAYS_SINCE_FIRST_OBS	Days elapsed between first accepted observation and OD_EPOCH (NOTE: may be positive or negative).	d		3.5	O
DAYS_SINCE_LAST_OBS	Days elapsed between last accepted observation and OD_EPOCH (NOTE: may be positive or negative).	d		1.2	O
RECOMMENDED_OD_SPAN	Number of days of observations recommended for the OD of the object (<i>useful only for Batch OD systems</i>).	d		5.2	O
ACTUAL_OD_SPAN	Actual time span in days used for the OD of the object (NOTE: should equal (DAYS_SINCE_FIRST_OBS - DAYS_SINCE_LAST_OBS)).	d		2.3	O
OBS_AVAILABLE	The number of observations available within the actual OD time span.			100	O
OBS_USED	The number of observations accepted within the actual OD time span.			90	O
TRACKS_AVAILABLE	The number of sensor tracks available for the OD within the actual time span (see definition of “tracks”, Section 1.5.3).			33	O
TRACKS_USED	The number of sensor tracks accepted for the OD within the actual time span (see definition of “tracks”, Section 1.5.3).			30	O
MAXIMUM_OBS_GAP	The maximum time between observations in the OD of the object.	d		1.0	O
OD_EPOCH_EIGMAJ	Positional error ellipsoid 1σ major eigenvalue at the epoch of the OD.	m		58.73	O
OD_EPOCH_EIGMED	Positional error ellipsoid 1σ intermediate eigenvalue at the epoch of the OD.	m		35.7	O
OD_EPOCH_EIGMIN	Positional error ellipsoid 1σ minor eigenvalue at the epoch of the OD.	m		21.5	O
OD_MAX_PRED_EIGMAJ	The resulting maximum predicted major eigenvalue of the 1σ positional error ellipsoid over the entire TIME_SPAN of the OCM, stemming from this OD.	m		21.5	O
OD_MIN_PRED_EIGMIN	The resulting minimum predicted minor eigenvalue of the 1σ positional error ellipsoid over the entire TIME_SPAN of the OCM, stemming from this OD.	m		21.5	O
OD_CONFIDENCE	OD confidence metric, which by definition spans 0 to 100%. (useful only for Filter-based OD systems). The OD confidence metric shall be as mutually defined by message exchange participants.	%		95.3	O
GDOP	Generalized Dilution Of Precision for this orbit determination, based on the observability grammian as defined in ANNEX N, References [N-15] and [N-16] and expressed in Informative Annex C4. GDOP provides a rating metric of the observability of the element set from the OD. Alternate GDOP formations may be used as mutually defined by message exchange participants.			.857	O
SOLVE_N	The number of solve-for states in the orbit determination.			6	O
SOLVE_STATES	Free-text comma-delimited description of the state elements solved for in the orbit determination.			POS[3], VEL[3]	O
CONSIDER_N	The number of consider parameters used in the orbit determination.			3	O
CONSIDER_PARAMS	Free-text comma-delimited description of the consider parameters used in the orbit determination.			DRAG, SRP	O
SENSORS_N	The number of sensors used in the orbit determination.			3	O

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Keyword	Description	Units	Default (if any)	Examples of Values	M/O/C
SENSORS	Free-text comma-delimited description of the sensors used in the orbit determination.			EGLIN, FYLINGDALES	O
WEIGHTED_RMS	<p>(Useful / valid only for Batch OD systems).</p> <p>The weighted RMS residual ratio, defined as:</p> $\text{Weighted RMS} = \sqrt{\frac{\sum_{i=1}^N w_i (y_i - \hat{y}_i)^2}{N}}$ <p>Where y_i is the observation measurement at the ith time</p> <p>\hat{y}_i is the current estimate of y_i,</p> <p>$w_i = \frac{1}{\sigma_i^2}$ is the weight (sigma) associated with the measurement at the ith time and N is the number of observations.</p> <p>This is a value that can generally identify the quality of the most recent vector update, and is used by the analyst in evaluating the OD process. A value of 1.00 is ideal.</p>	(measurement units)		1.3	O
DATA_TYPES	Comma-separated list of observation data types utilized in this orbit determination. Although this is a free-text field, it is recommended at a minimum to use data type descriptor(s) as provided in Table 3-5 of the TDM standard [9] (excluding the DATA_START, DATA_STOP, and COMMENT keywords). Additional descriptors/detail is encouraged if the descriptors of Table 3-5 are not sufficiently clear, e.g., could replace ANGLE_1 and ANGLE_2 with RADEC (e.g., from a telescope), AZEL (e.g., from a ground radar), RANGE (whether from radar or laser ranging), etc.			ANGLE_1, ANGLE_2	O
OD_STOP	End of an orbit determination data section.			n/a	M

6.2.10 OCM DATA: USER-DEFINED PARAMETERS

6.2.10.1 A single section of User-Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User-Defined Parameters, if included, should be used as sparingly as possible; their use is not encouraged.

6.2.10.2 At most, only one User-Defined Parameters section shall appear in an OCM.

6.2.10.3 Table 6-12 provides an overview of the OCM user-defined data section. Only those keywords shown in Table 6-12 shall be used in OCM user-defined data specification.

Table 6-12: OCM Data: User-Defined Parameters

Keyword	Description	Units	Examples of Values	M/O/C
USER_START	Start of a User-Defined Parameters data block.			M
COMMENT	Comments (a contiguous set of one or more comment lines may be provided immediately following the USER_START keyword (see Section 7.8 for formatting rules.)).		This is a comment	O
USER_DEFINED_x	User-defined parameter, where 'x' is replaced by a variable length user specified character string. Any number of user-defined parameters may be included, if necessary to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED EARTH MODEL = WGS-84		USER_DEFINED_EARTH_MODEL = WGS-84	M
USER_STOP	End of a User-Defined Parameters data block.			M

7 ORBIT DATA MESSAGE SYNTAX

7.1 OVERVIEW

This section details the syntax requirements for each of the Orbit Data Messages.

7.2 GENERAL

The Orbit Data Messages (OPM, OMM, OEM, and OCM) shall observe the syntax described in 7.3 through 7.8.

7.3 ODM LINES

7.3.1 Each OPM, OMM, OEM, or OCM line shall be one of the following:

- Header line;
- Metadata line;
- Data line;
- Comment line; or
- Blank line.

7.3.2 Each OPM, OMM, or OEM line must not exceed 254 ASCII characters and spaces (excluding line termination character[s]).

7.3.3 OCM lines may be of arbitrary length. If exchange between the two parties requires a maximum line length, that limit should be mutually agreed between message exchange partners.

7.3.4 Only printable ASCII characters and blanks shall be used. Control characters (such as TAB, etc.) shall not be used, with the exception of the line termination characters specified below.

7.3.5 Blank lines may be used at any position within the file. Blank lines shall have no assignable meaning, and may be ignored.

7.3.6 The first header line must be the first non-blank line in the file.

7.3.7 All lines shall be terminated by a single Carriage Return or a single Line Feed, a Carriage Return/Line Feed pair, or a Line Feed/Carriage Return pair.

7.4 ORBIT DATA MESSAGES IN “KEYWORD = VALUE NOTATION” (I.E., NON-XML) AND ORDER OF ASSIGNMENT STATEMENTS

7.4.1 For the OPM and OMM, all header, metadata, and data lines shall use ‘keyword = value’ notation, abbreviated as KVN.

7.4.1.1 For the OEM, all header and metadata elements shall use KVN notation.

7.4.1.2 OEM ephemeris data lines shall not use KVN format; rather, the OEM ephemeris data line has a fixed structure containing seven required fields (epoch time, three position components, three velocity components), and three optional acceleration components. (see Section 5.2.4.)

7.4.1.3 OEM covariance matrix epoch and covariance reference frame (if used) shall use KVN format. The OEM covariance data lines shall not use KVN format; rather, the OEM covariance data line has a fixed structure containing from one to six required fields (a row from the 6x6 lower triangular form covariance matrix). (see Section 5.2.5.)

7.4.1.4 For the OCM, all header and metadata elements shall use KVN notation.

7.4.1.5 OCM orbit state time history data lines shall not use KVN format; rather, the structure of such OCM orbit state time history data shall be comprised of a contiguous set of lines containing time tag followed by the parameters corresponding to the selected orbit set (see Section 6.2.4).

7.4.1.6 OCM covariance matrix epoch and covariance reference frame (if used) shall use KVN format. The OCM covariance data lines shall not use KVN format; rather, OCM covariance time history data shall be comprised of a contiguous set of lines, with each line corresponding to a single time, time and covariance entries separated by at least one white space character, and the data line containing the time tag followed by either the Lower Triangular Matrix (LTM) content (e.g., [1,1], [2,1], [2,2], [3,1], ...) of the covariance matrix corresponding to the selected covariance type, or followed by the one-sigma standard deviation along the major, intermediate, and minor, axes and the corresponding major, intermediate, and minor eigenvectors (see Section 6.2.6.).

7.4.1.7 OCM maneuver data lines shall not use KVN format; rather, OCM maneuver data shall be comprised of a contiguous set of lines containing a fixed structure as presented in Sec. 6.2.7.

7.4.2 The keywords ‘COMMENT’, *‘_START’ and *‘_STOP’ are exceptions to the KVN syntax assignment.

7.4.3 Only a single ‘keyword = value’ assignment shall be made on a line.

7.4.4 Keywords must be uppercase and must not contain blanks.

7.4.5 Any white space immediately preceding or following the keyword shall not be significant.

7.4.6 Any white space immediately preceding or following the ‘equals’ sign shall not be significant.

7.4.7 Any white space immediately preceding the end of line shall not be significant.

7.4.8 The order of occurrence of mandatory and optional KVN assignments shall be fixed as shown in the tables in Sections 3, 4, 5, and 6 that describe the OPM, OMM, OEM, and OCM keywords.

7.5 VALUES

7.5.1 A non-empty value field must be assigned to each mandatory keyword except for *‘_START’ and *‘_STOP’ keyword values.

7.5.2 Comments and free-text value fields **may be in any case (or mix of case)** desired by the user.

7.5.3 Apart from comments and free-text fields, normative text value fields shall be constructed using only **exclusively all uppercase** or **exclusively all lowercase**.

7.5.4 Integer values shall consist of a sequence of decimal digits with an optional leading sign (+’ or -’). If the sign is omitted, +’ shall be assumed. Leading zeroes may be used. The range of values that may be expressed as an integer is:

$$-2,147,483,648 \leq x \leq +2,147,483,647 \quad (\text{i.e., } -2^{31} \leq x \leq 2^{31}-1).$$

NOTE – The commas in the range of values above are thousands separators and are used only for readability. They should not appear in an actual message.

7.5.5 Non-integer numeric values may be expressed in either fixed-point or floating-point notation. Both representations may be used within an OPM, OMM, OEM, or OCM.

7.5.6 Non-integer numeric values expressed in fixed-point notation shall consist of a sequence of decimal digits separated by a period as a decimal point indicator, with an optional leading sign (+’ or -’). If the sign is omitted, +’ shall be assumed. Leading and trailing zeroes may be used. At least one digit shall appear before and after a decimal point.

7.5.7 Non-integer numeric values may be expressed in either fixed-point or floating-point notation. Both representations may be used within an OPM, OMM, OEM or OCM.

7.5.8 Non-integer numeric values expressed in fixed-point notation shall consist of a sequence of decimal digits separated by a period as a decimal point indicator, with an optional leading sign (+’ or -’). If the sign is omitted, +’ shall be assumed. Leading and trailing zeroes may be used. At least one digit shall appear before and after a decimal point.

7.5.9 Non-integer numeric values expressed in floating point notation shall consist of a sign, a mantissa, an alphabetic character indicating the division between the mantissa and exponent, and an exponent, constructed according to the following rules:

- a) The sign may be '+' or '-'. If the sign is omitted, '+' shall be assumed.
- b) The mantissa must be a string of no more than 16 decimal digits with a decimal point ('.') in the second position of the ASCII string, separating the integer portion of the mantissa from the fractional part of the mantissa.
- c) The character used to denote exponentiation shall be 'E' or 'e'. If the character indicating the exponent and the following exponent are omitted, an exponent value of zero shall be assumed (essentially yielding a fixed point value).
- d) The exponent must be an integer, and may have either a '+' or '-' sign (if the sign is omitted, then '+' shall be assumed).
- e) The maximum positive floating-point value is approximately 1.798E+308, with 16 significant decimal digits precision. The minimum positive floating point value is approximately 4.94E-324, with 16 significant decimal digits precision.

7.5.10 Blanks shall not be permitted within numeric values and time strings.

7.5.11 In value fields that are text, an underscore shall be equivalent to a single blank. Individual blanks shall be retained (shall be significant), but multiple contiguous blanks shall be equivalent to a single blank.

7.5.12 In value fields that represent an absolute time tag or epoch, times shall be given in one of the following two formats:

YYYY-MM-DDThh:mm:ss[.d→d][Z]

or

YYYY-DDDThh:mm:ss[.d→d][Z]

where 'YYYY' is the year, 'MM' is the two-digit month, 'DD' is the two-digit day, 'DDD' is the three-digit day of year, 'T' is constant, 'hh:mm:ss[.d→d]' is the time in hours, minutes, seconds, and optional fractional seconds; 'Z' is an optional time code terminator (the only permitted value is 'Z' for Zulu, i.e., UTC). As many 'd' characters to the right of the period as required may be used to obtain the required precision, up to the maximum allowed for a fixed-point number. All fields shall have leading zeros. (see Section reference 2, ASCII Time Code A or B.).

NOTE: During a leap second introduction, the value of the two-digit integer seconds (ss) field shall be '60' as specified in reference [2].

7.5.13 The time system for CREATION_DATE is UTC; for all other keywords representing times or epochs, the time system is determined by the TIME_SYSTEM metadata keyword.

7.6 OCM VECTOR DATA TYPE

A number of OCM keywords may be set to values containing more than one number. Examples include DC_REF_DIR and DC_BODY_TRIGGER. Such vectors shall be space-delimited and provided serially on a single line following the equals “=” sign, adhering to the requirements for “numeric values” provided in the previous sections.

7.7 UNITS IN THE ORBIT DATA MESSAGES

7.7.1 OPM/OMM UNITS

7.7.1.1 For documentation purposes and clarity only, units may be included as ASCII text after a value in the OPM and OMM. If units are displayed, they must exactly match the units (including lower/upper case) as specified in tables 3-3, 4-3. If units are displayed, then:

- a) there must be at least one blank character between the value and the units text;
- b) the units must be enclosed within square brackets (e.g., ‘[km]’);
- c) multiplication shall be denoted by a single asterisk (e.g., ‘[N*m]’);
- d) division shall be denoted by a single forward slash (e.g., meters per second is m/s);
- e) exponents of units shall be denoted with a double asterisk (i.e., ‘**’, for example, $\text{km}^2=\text{km}^{**2}$);
- f) The usual order of operations ordering applies (e.g., exponents before multiplication).

7.7.1.2 Some of the items in the applicable tables are dimensionless. The table shows a unit value of ‘n/a’, which in this case means that there is no applicable units designator for these items (e.g., for ECCENTRICITY).

7.7.1.3 The notation ‘[n/a]’ shall not appear in an OPM or OMM.

7.7.2 OEM UNITS

7.7.2.1 In an OEM ephemeris data line, units shall be km, km/s, and km/s^{**2} for position, velocity, and acceleration components, respectively, but the units shall not be displayed.

7.7.2.2 In an OEM covariance matrix line, units shall be km^{**2} , km^{**2}/s , or $\text{km}^{**2}/\text{s}^{**2}$ depending on whether the element is computed from two position components, one position component and one velocity component, or two velocity components. The units shall not be displayed.

7.7.3 OCM UNITS

7.7.4 Units for OCM metadata keywords and OCM data block metadata sub-section keywords shall match the units specified in Table 6-3 through Table 6-12.

7.7.5 The units of orbit time state history data lines, where present, shall adhere to the specified units for orbit states as provided in the SANA Registry [11] for Orbital Elements (Annex B7).

7.7.6 The units of covariance time history data lines, where present, shall adhere to the specified units for covariance data as provided in the SANA Registry [11] for Orbital Elements (Annex B7) and Additional Covariance Representations (Annex B8).

7.7.7 The units of maneuver time history data lines, where present, shall adhere to the specified units for maneuver lines as provided in Table 6-8 and Table 6-9.

7.7.8 Apart from orbit state, covariance and maneuver data lines, units of OCM keyword values shall correspond to the normative “Units” column of the accompanying Keyword Value Tables (i.e., Table 6-3 through Table 6-12) for each section definition.

NOTE: The units used throughout the OCM are generally a combination of kilometers for distance and seconds for time (e.g., km/s for velocity, km/s**2 for acceleration, and so forth). Mass is in kilograms, and force is in Netwons).

7.7.8.1 For OCM keywords used to convey multipartite orbit state, covariance, or maneuver data lines, units may accompany these data lines via the ORB_UNITS, COV_UNITS and MAN_UNITS keywords, respectively. Units shall not be displayed in OCM orbit state, covariance, or maneuver data lines themselves.

7.7.8.2 For OCM keywords that are not used to convey multipartite orbit state, covariance, or maneuver data lines, units may be included as ASCII text after a value in the OCM for documentation purposes and clarity only. If units are displayed, they must exactly match the units (including lower/upper case) as specified in tables 6-3 through 6-12. If units are displayed, then:

- g) there must be at least one blank character between the value and the units text;
- h) the units must be enclosed within square brackets (e.g., '[m]');
- i) multiplication shall be denoted by a single asterisk (e.g., '[N*m]');
- j) division shall be denoted by a single forward slash (e.g., meters per second is m/s);
- k) exponents of units shall be denoted with a double asterisk (i.e., '**', for example, m²=m**2);
- l) The usual order of operations ordering applies (e.g., exponents before multiplication).

7.7.8.3 Some of the items in the applicable tables are dimensionless. The table shows a unit value of 'n/a', which in this case means that there is no applicable units designator for these items (e.g., for ECCENTRICITY) and no units displayed.

7.8 COMMENTS IN THE ORBIT DATA MESSAGES

7.8.1 There are certain pieces of information that provide clarity and remove ambiguity about the interpretation of the information in a file, yet are not standardized so as to fit cleanly into the 'keyword = value' paradigm. Rather than force the information to fit into a space limited to one line, the ODM producer should put further specifications and information into comments. Static information should be separately shared and/or mutually agreed between message exchange partners outside of the ODM.

7.8.2 Comments may be used to provide provenance information or to help describe dynamical events or other pertinent information associated with the data. This additional information is intended to aid in consistency checks and elaboration where needed, but shall not be required for successful processing of a file.

7.8.3 For the OPM, OMM, OEM, and OCM, comment lines shall be optional.

7.8.4 Comment text may be in any case (or mix of case) desired by the user.

7.8.5 All comment lines shall begin with the 'COMMENT' keyword followed by at least one space. This keyword must appear on every comment line, not just the first such line. The remainder of the line shall be the comment value. White space shall be retained (shall be significant) in comment values.

7.8.6 Placement of comments shall be as specified in the tables in Sections 3, 4, 5, and 6 that describe the OPM, OMM, OEM, and OCM keywords.

7.8.7 Comments in the OPM may appear in the OPM Header immediately after the 'CCSDS_OPM_VERS' keyword, at the very beginning of the OPM Metadata section, and at the beginning of a logical block in the OPM Data section. Comments must not appear between the components of any logical block in the OPM Data section.

NOTE – The logical blocks in the OPM Data section are indicated in Table 3-3.

7.8.8 Comments in the OMM may appear in the OMM Header immediately after the 'CCSDS_OMM_VERS' keyword, at the very beginning of the OMM Metadata section, and at the beginning of a logical block in the OMM Data section. Comments must not appear between the components of any logical block in the OMM Data section.

NOTE – The logical blocks in the OMM Data section are indicated in Table 4-3.

7.8.9 Comments in the OEM may appear in the OEM Header immediately after the 'CCSDS_OEM_VERS' keyword, at the very beginning of the OEM Metadata section (after the 'META_START' keyword), at the beginning of the OEM Ephemeris Data Section, and

at the beginning of the OEM Covariance Data section (after the ‘COV_START’ keyword). Comment lines must not appear within any block of ephemeris lines or covariance matrix lines.

7.8.10 Comments may appear in all logical blocks of the OCM, but only at the positions shown in the defining Tables (generally at the top of each section, following the *_START section delimiting keyword).

7.8.11 Extensive comments in an ODM are recommended in cases where that content is germane to the message and changes from message to message.

7.8.12 The following comments should be provided:

- a) Information regarding the genesis, history, interpretation, intended use, etc., of the state vector, spacecraft, maneuver, or ephemeris that may be of use to the receiver of the OPM, OMM, or OEM:

COMMENT Source: File created by JPL Multi-Mission Navigation Team as part
COMMENT of Launch Operations Readiness Test held on 20 April 2001.

- b) Natural body ephemeris information: When the Earth is not the center of motion, the ephemerides of the planets, satellites, asteroids, and/or comets (including associated constants) consistent with the ODM should be identified so that the recipient can, in a consistent manner, make computations involving other centers:

COMMENT Based on latest orbit solution which includes observations
COMMENT through 2000-May-15 relative to planetary ephemeris DE-0405.

- c) OEM accuracy vs. efficiency: If the covariance data section of the OEM is not utilized, the producer of an OEM should report in comment lines what the expected accuracy of the ephemeris is, so the user can smooth or otherwise compress the data without affecting the accuracy of the trajectory. The OEM producer also should strive to achieve not only the best accuracy possible, considering prediction errors, but also consider the efficiency of the trajectory representation (e.g., step sizes of fractional seconds between ephemeris lines may be necessary for precision scientific reconstruction of an orbit, but are excessive from the standpoint of antenna pointing predicts generation).

7.9 ORBIT DATA MESSAGE KEYWORDS

7.9.1 VERSION KEYWORDS

The Header of the OPM, OMM, OEM, and OCM shall provide a CCSDS Orbit Data Message version number that identifies the format version; this is included to anticipate future changes. The version keywords for the OPM, OMM, OEM, and OCM shall be CCSDS_OPM_VERS, CCSDS_OMM_VERS, CCSDS_OEM_VERS, and CCSDS_OCM_VERS, respectively. The

value shall have the form of ‘x.y’, where ‘y’ shall be incremented for corrections and minor changes, and ‘x’ shall be incremented for major changes. Version x.0 shall be reserved for versions accepted by the CCSDS as an official Recommended Standard (‘Blue Book’). Testing shall be conducted using OPM, OMM, OEM, and OCM version numbers less than 1.0 (e.g., 0.x). The specific OPM, OMM, OEM, and OCM version numbers to be used should be mutually agreed between message exchange partners. The following version numbers are supported (Blue Book) or have been supported in the past (Silver Book):

Version Keyword	Version Number	Applicable Recommendation
CCSDS_OPM_VERS	1.0	Silver Book 1.0, 09/2004
CCSDS_OPM_VERS	2.0	Silver Book 2.0, 09/2009
CCSDS_OPM_VERS	3.0	Blue Book 3.0 (this document)
CCSDS_OMM_VERS	2.0	Silver Book 2.0, 09/2009
CCSDS_OMM_VERS	3.0	Blue Book 3.0 (this document)
CCSDS_OEM_VERS	1.0	Silver Book 1.0, 09/2004
CCSDS_OEM_VERS	2.0	Silver Book 2.0, 09/2009
CCSDS_OEM_VERS	3.0	Blue Book 3.0 (this document)
CCSDS_OCM_VERS	3.0	Blue Book 3.0 (this document)

7.9.2 GENERAL KEYWORDS

7.9.2.1 Only those keywords shown in Table 3-1, Table 3-2, and Table 3-3 shall be used in an OPM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

7.9.2.2 Only those keywords shown in Table 4-1, Table 4-2, and Table 4-3 shall be used in an OMM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

7.9.2.3 Only those keywords shown in Table 5-2 and Table 5-3 shall be used in an OEM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

Only those keywords shown in tables Table 6-3, Table 6-4, Table 6-5, Table 6-6, Table 6-7, Table 6-8, Table 6-9, Table 6-10, Table 6-11, and Table 6-12 shall be used in an OCM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

7.10 VALIDATION AND INGEST OF KVN CONTENT VIA REGULAR EXPRESSIONS (OR “REGEX”)

7.10.1 BENEFITS OF USING REGULAR EXPRESSIONS WITH THE ODM

Unlike the schema validation feature native to XML versions of this message as described in Section 8, the KVN version of this message does not natively support such validations of KVN content. To accomplish validation and ingest of KVN versions of the ODM, the use of Regular Expressions (referred to as “Regex”) is strongly encouraged where possible. Most programming languages support the Regex feature, and Regex offers a detailed and rigorous way to ensure proper validation, interpretation and conformance to Orbit Data Message content.

7.10.2 SAMPLE REGULAR EXPRESSIONS FOR CCSDS MESSAGES

To facilitate the use of Regular Expressions when processing CCSDS Navigation Messages, Informative Annex C6 provides sample Regex patterns to rigorously match a variety of common KVN sequences.

8 CONSTRUCTING AN ODM/XML INSTANCE

8.1 OVERVIEW

This section provides detailed instructions for the user on how to create an Extensible Markup Language message (XML, reference [5]) based on one of the KVN-formatted messages described in Section 3, Section 4, Section 5, and Section 6. The section applies only to the XML representation of ODMs.

Overall information on using XML for Navigation Data Messages is provided in [5]. The ODM/XML schemas are available on the SANA Web site. SANA is the registrar for the protocol registries created under CCSDS. The ODM/XML schemas explicitly define the permitted data elements and values acceptable for the XML versions of the ODMs. The location of the ODM/XML schemas is:

OPM: <https://sanaregistry.org/r/ndmxml/ndmxml-1.0-opm-3.0.xsd>

OMM: <https://sanaregistry.org/r/ndmxml/ndmxml-1.0-omm-3.0.xsd>

OEM: <https://sanaregistry.org/r/ndmxml/ndmxml-1.0-oem-3.0.xsd>

OCM: <https://sanaregistry.org/r/ndmxml/ndmxml-1.0-ocm-3.0.xsd>

Figure 8-1 illustrates the basic structure of an ODM/XML instance. Defined structural elements are the header and body. The body then consists of one or more segments depending on the message type (one for the OPM, OMM, and OCM; one or more for the OEM). Each segment consists of a <metadata>/<data> pair. In an OEM, which could have more than one segment, the metadata/data pair is repeated in each segment.

```

<header>
</header>
<body>
    <segment>
        <metadata>
        </metadata>
        <data>
        </data>
    </segment>
</body>

```

Figure 8-1: ODM/XML Basic Structure

ODM/XML tags for keywords defined in in Section 3, Section 4, Section 5, and Section 6 appear just as in the KVN, i.e., all capital letters. Tags related to the XML message structure (i.e., that do not correspond directly to a KVN keyword) appear in ‘lowerCamelCase’ (e.g., <header>, <segment>, <metadata>, <stateVector>, <covarianceMatrix>, etc.).

8.2 XML VERSION

This section describes the eXtensible Markup Language (or XML) version of the Orbit Data Message. The first line of each XML instantiation shall specify the XML version:

```
<?xml version="1.0" encoding="UTF-8"?>
```

This line must appear on the first line of each instantiation, exactly as shown.

8.3 BEGINNING THE INSTANTIATION: ROOT ELEMENT TAG

8.3.1 Each instantiation shall have a ‘root element tag’ that identifies the message type and other information such as where to find the applicable schema, required attributes, etc.

8.3.2 The root element tag in an ODM/XML instantiation shall be one of those listed in Table 8-1.

Table 8-1: ODM/XML Root Element Tags

Root Element Tag	Message Type
<opm></opm>	Orbit Parameter Message
<omm></omm>	Orbit Mean Elements Message
<oem></oem>	Orbit Ephemeris Message
<ocm></ocm>	Orbit Comprehensive Message

8.3.3 The XML Schema Instance namespace attribute must appear in the root element tag of all ODM/XML instantiations, exactly as shown:

```
xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
```

If it is desired to validate an instantiation against the CCSDS Web-based schema, the `xsi:noNamespaceSchemaLocation` attribute must be coded as a single string of non-blank characters, with no line breaks exactly as shown:

```
xsi:noNamespaceSchemaLocation="http://sanaregistry.org/r/ndmxml/ndmxml-1.0-master.xsd"
```

NOTE – The value associated with the `xsi:noNamespaceSchemaLocation` attribute shown in this document is too long to appear on a single line.

8.3.4 For use in a local operations environment, the schema set may be downloaded from the SANA website to a local server that meets local requirements for operations robustness.

8.3.5 If a local version is used, the value associated with the `xsi:noNamespaceSchemaLocation` attribute must be changed to a URL that is accessible to the local server.

8.3.6 Two attributes shall appear in the root element tag of an ODM/XML single message instantiation, specifically, the `CCSDS_XXX_VERS` keyword that is also part of the standard KVN header, and the Blue Book version number. The final attributes of the root element tag shall be 'id' and 'version'.

8.3.7 The `CCSDS_XXX_VERS` keyword shall be supplied via the 'id' attribute of the root element tag. The 'id' attribute shall be `'id="CCSDS_XXX_VERS"'`, where xxx = OPM, OMM, OEM, or OCM.

8.3.8 The version number of the Blue Book to which the schema applies shall be supplied via the 'version' attribute. The 'version' attribute shall be `'version="3.0"'`.

NOTE – The following example root element tag for an OPM instantiation combines all the directions in the preceding several Sections:

```
<?xml version="1.0" encoding="UTF-8"?>
<opm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
  xsi:noNamespaceSchemaLocation="http://sanaregistry.org/r/nd
    mxml/ndmxml-1.0-master.xsd"
    id="CCSDS_OPM_VERS" version="3.0">
```

8.4 THE STANDARD ODM/XML HEADER SECTION

8.4.1 The ODMs shall share a standard header section, with tags `<header>` and `</header>`.

8.4.2 Immediately following the `<header>` tag the message may have any number of `<COMMENT>` elements.

8.4.3 The standard ODM header shall contain the `<CREATION_DATE>` and the `<ORIGINATOR>` elements.

8.4.4 The standard ODM header may contain the `<MESSAGE_ID>` element.

NOTE – An example `<header>` section is shown immediately below.

```
<header>
  <COMMENT>This is the common ODM/XML header</COMMENT>
  <COMMENT>I can put as many comments here as I want,
    </COMMENT>
  <COMMENT>including none. </COMMENT>
  <CREATION_DATE>2004-281T17:26:06</CREATION_DATE>
  <ORIGINATOR>AGENCY-X</ORIGINATOR>
  <MESSAGE_ID>XYZ123-2019</MESSAGE_ID>
```

</header>

8.5 THE ODM/XML BODY SECTION

8.5.1 After coding the <header>, the instantiation must include a <body> section.

8.5.2 Inside the <body> section must appear at least one <segment> section.

8.5.3 Each segment must be made up of one or more <metadata> and <data> sections, depending on the specific message type.

8.6 THE ODM/XML METADATA SECTION

8.6.1 All ODMs must have a metadata section.

8.6.2 The metadata section shall be delimited by the <metadata> element.

8.6.3 Between the <metadata> and </metadata> tags, the keywords shall be the same as those in the metadata sections in Section 3, Section 4, Section 5, and Section 6, with possible exceptions as noted in the sections below that discuss creating instantiations of the specific messages.

8.7 THE ODM/XML DATA SECTION

8.7.1 All ODMs must have a data section.

8.7.2 The data section shall follow the metadata section and shall be delimited by the <data> element.

8.7.3 Between the <data> and </data> tags, the keywords shall be the same as those in the data sections in Section 3, Section 4, Section 5, and Section 6, with possible exceptions as noted in the Sections that discuss creating instantiations of the specific messages.

8.8 CREATING AN OPM INSTANTIATION

8.8.1 An OPM instantiation shall be delimited with the <opm></opm> root element tags using the standard attributes documented in **8.3**.

8.8.2 The final attributes of the <opm> tag shall be 'id' and 'version'.

8.8.3 The 'id' attribute shall be 'id="CCSDS_OPM_VERS"'.

8.8.4 The 'version' attribute shall be 'version="3.0"'.

8.8.5 The standard NDM header shall follow the <opm> tag (see Section **8.4**).

8.8.6 The OPM <body> shall consist of a single <segment>.

8.8.7 The <segment> shall consist of a <metadata> section and a <data> section.

8.8.8 The keywords in the <metadata> and <data> sections shall be those specified in Section 3. The rules for including any of the keyword tags in the OPM/XML are the same as those specified for the OPM/KVN.

8.8.9 Tags for keywords specified in Section 3 shall be all uppercase.

8.8.10 Several of the OPM/XML keywords may have the units attribute.

8.8.11 In all cases, the units shall match those defined in the tables in Section 3.

8.8.12 The following table lists examples of the use of units in the OPM/XML.

Keyword	Units	Example
INCLINATION	deg	<INCLINATION units="deg">numeric-value</INCLINATION>
MASS	kg	<MASS units="kg">numeric-value</MASS>
X	km	<X units="km">numeric-value</X>
CX_X	km**2	<CX_X units="km**2">numeric-value</CX_X>
CX_DOT_X	km**2/s	<CX_DOT_X units="km**2/s">numeric-value</CX_DOT_X>
CX_DOT_X_DOT,	km**2/s**2	<CX_DOT_X_DOT units="km**2/s**2">numeric-value</CX_DOT_X_DOT>
GM	km**3/s**2	<GM units="km**3/s**2">numeric-value</GM>
X_DOT	km/s	<X_DOT units="km/s">numeric-value</X_DOT>
MAN_DV_1	km/s	<MAN_DV_1 units="km/s">numeric-value</MAN_DV_1>
SOLAR_RAD_AREA	m**2	<SOLAR_RAD_AREA units="m**2">numeric-value</SOLAR_RAD_AREA>
MAN_DURATION	s	<MAN_DURATION units="s">numeric-value</MAN_DURATION>

8.8.13 In addition to the OPM keywords specified in Section 3, there are several special tags associated with the OPM body as described in the next few Sections. The information content in the OPM is separated into “logical blocks.” Special tags in the OPM are used to encapsulate the information in the logical blocks of the OPM.

8.8.14 The OPM/XML tags used to delimit the logical blocks of the OPM shall be drawn from the following table:

OPM Logical Block	Associated ODM/XML OPM Tag
State Vector	<stateVector>
Keplerian Elements	<keplerianElements>
Spacecraft Parameters	<spacecraftParameters>
Covariance Matrix	<covarianceMatrix>
Maneuver Parameters	<maneuvers>
User-Defined Parameters	<userDefinedParameters>

8.8.15 Between the begin tag and end tag (i.e., between <spacecraftParameters> and </spacecraftParameters>), the user shall place the keywords required by the Spacecraft Parameters logical block as specified in Table 3-3 of Section 3.

8.9 CREATING AN OMM INSTANTIATION

8.9.1 An OMM instantiation shall be delimited with the `<omm></omm>` root element tags using the standard attributes documented in **8.3**.

8.9.2 The final attributes of the `<omm>` tag shall be 'id' and 'version'.

8.9.3 The 'id' attribute shall be 'id="CCSDS_OMM_VERS"'.

8.9.4 The 'version' attribute for the version of the OMM described in Section 4 shall be 'version="3.0"'.

8.9.5 The standard NDM header shall follow the `<omm>` tag (see Section **8.4**).

8.9.6 The OMM `<body>` shall consist of a single `<segment>`.

8.9.7 The `<segment>` shall consist of a `<metadata>` section and a `<data>` section.

8.9.8 The keywords in the `<metadata>` and `<data>` sections shall be those specified in Section 4. The rules for including any of the keyword tags in the OMM/XML are the same as those specified for the OMM/KVN in Section 4.

8.9.9 Tags for keywords specified in Section 4 shall be all uppercase.

8.9.10 Several of the OMM/XML keywords may have the unit attribute.

8.9.11 In all cases, the units shall match those defined in the tables in Section 4.

8.9.12 The following table lists examples of the use of units in the OMM/XML.

Keyword	Units	Example
BSTAR	1/ER	<code><BSTAR units="1/ER">numeric-value</BSTAR></code>
INCLINATION	deg	<code><INCLINATION units="deg">numeric-value</INCLINATION></code>
MASS	kg	<code><MASS units="kg">numeric-value</MASS></code>
SEMI_MAJOR_AXIS	km	<code><SEMI_MAJOR_AXIS units="km">numeric-value</SEMI_MAJOR_AXIS></code>
CX X	km**2	<code><CX X units="km**2">numeric-value</CX X></code>
CX_DOT_X	km**2/s	<code><CX_DOT_X units="km**2/s">numeric-value</CX_DOT_X></code>
CX_DOT_X_DOT	km**2/s**2	<code><CX_DOT_X_DOT units="km**2/s**2">numeric-value</CX_DOT_X_DOT></code>
GM	km**3/s**2	<code><GM units="km**3/s**2">numeric-value</GM></code>
SOLAR_RAD_AREA	m**2	<code><SOLAR_RAD_AREA units="m**2">numeric-value</SOLAR_RAD_AREA></code>
MEAN_MOTION	rev/day	<code><MEAN_MOTION units="rev/day">numeric-value</MEAN_MOTION></code>
MEAN_MOTION_DOT	rev/day**2	<code><MEAN_MOTION_DOT units="rev/day**2">numeric-value</MEAN_MOTION_DOT></code>

8.9.13 In addition to the OMM keywords specified in Section 4, there are several special tags associated with the OMM body as described in the next few sections. The information content

in the OMM is separated into constructs described in Section 4 as ‘logical blocks’. Special tags in the OMM are used to encapsulate the information in the logical blocks of the OMM.

8.9.14 The OMM/XML tags used to delimit the logical blocks of the OMM shall be drawn from the following table:

OMM Logical Block	Associated ODM/XML OMM Tag
Mean Keplerian Elements	<meanElements>
Spacecraft Parameters	<spacecraftParameters>
TLE Parameters	<tleParameters>
Covariance Matrix	<covarianceMatrix>
User-Defined Parameters	<userDefinedParameters>

8.9.15 Between the begin tag and end tag (i.e., between <spacecraftParameters> and </spacecraftParameters>), the user must place the keywords required by the Spacecraft Parameters logical block as specified in Table 4-3 of Section 4.

8.10 CREATING AN OEM INSTANTIATION

8.10.1 An OEM instantiation shall be delimited with the `<oem></oem>` root element tags using the standard attributes documented in 8.3.

8.10.2 The final attributes of the `<oem>` tag shall be 'id' and 'version'.

8.10.3 The 'id' attribute shall be 'id="CCSDS_OEM_VERS"'.

8.10.4 The 'version' attribute for the version of the OEM described in Section 5 shall be 'version="3.0"'.

8.10.5 The standard NDM header shall follow the `<oem>` tag (see Section 8.4).

8.10.6 The OEM `<body>` shall consist of one or more `<segment>` constructs.

8.10.7 Each `<segment>` shall consist of a `<metadata>` section and a `<data>` section.

8.10.8 The keywords in the `<metadata>` and `<data>` sections shall be those specified in Section 5. The rules for including any of the keyword tags in the OEM/XML are the same as those specified for the OEM/KVN in Section 5.

8.10.9 Tags for keywords specified in Section 5 shall be all uppercase.

8.10.10 Several of the OEM/XML keywords may have the unit attribute.

8.10.11 In all cases, the units shall match those defined in Section 5.

8.10.12 In addition to the OEM keywords specified in Section 5, there are some special tags associated with the OEM body as described in the next Sections.

8.10.13 The `<stateVector>` tag shall encapsulate the keywords associated with one of the ephemeris data lines in the OEM.

8.10.14 In the XML representation of the OEM, the components of the `<stateVector>` ephemeris data line must be represented with keywords (i.e., a tag).

8.10.15 The `<stateVector>` keywords shall be the same as those defined for the same construct in the OPM.

8.10.16 The OEM/XML tags used within the `<stateVector>` structure shall be drawn from the following table:

OEM Tag	Represents	Example
<EPOCH>	time tag of the state	<EPOCH>2007-09-20T17:41:00</EPOCH>
<X>	x component of position	<X units="km">6678.0</X>
<Y>	y component of position	<Y units="km">0.0</Y>
<Z>	z component of position	<Z units="km">0.0</Z>
<X_DOT>	x component of velocity	<X_DOT units="km/s">0</X_DOT>
<Y_DOT>	y component of velocity	<Y_DOT units="km/s">7.73</Y_DOT>
<Z_DOT>	z component of velocity	<Z_DOT units="km/s">0.0</Z_DOT>
<X_DDOT>	x component of acceleration	<X_DDOT units="km/s**2">0.0</X_DDOT>
<Y_DDOT>	y component of acceleration	<Y_DDOT units="km/s**2">0.50</Y_DDOT>
<Z_DDOT>	z component of acceleration	<Z_DDOT units="km/s**2">0.0</Z_DDOT>

8.10.17 Between the begin tag and end tag (i.e., between <stateVector> and </stateVector>), the user shall place the values required by the ephemeris data line as specified in Section 5.

8.10.18 Since the state vector structure is shared by the OPM schema and OEM schema, units may optionally appear in the XML version of the OEM ephemeris data line.

8.10.19 The <covarianceMatrix> tag shall encapsulate the keywords associated with the covariance matrix lines in the OEM.

8.10.20 In the XML representation of the OEM, the covariance data line must be represented with keywords (i.e., a tag).

8.10.21 The OEM <covarianceMatrix> keywords shall be the same as those defined for the same construct in the OPM and OMM.

NOTE – In the KVN representations of the OEM covariance matrix data lines, keywords are not used. Rather, the components of the covariance matrix data line appear in an order defined in Section 5. Similarly, units are not used in the KVN version of the OEM covariance matrix; however, they are optional in the OPM and OMM.

8.10.22 Since the covariance matrix structure is shared by the OPM, OMM, and OEM, units may optionally appear in the XML version of the OEM covariance matrix line.

8.10.23 The OEM/XML tags used within the <covarianceMatrix> structure shall be drawn from the following table:

Keyword	Units	Example
CX_X, CY_X, CY_Y, CZ_X, CZ_Y, CZ_Z	km**2	<CX_X units="km**2">numeric-value</CX_X>
CX_DOT_X, CX_DOT_Y, CX_DOT_Z, CY_DOT_X, CY_DOT_Y, CY_DOT_Z, CZ_DOT_X, CZ_DOT_Y, CZ_DOT_Z	km**2/s	<CX_DOT_X units="km**2/s">numeric-value</CX_DOT_X>
CX_DOT_X_DOT, CY_DOT_X_DOT, CY_DOT_Y_DOT, CZ_DOT_X_DOT, CZ_DOT_Y_DOT, CZ_DOT_Z_DOT	km**2/s**2	<CX_DOT_X_DOT units="km**2/s**2">numeric-value</CX_DOT_X_DOT>

8.10.24 Between the begin tag and end tag (i.e., between <covarianceMatrix> and </covarianceMatrix>), the user shall place the values required by the covariance matrix line type as specified in e.g., Table 5-4 of Section 5.

8.11 CREATING AN OCM INSTANTIATION

8.11.1 An OCM instantiation shall be delimited with the `<ocm></ocm>` root element tags using the standard attributes documented in 8.3.

8.11.2 The final attributes of the `<ocm>` tag shall be 'id' and 'version'.

8.11.3 The 'id' attribute shall be 'id="CCSDS_OCM_VERS"'.

8.11.4 The 'version' attribute for the version of the OCM described in Section 6 shall be 'version="3.0"'.

8.11.5 The standard NDM header shall follow the `<ocm>` tag (see Section 8.4).

8.11.6 The OCM `<body>` shall consist of a single `<segment>` construct.

8.11.7 The `<segment>` shall consist of a `<metadata>` section and a `<data>` section.

8.11.8 The keywords in the `<metadata>` and `<data>` sections shall be those specified in Section 6. The rules for including any of the keyword tags in the OCM/XML are the same as those specified for the OCM in Section 6.

8.11.9 Tags for keywords specified in Section 6 shall be all uppercase.

8.11.10 Several of the OCM/XML keywords may have the unit attribute.

8.11.11 In all cases, the units shall match those defined in the SANA Registry (detailed in ANNEX B) and tables in Section 6, including orbit, covariance and maneuver data units as specified by ORB_UNITS, COV_UNITS, and MAN_UNITS, respectively.

8.9.16 The following table lists examples of the use of units in the OCM/XML.

Keyword	Units	Example
ACTUAL_OD_SPAN	d	<code><ACTUAL_OD_SPAN units="d">numeric-value</ACTUAL_OD_SPAN></code>
DC_PA_START_ANGLE	deg	<code><DC_PA_START_ANGLE units="deg">numeric-value</DC_PA_START_ANGLE></code>
CENTRAL_BODY_ROTATION	deg/s	<code><CENTRAL_BODY_ROTATION units="deg/s">numeric-value</CENTRAL_BODY_ROTATION></code>
WET_MASS	kg	<code><MASS units="kg">numeric-value</WET_MASS></code>
IXX	kg*m**2	<code><IXX units="kg*m**2">numeric-value</IXX></code>
EQUATORIAL_RADIUS	km	<code><EQUATORIAL_RADIUS units="km">numeric-value</EQUATORIAL_RADIUS></code>
GM	km**3/s**2	<code><GM units="km**3/s**2">numeric-value</GM></code>
DV_BOL	km/s	<code><DV_BOL units="km/s">numeric-value</DV_BOL></code>
OEB_MAX	m	<code><OEB_MAX units="m">numeric-value</OEB_MAX></code>
DRAG_CONST_AREA	m**2	<code><DRAG_CONST_AREA units="m**2">numeric-value</DRAG_CONST_AREA></code>
MAX_THRUST	N	<code><MAX_THRUST units="N">numeric-value</MAX_THRUST></code>
FIXED_GEOMAG_KP	nT	<code><FIXED_GEOMAG_KP units="nT">numeric-value</FIXED_GEOMAG_KP></code>

Keyword	Units	Example
UT1MUTC_AT_TZERO	s	<UT1MUTC_AT_TZERO units="s">numeric-value</UT1MUTC_AT_TZERO>
FIXED_F10P7	SFU	<FIXED_F10P7 units="SFU">numeric-value</FIXED_F10P7>
DRAG_UNCERTAINTY	%	<DRAG_UNCERTAINTY units="%">numeric-value</DRAG_UNCERTAINTY>

8.11.12 In addition to the OCM keywords specified in Section 6, there are some special tags associated with the OCM body as described in the next Sections.

OCM Logical Block	ODM/XML OCM Section Tags	Data Line Tag
Orbit Data	<orb>	<orbLine>
Space Object Physical Characteristics	<phys>	N/A
Covariance Data	<covar>	<covLine>
Maneuver Data	<man>	<manLine>
Perturbations Parameters	<pert>	N/A
Orbit Determination Data	<od>	N/A
User-Defined Parameters	<userDef>	N/A

8.11.13 Between the begin tag and end tag (e.g., between <orb> and </orb>), the user must place the keywords required by the specific OCM section as specified in e.g., Table 6-4 of Section 6.

8.11.14 The data type of the <orbLine>, <covLine>, and <manLine> elements is "xsd:string", i.e., there is no validation of the contents and the line must be parsed by the OCM recipient to access the individual components of the orbit, covariance, or maneuver data line.

8.11.15 The number of individual components in the multipartite <orbLine> shall be determined by the number of components in the value for the ORB_TYPE keyword, plus one for the timetag.

8.11.16 The number of individual components in the single multipartite <covLine> shall be 13 if COV_TYPE=SIG3EIGVEC3 is selected, or 14 if COV_TYPE=TSIG3EIGVEC3 is selected. Otherwise, if "N" is the dimension of the covariance matrix, then the number of individual components in the single multipartite <covLine> shall be $(N**2+N)/2$ corresponding to the Lower Triangular Matrix format, plus one for the timetag.

8.11.17 The number of individual components in the multipartite <manLine> shall be determined by the number of comma-separated values in the MAN_COMPOSITION keyword, plus one for the timetag.

8.12 CREATING A COMBINED INSTANTIATION

8.12.1 An ODM user may create an XML instance that incorporates any number of messages from Sections 3 through 6 of this document in a logical suite called an ‘NDM (Navigation Data Message) Combined Instantiation’. Such combined instantiations may be useful for some situations, for example:

- A maneuver scenario where both "no burn" ephemeris and "with burn" ephemeris are combined in a single message.
- A constellation scenario where states (OPM, OMM) and/or ephemeris data (OEM, OCM) for all the spacecraft in the constellation are combined in a single XML message.
- A full OEM or OCM ephemeris with detail on important states reflected in some number of OPMs. The OEM/OCM and the multiple OPMs can be conveniently conveyed in a single NDM.

8.12.2 An NDM combined instantiation shall be delimited with the `<ndm></ndm>` root element tags instead of one of the individual message tags described in **8.3.2**

8.12.3 The standard attributes documented in 8.3 shall be used with the `<ndm>` tag, with the exception that neither ‘id’ nor ‘version’ attributes are associated with the `<ndm>` tag.

8.12.4 In the NDM combined instantiation, the only attributes that shall appear on the constituent message tags (i.e., the tags listed in 8.3.2) are the ‘id’ and ‘version’ attributes.

8.12.5 Between the `<ndm></ndm>` tags, the desired messages described in **8.8** through **8.11** may be combined as needed to meet user requirements.

8.12.6 Any combination of constituent ODM types may be used in an NDM combined instantiation.

8.12.7 **Figure 8-1** and **Figure 8-2** illustrate the basic structure of an NDM combined instantiation. All detail has been removed from **Figure 8-1** in order to contrast the single message ODM with an NDM combined instantiation. As shown in **Figure 8-2**, in an NDM combined instantiation the individual message tags still have the ‘id’ and ‘version’ attributes, but the namespace attributes and schema location attributes are associated with the `<ndm>` root element.

Single Message OPM	NDM Combined Instantiation
<pre> <opm> <header> </header> <body> </body> </opm> </pre>	<pre> <ndm> <opm> <header> </header> <body> </body> </opm> . . . <opm> <header> </header> <body> </body> </opm> </ndm> </pre>

Figure 8-2: Comparison of Single Message OPM with NDM Combined Instantiation

8.12.8 The OPMs shown in Figure 8-2 may be replaced with any number of OMM, OEM, or OCM messages in any combination as needed to meet user requirements, as shown in Figure 8-3 below.

```

<?xml version="1.0" encoding="UTF-8"?>
<ndm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
      xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml/ndmxml-1.0-
master.xsd">

<COMMENT>This figure combines multiple ODM/XML messages into a single message</COMMENT>
<COMMENT>Message detail is deleted in order to focus on the message structure</COMMENT>
<COMMENT>Note use of "<ndm>" root element, and ODM/version attributes</COMMENT>

  <opm id="CCSDS_OPM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </opm>

  <oem id="CCSDS_OEM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </oem>

  <opm id="CCSDS_OPM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </opm>

  <omm id="CCSDS_OMM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </omm>

  <ocm id="CCSDS_OCM_VERS" version="3.0">
    <header>
    </header>
    <body>
    </body>
  </om>

</ndm>

```

Figure 8-3: NDM Combined Instantiation Showing Mix of ODMs and Use of Attributes

NOTE: See Fig. H-1 for a full example of one use case combining multiple ODMs in a single XML message. For instructions on creating a combined instantiation that incorporates ODM/XML messages combined with other navigation related messages, see reference [5].

8.13 SPECIAL SYNTAX RULES FOR ODM/XML

8.13.1 Most of the KVN syntax rules apply for ODM/XML instantiations of an ODM, however, there are a few variations described in this section that shall be observed.

8.13.2 Each mandatory XML tag must be present and contain a valid value.

8.13.3 Integer values shall follow the conventions of the integer data type per Reference [12]. Additional restrictions on the allowable range of values permitted for any integer data element may also be defined in the ODM/XML Schema.

NOTE – Examples of such restrictions may include a defined range (e.g., 0 - 100, 1 - 10, etc.), a set of enumerated values (e.g., 0,1,2,4,8), a pre-defined specific variation such as positiveInteger, or a user-defined data type variation.

8.13.4 Non-integer numeric values may be expressed in either fixed-point or floating-point notation. Numeric values shall follow the conventions of the double data type per Reference [13]. Additional restrictions on the allowable range of values permitted for any numeric data element may also be defined in the ODM/XML Schema.

NOTE – Examples of such restrictions may include a defined range (e.g., 0.0-100.0, etc.), or a user-defined data type variation.

8.13.5 Text values shall follow the conventions of the string data type per Reference [14]. Additional restrictions on the allowable range or values permitted for any data element may also be defined in the ODM/XML Schema.

NOTE – Examples of such restrictions may include a set of enumerated values (e.g., ‘YES’/‘NO’) or other user-defined data type variation.

8.13.6 The units in the ODM/XML shall be the same units used in the KVN-formatted ODM described in Section 7.7.3, or as mandated in the SANA registry per ANNEX B. XML attributes shall be used to explicitly define the units or other important information associated with the given data element. See the tables in this section for the OPM, OMM, OEM, and OCM for examples of coding units in ODM/XML instantiations.

8.13.7 Comments must be displayed as values between the <COMMENT> and </COMMENT> tags.

ANNEX A

IMPLEMENTATION CONFORMANCE**STATEMENT PRO FORMA****(NORMATIVE)****A1 INTRODUCTION****A1.1 OVERVIEW**

This annex provides the Implementation Conformance Statement (ICS) Requirements List (RL) for an implementation of the Orbit Data Messages (CCSDS 502.0). The ICS for an implementation is generated by completing the RL in accordance with the instructions below. An implementation shall satisfy the mandatory conformance requirements referenced in the RL.

- The RL in this annex is blank. An implementation's completed RL is called the ICS. The ICS states which capabilities and options have been implemented. The following can use the ICS:
 - the implementer, as a checklist to reduce the risk of failure to conform to the standard through oversight;
 - a supplier or potential acquirer of the implementation, as a detailed indication of the capabilities of the implementation, stated relative to the common basis for understanding provided by the standard ICS proforma;
 - a user or potential user of the implementation, as a basis for initially checking the possibility of interworking with another implementation (it should be noted that, while interworking can never be guaranteed, failure to interwork can often be predicted from incompatible ICS lists);
 - a tester, as the basis for selecting appropriate tests against which to assess the claim for conformance of the implementation.

A1.2 ABBREVIATIONS AND CONVENTIONS

The RL consists of information in tabular form. The status of features is indicated using the abbreviations and conventions described below.

Item Column

The item column contains sequential numbers for items in the table.

Feature Column

The feature column contains a brief descriptive name for a feature. It implicitly means “Is this feature supported by the implementation?”

Status Column

The status column uses the following notations:

- M mandatory;
- O optional;
- C conditional;
- X prohibited;
- I out of scope;
- N/A not applicable.

Support Column Symbols

The support column is to be used by the implementer to state whether a feature is supported by entering Y, N, or N/A, indicating:

- Y Yes, supported by the implementation.
- N No, not supported by the implementation.
- N/A Not applicable.

The support column should also be used, when appropriate, to enter values supported for a given capability.

A1.3 INSTRUCTIONS FOR COMPLETING THE RL

An implementer shows the extent of compliance to the Recommended Standard by completing the RL; that is, the state of compliance with all mandatory requirements and the options supported are shown. The resulting completed RL is called an ICS. The implementer shall complete the RL by entering appropriate responses in the support or values supported column, using the notation described in A1.2. If a conditional requirement is inapplicable, N/A should be used. If a mandatory requirement is not satisfied, exception information must be supplied by entering a reference X_i , where i is a unique identifier, to an accompanying rationale for the noncompliance.

A2 ICS PROFORMA FOR ORBIT DATA MESSAGES

A2.1 IDENTIFICATION OF ICS

Date of Statement (DD/MM/YYYY)	
ICS serial number	
System Conformance statement cross-reference	

A2.2 IDENTIFICATION OF IMPLEMENTATION UNDER TEST (IUT)

Implementation name	
Implementation version	
Special Configuration	
Other Information	

A2.3 IDENTIFICATION OF SUPPLIES

Supplier	
Contact Point for Queries	
Implementation Name(s) and Versions	
Other information necessary for full identification, e.g., name(s) and version(s) for machines and/or operating systems; System Name(s)	

A2.4 DOCUMENT VERSIONS

CCSDS 502.0 Document Version	3
<p>Have any exceptions been required?</p> <p>(Note: A YES answer means that the implementation does not conform to the Recommended Standard. Non-supported mandatory capabilities are to be identified in the ICS, with an explanation of why the implementation is non-conforming.)</p>	Yes _____ No_____

A2.5 REQUIREMENTS LISTS

[See CCSDS A20.1-Y-1, *CCSDS Implementation Conformance Statements* (Yellow Book, Issue 1, April 2014).]

A2.6 ORBIT PARAMETER MESSAGE REQUIREMENTS LIST

Note to Tom Gannett: these sections should probably be A2.5.1 etc. (Annex level three) but the line type is not defined and I'm not confident I know how to do that.

Note 2 to Tom Gannett: Would it be okay to highlight in BLUE or otherwise denote (with a bolder row-wise border) the separate data blocks as I've done below?

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OPM Header	N/A	Table 3-1	M	
2	OPM Version	CCSDS_OPM_VERS	Table 3-1	M	
3	Comment	COMMENT	Table 3-1	O	
4	Message creation date and time	CREATION_DATE	Table 3-1	M	
5	Message originator	ORIGINATOR	Table 3-1	M	
6	Unique message identifier	MESSAGE_ID	Table 3-1	O	
7	OPM Metadata	N/A	Table 3-2	M	
8	Comment	COMMENT	Table 3-2	O	
9	Name of space object	OBJECT_NAME	Table 3-2	M	
10	Identifier of space object	OBJECT_ID	Table 3-2	M	
11	Orbit center	CENTER_NAME	Table 3-2	M	
12	Reference frame	REF_FRAME	Table 3-2	M	
13	Epoch of reference frame	REF_FRAME_EPOCH	Table 3-2	C	
14	Time system applicable to data	TIME_SYSTEM	Table 3-2	M	
15	OPM Data	N/A	Table 3-3	M	
16	State Vector logical block	N/A	Table 3-3	M	
17	Comment	COMMENT	Table 3-3	O	
18	Epoch of the state vector	EPOCH	Table 3-3	M	
19	X component of position	X	Table 3-3	M	
20	Y component of position	Y	Table 3-3	M	
21	Z component of position	Z	Table 3-3	M	
22	X component of velocity	X_DOT	Table 3-3	M	
23	Y component of velocity	Y_DOT	Table 3-3	M	

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Item	Feature	Keyword	Reference	Status M/O/C	Support
24	Z component of velocity	Z_DOT	Table 3-3	M	
25	Keplerian Elements logical block	N/A	Table 3-3	O	
26	Comment	COMMENT	Table 3-3	O	
27	Semi-major axis of orbit	SEMI_MAJOR_AXIS	Table 3-3	C	
28	Eccentricity of orbit	ECCENTRICITY	Table 3-3	C	
29	Inclination of orbit	INCLINATION	Table 3-3	C	
30	Right ascension of ascending node of orbit	RA_OF_ASC_NODE	Table 3-3	C	
31	Argument of pericenter of orbit	ARG_OF_PERICENTER	Table 3-3	C	
32	True or mean anomaly of orbit	TRUE_ANOMALY or MEAN_ANOMALY	Table 3-3	C	
33	Gravitational coefficient of the central body	GM	Table 3-3	C	
34	Spacecraft Parameters logical block	N/A	Table 3-3	O	
35	Comment	COMMENT	Table 3-3	O	
36	Mass of the spacecraft	MASS	Table 3-3	C	
37	Solar radiation area of the spacecraft	SOLAR_RAD_AREA	Table 3-3	C	
38	Solar radiation coefficient of the spacecraft	SOLAR_RAD_COEFF	Table 3-3	C	
39	Drag area of the spacecraft	DRAG_AREA	Table 3-3	C	
40	Drag coefficient of the spacecraft	DRAG_COEFF	Table 3-3	C	
41	Position/velocity Covariance Matrix logical block	N/A	Table 3-3	O	
42	Comment	COMMENT	Table 3-3	O	
43	Cov reference frame	COV_REF_FRAME	Table 3-3	C	
44	Covariance matrix [1,1]	CX_X	Table 3-3	C	
45	Covariance matrix [2,1]	CY_X	Table 3-3	C	
46	Covariance matrix [2,2]	CY_Y	Table 3-3	C	
47	Covariance matrix [3,1]	CZ_X	Table 3-3	C	
48	Covariance matrix [3,2]	CZ_Y	Table 3-3	C	
49	Covariance matrix [3,3]	CZ_Z	Table 3-3	C	

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Item	Feature	Keyword	Reference	Status M/O/C	Support
50	Covariance matrix [4,1]	CX_DOT_X	Table 3-3	C	
51	Covariance matrix [4,2]	CX_DOT_Y	Table 3-3	C	
52	Covariance matrix [4,3]	CX_DOT_Z	Table 3-3	C	
53	Covariance matrix [4,4]	CX_DOT_X_DOT	Table 3-3	C	
54	Covariance matrix [5,1]	CY_DOT_X	Table 3-3	C	
55	Covariance matrix [5,2]	CY_DOT_Y	Table 3-3	C	
56	Covariance matrix [5,3]	CY_DOT_Z	Table 3-3	C	
57	Covariance matrix [5,4]	CY_DOT_X_DOT	Table 3-3	C	
58	Covariance matrix [5,5]	CY_DOT_Y_DOT	Table 3-3	C	
59	Covariance matrix [6,1]	CZ_DOT_X	Table 3-3	C	
60	Covariance matrix [6,2]	CZ_DOT_Y	Table 3-3	C	
61	Covariance matrix [6,3]	CZ_DOT_Z	Table 3-3	C	
62	Covariance matrix [6,4]	CZ_DOT_X_DOT	Table 3-3	C	
63	Covariance matrix [6,5]	CZ_DOT_Y_DOT	Table 3-3	C	
64	Covariance matrix [6,6]	CZ_DOT_Z_DOT	Table 3-3	C	
65	Maneuver Parameters logical block	N/A	Table 3-3	O	
66	Comment	COMMENT	Table 3-3	O	
67	Time of maneuver start	MAN_EPOCH_IGNITION	Table 3-3	O	
68	Duration of maneuver	MAN_DURATION	Table 3-3	O	
69	Change of mass due to maneuver	MAN_DELTA_MASS	Table 3-3	O	
70	Relevant reference frame for maneuver	MAN_REF_FRAME	Table 3-3	O	
71	1st component of velocity change	MAN_DV_1	Table 3-3	O	
72	2nd component of velocity change	MAN_DV_2	Table 3-3	O	
73	3rd component of velocity change	MAN_DV_3	Table 3-3	O	
74	User-Defined Parameters logical block	N/A	Table 3-3	O	
75	As defined by user, "essential	USER_DEFINED_x	Table 3-3	O	

Item	Feature	Keyword	Reference	Status M/O/C	Support
	information that cannot be conveyed in COMMENT statements"				

A2.7 ORBIT MEAN ELEMENTS MESSAGE REQUIREMENTS LIST

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OMM Header	N/A	Table 4-1	M	
2	OMM Version	CCSDS_OMM_VERS	Table 4-1	M	
3	Comment	COMMENT	Table 4-1	O	
4	Message creation date and time	CREATION_DATE	Table 4-1	M	
5	Message originator	ORIGINATOR	Table 4-1	M	
6	Unique message identifier	MESSAGE_ID	Table 4-1	O	
7	OMM Metadata	N/A	Table 4-2	M	
8	Comment	COMMENT	Table 4-2	O	
9	Name of space object	OBJECT_NAME	Table 4-2	M	
10	Identifier of space object	OBJECT_ID	Table 4-2	M	
11	Orbit center	CENTER_NAME	Table 4-2	M	
12	Reference frame	REF_FRAME	Table 4-2	M	
13	Epoch of reference frame	REF_FRAME_EPOCH	Table 4-2	C	
14	Time system applicable to data	TIME_SYSTEM	Table 4-2	M	
15	Mean element set theory of data	MEAN_ELEMENT_THEORY	Table 4-2	M	
16	OMM Data	N/A	Table 4-3	M	
17	Mean Keplerian elements logical block	N/A	Table 4-3	M	
18	Comment	COMMENT	Table 4-3	O	
19	Epoch of the state vector	EPOCH	Table 4-3	M	
20	Semi-major axis or mean motion	SEMI_MAJOR_AXIS or MEAN MOTION	Table 4-3	M	
21	Eccentricity	ECCENTRICITY	Table 4-3	M	
22	Inclination	INCLINATION	Table 4-3	M	
23	Right ascension of ascending node	RA_OF_ASC_NODE	Table 4-3	M	
24	Argument of pericenter	ARG_OF_PERICENTER	Table 4-3	M	
25	Mean anomaly	MEAN_ANOMALY	Table 4-3	M	

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Item	Feature	Keyword	Reference	Status M/O/C	Support
26	Gravitational Coefficient	GM	Table 4-3	O	
27	Spacecraft Parameters logical block	N/A	Table 4-3	O	
28	Comment	COMMENT	Table 4-3	O	
29	Spacecraft Mass	MASS	Table 4-3	O	
30	Solar Radiation Pressure Area	SOLAR_RAD_AREA	Table 4-3	O	
31	Solar Radiation Pressure Coefficient	SOLAR_RAD_COEFF	Table 4-3	O	
32	Drag Area	DRAG_AREA	Table 4-3	O	
33	Drag Coefficient	DRAG_COEFF	Table 4-3	O	
34	TLE logical block	N/A	Table 4-3	O	
35	Comment	COMMENT	Table 4-3	O	
36	Ephemeris Type	EPHEMERIS_TYPE	Table 4-3	O	
37	Classification Type	CLASSIFICATION_TYPE	Table 4-3	O	
38	NORAD Catalog Number	NORAD_CAT_ID	Table 4-3	O	
39	Element set number	ELEMENT_SET_NO	Table 4-3	O	
40	Revolution Number	REV_AT_EPOCH	Table 4-3	O	
41	SGP/SGP4 drag-like coefficient	BSTAR	Table 4-3	O	
42	First Time Derivative of the Mean Motion	MEAN_MOTION_DOT	Table 4-3	O	
43	Second Time Derivative of Mean Motion	MEAN_MOTION_DDOT	Table 4-3	O	
44	Pos/Vel/Cov logical block		Table 4-3	O	
45	Comment	COMMENT	Table 4-3	O	
46	Cov reference frame	COV_REF_FRAME	Table 4-3	C	
47	Covariance[1,1]	CX_X	Table 4-3	C	
48	Covariance[2,1]	CY_X	Table 4-3	C	
49	Covariance[2,2]	CY_Y	Table 4-3	C	
50	Covariance[3,1]	CZ_X	Table 4-3	C	
51	Covariance[3,2]	CZ_Y	Table 4-3	C	
52	Covariance[3,3]	CZ_Z	Table 4-3	C	
53	Covariance[4,1]	CX_DOT_X	Table 4-3	C	
54	Covariance[4,2]	CX_DOT_Y	Table 4-3	C	
55	Covariance[4,3]	CX_DOT_Z	Table 4-3	C	
56	Covariance[4,4]	CX_DOT_X_DOT	Table 4-3	C	
57	Covariance[5,1]	CY_DOT_X	Table 4-3	C	
58	Covariance[5,2]	CY_DOT_Y	Table 4-3	C	
59	Covariance[5,3]	CY_DOT_Z	Table 4-3	C	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
60	Covariance[5,4]	CY_DOT_X_DOT	Table 4-3	C	
61	Covariance[5,5]	CY_DOT_Y_DOT	Table 4-3	C	
62	Covariance[6,1]	CZ_DOT_X	Table 4-3	C	
63	Covariance[6,2]	CZ_DOT_Y	Table 4-3	C	
64	Covariance[6,3]	CZ_DOT_Z	Table 4-3	C	
65	Covariance[6,4]	CZ_DOT_X_DOT	Table 4-3	C	
66	Covariance[6,5]	CZ_DOT_Y_DOT	Table 4-3	C	
67	Covariance[6,6]	CZ_DOT_Z_DOT	Table 4-3	C	
68	User-Defined Parameters logical block	N/A	Table 4-3	O	
69	As defined by user, "essential information that cannot be conveyed in COMMENT statements"	USER_DEFINED_x	Table 4-3	O	

A2.8 ORBIT EPHEMERIS MESSAGE REQUIREMENTS LIST

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OEM Header	N/A	Table 5-2	M	
2	OEM Version	CCSDS_OEM_VERS	Table 5-2	M	
3	Comment	COMMENT	Table 5-2	O	
4	Message creation date and time	CREATION_DATE	Table 5-2	M	
5	Message originator	ORIGINATOR	Table 5-2	M	
6	Unique message identifier	MESSAGE_ID	Table 5-2	O	
7	Metadata logical block	N/A	Table 5-3	M	
8	Start of OEM Metadata	META_START	Table 5-3	M	
9	Comment	COMMENT	Table 5-3	O	
10	Name of space object	OBJECT_NAME	Table 5-3	M	
11	Identifier of space object	OBJECT_ID	Table 5-3	M	
12	Orbit center	CENTER_NAME	Table 5-3	M	
13	Reference frame	REF_FRAME	Table 5-3	M	
14	Epoch of reference frame	REF_FRAME_EPOCH	Table 5-3	C	
15	Time system applicable to data	TIME_SYSTEM	Table 5-3	M	
16	Start of TOTAL time span covered by data	START_TIME	Table 5-3	M	
17	Start of useable orbit data	USEABLE_START_TIME	Table 5-3	O	
18	End of useable orbit data	USEABLE_STOP_TIME	Table 5-3	O	
19	End of TOTAL time span covered by data	STOP_TIME	Table 5-3	M	
20	Recommended interpolation method	INTERPOLATION	Table 5-3	O	
21	Recommended interpolation degree	INTERPOLATION_DEGREE	Table 5-3	C	
22	End of OEM Metadata	META_STOP	Table 5-3	M	
23	Ephemeris lines	... <insert ephemeris data lines here.>	Table 5-3	M	
24	OEM Covariance logical block	N/A	Table 5-3	O	
25	Start of OEM Covariance logical block	COVARIANCE_START	Table 5-3	M	
26	Epoch of the navigation solution	EPOCH	Table 5-3	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	related to the covariance matrix				
27	Reference frame of the covariance matrix, if different from that of the states in the ephemeris	COV_REF_FRAME	Table 5-3	C	
28	Covariance lines	... <insert covariance matrices here>	Table 5-3	O	
29	End of OEM Covariance logical block	COVARIANCE_STOP	Table 5-3	M	

A2.9 ORBIT COMPREHENSIVE MESSAGE REQUIREMENTS LIST

OCM Header

Item	Feature	Keyword	Reference	Status M/O/C	Support
1	OCM Header	N/A	Table 6-2	M	
2	OCM Version	CCSDS_OCM_VERS	Table 6-2	M	
3	Comment	COMMENT	Table 6-2	O	
4	Message creation date and time	CREATION_DATE	Table 6-2	M	
5	Message originator	ORIGINATOR	Table 6-2	M	
6	Unique message identifier	MESSAGE_ID	Table 6-2	O	

OCM Metadata

Item	Feature	Keyword	Reference	Status M/O/C	Support
7	Metadata logical block	N/A	Table 6-3	M	
8	OCM Metadata Start	META_START	Table 6-3	M	
9	Comment	COMMENT	Table 6-3	O	
10	Message classification/caveats	CLASSIFICATION	Table 6-2	O	
11	Spacecraft name for the object	OBJECT_NAME	Table 6-3	O	
12	International designator for the object	INTERNATIONAL_DESIGNATOR	Table 6-3	O	
13	Satellite catalog source	CATALOG_NAME	Table 6-3	O	
14	Unique satellite identification designator	OBJECT_DESIGNATOR	Table 6-3	O	
15	Alternate name(s) of space object used by spacecraft operator, State Actors, commercial SSA providers and/or media.	ALTERNATE_NAMES	Table 6-3	O	
16	Message originator or programmatic Point-of-Contact	ORIGINATOR_POC	Table 6-3	O	
17	Contact position of the originator PoC	ORIGINATOR_POSITION	Table 6-3	O	
18	Originator PoC phone number	ORIGINATOR_PHONE	Table 6-3	O	
19	Originator PoC address	ORIGINATOR_ADDRESS	Table 6-3	O	
20	Creating agency or operator	TECH_ORG	Table 6-3	O	
21	Technical Point-of-Contact	TECH_POC	Table 6-3	O	
22	Contact position of the technical PoC	TECH_POSITION	Table 6-3	O	
23	Technical PoC phone number	TECH_PHONE	Table 6-3	O	
24	Technical PoC address	TECH_ADDRESS	Table 6-3	O	
25	ID that uniquely identifies the previous message from this message originator for this particular space object	PREVIOUS_MESSAGE_ID	Table 6-3	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
26	ID that uniquely identifies the next message from this message originator for this particular space object	NEXT_MESSAGE_ID	Table 6-3	O	
27	Unique identifier of linked Attitude Data Message(s)	ADM_MSG_LINK	Table 6-3	O	
28	Unique identifier of linked Conjunction Data Message(s)	CDM_MSG_LINK	Table 6-3	O	
29	Unique identifier of linked Pointing Request Message(s)	PRM_MSG_LINK	Table 6-3	O	
30	Unique identifier of linked Reentry Data Message(s)	RDM_MSG_LINK	Table 6-3	O	
31	Unique identifier of linked Tracking Data Message(s)	TDM_MSG_LINK	Table 6-3	O	
32	Operator of the space object	OPERATOR	Table 6-3	O	
33	Owner of the space object	OWNER	Table 6-3	O	
34	Country where the owner or responsible party is based	COUNTRY	Table 6-3	O	
35	Name of the constellation	CONSTELLATION	Table 6-3	O	
36	Type of object	OBJECT_TYPE	Table 6-3	O	
37	Time system for all absolute time stamps in this OCM including EPOCH_TZERO	TIME_SYSTEM	Table 6-3	O	
38	Default epoch to which all relative times are referenced	EPOCH_TZERO	Table 6-3	M	
39	Operational status of the space object	OPS_STATUS	Table 6-3	O	
40	Type of orbit	ORBIT_CATEGORY	Table 6-3	O	
41	Elements of information included in this message	OCM_DATA_ELEMENTS	Table 6-3	O	
42	Spacecraft clock epoch	SCLK_OFFSET_AT_EPOCH	Table 6-3	C	
43	Spacecraft clock rate	SCLK_SEC_PER_SI_SEC	Table 6-3	C	
44	Creation epoch of the previous message from this originator for this	PREVIOUS_MESSAGE_EPOCH	Table 6-3	O	

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Item	Feature	Keyword	Reference	Status M/O/ C	Support
	particular space object				
45	Anticipated (or actual) epoch of the next message from this originator for this particular space object	NEXT_MESSAGE_EPOCH	Table 6-3	O	
46	Time of the earliest data	START_TIME	Table 6-3	O	
47	Time of the latest data	STOP_TIME	Table 6-3	O	
48	Span of time that the OCM covers, measured in days	TIME_SPAN	Table 6-3	O	
49	Difference (TAI – UTC) in seconds	TAIMUTC_AT_TZERO	Table 6-3	O	
50	Difference (UT1 – UTC) in seconds	UT1MUTC_AT_TZERO	Table 6-3	O	
51	Source and version of the message originator's Earth Orientation Parameters (EOP)	EOP_SOURCE	Table 6-3	O	
52	Method used to select or interpolate sequential EOP data	INTERP_METHOD_EOP	Table 6-3	O	
53	Method used to select or interpolate sequential space weather data	INTERP_METHOD_SW	Table 6-3	O	
54	Source and version of the message originator's celestial body (e.g., Sun/Earth/Planetary) ephemeris data	CELESTIAL_SOURCE	Table 6-3	O	
55	Metadata Stop	META_STOP	Table 6-3	M	

OCM Data: Orbit State Time History

Item	Feature	Keyword	Reference	Status M/O/C	Support
	Orbit state time history logical block	N/A	Table 6-4	O	
	Orbit state time history start	ORB_START	Table 6-4	M	
	Comment	COMMENT	Table 6-4	O	
	Identification number for this orbit state time history block	ORB_ID	Table 6-4	O	
	Identification number for the previous orbit time history	ORB_PREV_ID	Table 6-4	O	
	Identification number for the next orbit time history	ORB_NEXT_ID	Table 6-4	O	
	Basis of this Orbit State time history data	ORB_BASIS	Table 6-4	O	
	Identification number for the orbit determination, navigation solution, or simulation	ORB_BASIS_ID	Table 6-4	O	
	Recommended interpolation method	INTERPOLATION	Table 6-4	O	
	Recommended interpolation degree	INTERPOLATION_DEGREE	Table 6-4	C	
	Orbit averaging technique used	ORB_AVERAGING	Table 6-4	O	
	Origin of the orbit reference frame	CENTER_NAME	Table 6-4	O	
	Reference frame of the orbit state time history	ORB_REF_FRAME	Table 6-4	O	
	Epoch of the orbit data reference frame	ORB_FRAME_EPOCH	Table 6-4	C	
	Start of useable orbit data	USEABLE_START_TIME	Table 6-4	O	
	End of useable orbit data	USEABLE_STOP_TIME	Table 6-4	O	
	Orbit element set type	ORB_TYPE	Table 6-4	O	
	Orbital element units for data elements that follow the time tag	ORB_UNITS	Table 6-4	O	
	OCM Orbit state time history	... <insert orbit state time history here>	Table 6-4	M	
	Orbit state time history end	ORB_STOP	Table 6-4	M	

OCM Data: Space Object Physical Characteristics

Item	Feature	Keyword	Reference	Status M/O/C	Support
	Space Object Physical Characteristics logical block	N/A	Table 6-5	O	
	Start of a Space Object Physical Characteristics specification	PHYS_START	Table 6-5	M	
	Comment	COMMENT	Table 6-5	O	
	Satellite manufacturer name	MANUFACTURER	Table 6-5	O	
	Satellite manufacturer's spacecraft bus model name	BUS_MODEL	Table 6-5	O	
	Space objects that this object is docked to	DOCKED_WITH	Table 6-5	O	
	Additional drag Area facing the relative wind vector beyond that represented by AREA_ALONG_OEB_<MAX, MED, MIN>	DRAG_CONST_AREA	Table 6-5	O	
	Drag coefficient	DRAG_COEFF_NOM	Table 6-5	O	
	Drag coeff. 1σ uncertainty	DRAG_UNCERTAINTY	Table 6-5	O	
	Space object total mass at beginning of life	INITIAL_MASS	Table 6-5	O	
	Space object wet mass	WET_MASS	Table 6-5	O	
	Space object dry mass	DRY_MASS	Table 6-5	O	
	Parent reference frame which maps to the Optimally Enclosing Box (OEB) frame	OEB_PARENT_FRAME	Table 6-5	O	
	Epoch of the OEB parent reference frame	OEB_PARENT_FRAME_EPOCH	Table 6-5	C	
	Quaternion q ₁	OEB_Q1	Table 6-5	O	
	Quaternion q ₂	OEB_Q2	Table 6-5	O	
	Quaternion q ₃	OEB_Q3	Table 6-5	O	
	Quaternion q _c	OEB_QC	Table 6-5	O	
	Maximum physical OEB dimension	OEB_MAX	Table 6-5	O	
	Medium physical OEB dimension	OEB_INT	Table 6-5	O	
	Minimum physical dimension	OEB_MIN	Table 6-5	O	
	Cross-sectional area viewed along max OEB dimension	AREA_ALONG_OEB_MAX	Table 6-5	O	
	Cross-sectional area viewed along intermediate OEB dimension	AREA_ALONG_OEB_INT	Table 6-5	O	
	Cross-sectional area viewed along min OEB dimension	AREA_ALONG_OEB_MIN	Table 6-5	O	
	Minimum cross-sectional area for collision probability	AREA_MIN_FOR_PC	Table 6-5	O	
	Maximum cross-sectional area for collision probability	AREA_MAX_FOR_PC	Table 6-5	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	Typical cross-sectional area for collision probability	AREA_TYP_FOR_PC	Table 6-5	O	
	Typical (50th percentile) effective Radar Cross Section	RCS	Table 6-5	O	
	Minimum Radar Cross Section	RCS_MIN	Table 6-5	O	
	Maximum Radar Cross Section	RCS_MAX	Table 6-5	O	
	SRP additional area	SRP_CONST_AREA	Table 6-5	O	
	Solar Radiation Pressure Coefficient	SOLAR_RAD_COEFF	Table 6-5	O	
	Solar Radiation Pressure 1	SOLAR_RAD_UNCERTAINTY	Table 6-5	O	
	Typical (50th percentile) absolute Visual Magnitude	VM_ABSOLUTE	Table 6-5	O	
	Minimum apparent Visual Magnitude	VM_APPARENT_MIN	Table 6-5	O	
	Typical (50th percentile) apparent Visual Magnitude	VM_APPARENT	Table 6-5	O	
	Maximum apparent Visual Magnitude	VM_APPARENT_MAX	Table 6-5	O	
	Typical (50th percentile) coefficient of reflectivity	REFLECTIVITY	Table 6-5	O	
	Primary mode of attitude control	ATT_CONTROL_MODE	Table 6-5	O	
	Type of actuator for attitude control	ATT_ACTUATOR_TYPE	Table 6-5	O	
	Accuracy of attitude knowledge	ATT_KNOWLEDGE	Table 6-5	O	
	Ability (e.g. deadband) to control attitude	ATT_CONTROL	Table 6-5	O	
	Combined ability to both knowledge and control attitude	ATT_POINTING	Table 6-5	O	
	Average maneuver frequency	AVG_MANEUVER_FREQ	Table 6-5	O	
	Maximum composite thrust	MAX_THRUST	Table 6-5	O	
	Total ΔV capability of the spacecraft at beginning of life	DV_BOL	Table 6-5	O	
	Total ΔV remaining	DV_REMAINING	Table 6-5	O	
	Moment of Inertia about the X-axis	IXX	Table 6-5	O	
	Moment of Inertia about the Y-axis	IYY	Table 6-5	O	
	Moment of Inertia about the Z-axis	IZZ	Table 6-5	O	
	Inertia Cross Product of the X & Y axes	IXY	Table 6-5	O	
	Inertia Cross Product of the X & Z axes	IXZ	Table 6-5	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	Inertia Cross Product of the Y & Z axes	IYZ	Table 6-5	O	
	End of the Space Object Physical Characteristics specification	PHYS_STOP	Table 6-5	M	

OCM Data: Covariance Time History

Item	Feature	Keyword	Reference	Status M/O/C	Support
	Covariance time history logical block	N/A	Table 6-6	O	
	OCM start of a covariance time history section	COV_START	Table 6-6	M	
	Comment	COMMENT	Table 6-6	O	
	Identification number for this covariance time history block	COV_ID	Table 6-6	O	
	Identification number for the previous covariance time history	COV_PREV_ID	Table 6-6	O	
	Identification number for the next covariance time history	COV_NEXT_ID	Table 6-6	O	
	Basis of this covariance time history	COV_BASIS	Table 6-6	O	
	Identification number for the orbit determination, navigation solution, or simulation	COV_BASIS_ID	Table 6-6	O	
	Reference frame of the covariance time history	COV_REF_FRAME	Table 6-6	O	
	Epoch of the covariance data reference frame	COV_FRAME_EPOCH	Table 6-6	C	
	Minimum scale factor to apply to this covariance data	COV_SCALE_MIN	Table 6-6	O	
	Maximum scale factor to apply to this covariance data	COV_SCALE_MAX	Table 6-6	O	
	Confidence in the covariance errors matching reality	COV_CONFIDENCE	Table 6-6	O	
	Covariance composition	COV_TYPE	Table 6-6	O	
	Units of covariance data line standard deviations (i.e., the square root of the variances supplied in the covariance matrix diagonal)	COV_UNITS	Table 6-6	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	elements) that follow the time tag				
	Covariance data	... <Insert covariance data here>	Table 6-6	M	
	End of a covariance time history section	COV_STOP	Table 6-6	M	

OCM Data: Maneuver Specification

Item	Feature	Keyword	Reference	Status M/O/C	Support
	Maneuver time history logical block	N/A	Table 6-7	O	
	Start of a maneuver data block	MAN_START	Table 6-7	M	
	Comment	COMMENT	Table 6-7	O	
	Identification number for this maneuver	MAN_ID	Table 6-7	M	
	Identification number for the previous maneuver	MAN_PREV_ID	Table 6-7	O	
	Identification number for the next maneuver	MAN_NEXT_ID	Table 6-7	O	
	Basis of this maneuver time history data	MAN_BASIS	Table 6-7	O	
	Identification number of the orbit determination	MAN_BASIS_ID	Table 6-7	O	
	Maneuver device identifier	MAN_DEVICE_ID	Table 6-7	M	
	Completion time of the previous maneuver	MAN_PREV_EPOCH	Table 6-7	O	
	Start time of the next maneuver	MAN_NEXT_EPOCH	Table 6-7	O	
	Specifies the purpose of the maneuver	MAN_PURPOSE	Table 6-7	O	
	Specifies the source of the orbit and/or attitude state(s)	MAN_PRED_SOURCE	Table 6-7	O	
	Reference frame in which the maneuver vector direction data is provided	MAN_REF_FRAME	Table 6-7	O	
	Epoch of the maneuver data reference frame	MAN_FRAME_EPOCH	Table 6-7	C	
	Origin of maneuver gravitational assist body	GRAV_ASSIST_NAME	Table 6-7	O	
	Duty cycle type to use for this maneuver time history	DC_TYPE	Table 6-7	O	
	Start time of the duty cycle-based maneuver window	DC_WIN_OPEN	Table 6-7	C	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	End time of the duty cycle-based maneuver window	DC_WIN_CLOSE	Table 6-7	C	
	Minimum number of "ON" duty cycles	DC_MIN_CYCLES	Table 6-7	O	
	Maximum number of "ON" duty cycles	DC_MAX_CYCLES	Table 6-7	O	
	Start time of the initial duty cycle-based maneuver sequence	DC_EXEC_START	Table 6-7	C	
	End time of the final duty cycle-based maneuver sequence	DC_EXEC_STOP	Table 6-7	C	
	THRUST duty cycle reference time tag	DC_REF_TIME	Table 6-7	C	
	Thruster pulse "ON" duration	DC_TIME_PULSE_DURATION	Table 6-7	C	
	Elapsed time between the start of one pulse and the start of the next	DC_TIME_PULSE_PERIOD	Table 6-7	C	
	Specifies the "ON" reference unit vector direction	DC_REF_DIR	Table 6-7	C	
	Body frame reference unit vector "trigger" direction	DC_BODY_TRIGGER	Table 6-7	C	
	Phase angle offset of thruster pulse start	DC_PA_START_ANGLE	Table 6-7	C	
	Phase angle of thruster pulse stop	DC_PA_STOP_ANGLE	Table 6-7	C	
	Set of maneuver elements of information to follow the maneuver time tag	MAN_COMPOSITION	Table 6-7	M	
	Units of maneuver data line elements that follow the time tag(s)	MAN_UNITS	Table 6-7	O	
	Maneuver time history data	... < Insert maneuver data here >	Table 6-7	M	
	End maneuver data block	MAN_STOP	Table 6-7	M	

OCM Data: Perturbations Specification

Item	Feature	Keyword	Reference	Status M/O/C	Support
	Orbit perturbations parameters	N/A	Table 6-10	O	
	Start of the perturbations specification	PERT_START	Table 6-10	M	
	Comment	COMMENT	Table 6-10	O	
	Atmosphere model used in the simulation	ATMOSPHERIC_MODEL	Table 6-10	O	
	Gravity model used in the simulation	GRAVITY_MODEL	Table 6-10	O	
	Oblate spheroid equatorial radius	EQUATORIAL_RADIUS	Table 6-10	O	
	Gravitational coefficient of attracting body	GM	Table 6-10	O	
	"N-body" gravitational perturbations bodies used	N_BODY_PERTURBATIONS	Table 6-10	O	
	Central body angular rotation rate	CENTRAL_BODY_ROTATION	Table 6-10	O	
	Inverse of the central body's oblate spheroid oblateness	OBLATE_FLATTENING	Table 6-10	O	
	Ocean tides model	OCEAN_TIDES_MODEL	Table 6-10	O	
	Solid tides model	SOLID_TIDES_MODEL	Table 6-10	O	
	Reduction theory used for precession and nutation modeling	REDUCTION_THEORY	Table 6-10	O	
	Albedo model	ALBEDO_MODEL	Table 6-10	O	
	Number of grid points used in the albedo model	ALBEDO_GRID_SIZE	Table 6-10	O	
	Shadow model used for Solar Radiation Pressure	SHADOW_MODEL	Table 6-10	O	
	List of planetary bodies for which SRP shadowing is modeled	SHADOW_BODIES	Table 6-10	O	
	SRP model used	SRP_MODEL	Table 6-10	O	
	Source and version of the Space Weather data used	SW_DATA_SOURCE	Table 6-10	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	in the creation of this message				
	Epoch of the Space Weather data	SW_DATA_EPOCH	Table 6-10	O	
	Method used to select or interpolate any and all sequential space weather data	SW_INTERP_METHOD	Table 6-10	O	
	A fixed (time invariant) value of the planetary 3-hour-range geomagnetic index Kp	FIXED_GEOMAG_KP	Table 6-10	O	
	A fixed (time invariant) value of the 3-hourly (equivalent range) geomagnetic index Ap	FIXED_GEOMAG_AP	Table 6-10	O	
	A fixed (time invariant) value of the planetary 1-hour-range geomagnetic index Dst	FIXED_GEOMAG_DST	Table 6-10	O	
	A fixed (time invariant) value of the solar flux daily proxy F10.7	FIXED_F10P7	Table 6-10	O	
	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy F10.7	FIXED_F10P7_MEAN	Table 6-10	O	
	A fixed (time invariant) value of the solar flux daily proxy M10.7	FIXED_M10P7	Table 6-10	O	
	A fixed (time invariant) value of the solar flux 81-day running center-averaged proxy M10.7	FIXED_M10P7_MEAN	Table 6-10	O	
	A fixed (time invariant) value of the solar flux daily proxy S10.7	FIXED_S10P7	Table 6-10	O	
	A fixed (time invariant) value of the solar flux 81-day running center-	FIXED_S10P7_MEAN	Table 6-10	O	

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Item	Feature	Keyword	Reference	Status M/O/C	Support
	averaged proxy S10.7				
	A fixed (time invariant) value of the solar flux daily proxy Y10.7	FIXED_Y10P7	Table 6- 10	O	
	A fixed (time invariant) value of the solar flux 81-day running center- averaged proxy Y10.7	FIXED_Y10P7_MEAN	Table 6- 10	O	
	Perturbations Data Block Stop	PERT_STOP	Table 6- 10	M	

OCM Data: Orbit Determination Data

Item	Feature	Keyword	Reference	Status M/O/C	Support
	Orbit determination parameters logical block	N/A	Table 6-11	O	
	Start of an orbit determination data section	OD_START	Table 6-11	M	
	Comment	COMMENT	Table 6-11	O	
	Identification number for this orbit determination	OD_ID	Table 6-11	M	
	Identification number for the previous orbit determination	OD_PREV_ID	Table 6-11	O	
	Type of orbit determination method used	OD_METHOD	Table 6-11	M	
	UTC epoch of the orbit determination solved-for state	OD_EPOCH	Table 6-11	M	
	Days elapsed between first accepted observation and OD_EPOCH	DAYS_SINCE_FIRST_OBS	Table 6-11	O	
	Days elapsed between last accepted observation and OD_EPOCH	DAYS_SINCE_LAST_OBS	Table 6-11	O	
	Number of days of observations recommended for the OD	RECOMMENDED_OD_SPAN	Table 6-11	O	
	Actual time span in days used for the OD	ACTUAL_OD_SPAN	Table 6-11	O	
	Number of observations available within the actual OD time span	OBS_AVAILABLE	Table 6-11	O	
	Number of observations accepted within the actual OD time span	OBS_USED	Table 6-11	O	
	Number of sensor tracks available for the OD within the actual time span	TRACKS_AVAILABLE	Table 6-11	O	
	Number of sensor tracks accepted for	TRACKS_USED	Table 6-11	O	

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Item	Feature	Keyword	Reference	Status M/O/C	Support
	the OD within the actual time span				
	Maximum time between observations in the OD	MAXIMUM_OBS_GAP	Table 6-11	O	
	Positional error ellipsoid 1σ major eigenvalue at OD	OD_EPOCH_EIGMAJ	Table 6-11	O	
	Positional error ellipsoid 1σ intermediate eigenvalue at OD	OD_EPOCH_EIGMED	Table 6-11	O	
	Positional error ellipsoid 1σ minor eigenvalue at OD	OD_EPOCH_EIGMIN	Table 6-11	O	
	Max positional error ellipsoid 1σ major eigenvalue	OD_MAX_PRED_EIGMAJ	Table 6-11	O	
	Min positional error ellipsoid 1σ minor eigenvalue	OD_MIN_PRED_EIGMIN	Table 6-11	O	
	OD confidence metric, which by definition spans 0 to 100%	OD_CONFIDENCE	Table 6-11	O	
	Generalized Dilution of Precision for this orbit determination	GDOP	Table 6-11	O	
	Number of solve-for states in the orbit determination	SOLVE_N	Table 6-11	O	
	State elements solved for in the orbit determination	SOLVE_STATES	Table 6-11	O	
	Number of consider parameters used in the orbit determination	CONSIDER_N	Table 6-11	O	
	Consider parameters used in the orbit determination	CONSIDER_PARAMS	Table 6-11	O	
	Number of sensors used in the orbit determination	SENSORS_N	Table 6-11	O	
	Sensors used in the orbit determination	SENSORS	Table 6-11	O	
	Weighted RMS residual ratio	WEIGHTED_RMS	Table 6-11	O	
	File name(s) and/or associated identification	TDM_MSG_LINK	Table 6-11	O	

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Item	Feature	Keyword	Reference	Status M/O/C	Support
	number(s) of Tracking Data Message (TDM) [9] observations				
	Observation data types utilized in this orbit determination	DATA_TYPES	Table 6-11	O	
	End of an orbit determination data section	OD_STOP	Table 6-11	M	

OCM Data: User-Defined Parameters

Item	Feature	Keyword	Reference	Status M/O/C	Support
	User-Defined Parameters logical block	N/A	Table 6-12	O	
	OCM User-Defined Parameters start	USER_START	Table 6-12	M	
	Comment	COMMENT	Table 6-12	O	
	As defined by user, "essential information that cannot be conveyed in COMMENT statements"	(User-defined keywords)	Table 6-12	M	
	OCM User-Defined Parameters end	USER_STOP	Table 6-12	M	

ANNEX B

SANA REGISTRY SOURCES FOR SELECTED KEYWORD VALUES**(NORMATIVE)**

The values in this annex represent the recommended values for selected keywords present in OPM, OMM, OEM or OCM message. For details and descriptions of the keyword interpretations, the reader is directed to ANNEX N, Reference [N-1] and [N-7]. The message creator should seek to confirm with the recipient(s) that their software can support the selected keyword value, particularly for more complex content such as reference frames, orbital elements and covariance definitions.

These recommended values are stored on the SANA Registry, globally accessible on the CCSDS SANA registry website located at:

https://sanaregistry.org/r/navigation_standard_normative_annexes

Note that the message creator or recipient may wish to automate processing of SANA registry normative content, which can be done by ingesting and processing of such content in electronic format. These formats can be accessed via the “Actions” link on each registry, e.g. for the Orbital Elements registry, a comma separated value (CSV) format can be exported at: https://www.sanaregistry.org/r/orbital_elements?_export=csv and a (JSON) format at: https://www.sanaregistry.org/r/orbital_elements?_export=json. Note that both the registry and these electronic data formats contain the specification of how many vector elements corresponding to each keyword value.

Exchange partners may submit additional (new) keyword values for consideration of future inclusion into the SANA registry by submitting a detailed email request (<mailto:info@sanaregistry.org>) per ANNEX M, Section M2. The CCSDS Area or Working Group responsible for the maintenance of the ODM at the time of the request is the approval authority. Until a suggested value is included in the SANA registry, exchange partners may define and use values that are not listed in the SANA registry if mutually agreed between message exchange partners.

B1 MESSAGE ORIGINATORS

The set of recommended values for the **ORIGINATOR** keyword is enumerated in the *SANA Registry of Organizations*, located at:

<https://sanaregistry.org/r/organizations>

B2 REFERENCE FRAME CENTERS AND THIRD-BODY PERTURBATIONS

A set of allowed values for the reference frame center keywords (**CENTER_NAME** for OPM, OEM, OMM, and OCM, as well as **N_BODY_PERTURBATIONS** in the OCM) is enumerated in the *SANA Registry of Orbit Centers*, located at:

https://sanaregistry.org/r/orbit_centers

Note that these values may also be useful to specify another platform (satellite, airframe, ground vehicle, etc.) as the reference frame origin to permit the specification of relative positional state time history data. In this case, message authors shall clearly communicate to recipients that the orbit center is not a gravitational center, that propagation of ephemeris vectors or extrapolation of ephemeris start/stop states is not advisable, and that interpolation of state time histories should not be accomplished using classical orbit propagation forces, e.g., gravitational constants, drag.

B3 TIME SYSTEMS

A set of allowed values for the **TIME_SYSTEM** keyword is enumerated in the *SANA Registry of Time Systems*, located at:

https://sanaregistry.org/r/time_systems

B4 NON-ORBIT-RELATIVE REFERENCE FRAMES

A set of allowed non-orbit-relative reference frame values for ***_REF_FRAME** keywords is enumerated in the *SANA Registry of Celestial Body Reference Frames*, located at:

https://sanaregistry.org/r/celestial_body_reference_frames

B5 ORBIT-RELATIVE REFERENCE FRAMES

In addition to the above reference frames, maneuver and covariance data may be selected from the list of allowed orbit-relative reference frames using ***_REF_FRAME** keyword values enumerated in the *SANA Registry of Orbit-Relative Reference Frames*, located at:

https://sanaregistry.org/r/orbit_relative_reference_frames

Note that two types of orbit-relative local reference frames exist: inertial and rotating. When transforming velocity terms between inertial and rotating frames, remember to properly incorporate the $(\bar{\omega} \times \bar{r})$ contribution.

B6 ADDITIONAL SPACECRAFT AND ATTITUDE REFERENCE FRAMES

An additional allowed set of spacecraft and attitude control reference frame values for *_REF_FRAME keywords is enumerated in the *SANA Registry of Spacecraft and Attitude Control Reference Frames*, located at:

https://sanaregistry.org/r/spacecraft_body_reference_frames

In the numerous instances in these spacecraft body reference frames are specified by a keyword followed by an “i” (e.g., ACTUATOR_i), the “i” should be replaced by an integer value (1, 2, ...) to denote the “ith” reference frame within the set of those particular reference frames.

B7 ORBITAL ELEMENTS

A set of allowed values for the ORB_TYPE keyword is enumerated in the *SANA Registry of Orbital Elements*, located at:

https://sanaregistry.org/r/orbital_elements

Unique to the Orbit Comprehensive Message (OCM), orbit element states and/or time histories may be specified in multiple element set types.

Orbit elements shall be interpreted as osculating elements unless either explicitly specified via the ORB_AVERAGING keyword or as mutually agreed between message exchange partners to contain mean elements (e.g. singly- or doubly-averaged elements based upon Kozai, Brouwer, or other theories).

Inertial reference frames shall be specified when employing inertial element sets.

When employing non-inertial element sets, inertial reference frames shall not be specified.

B8 ADDITIONAL COVARIANCE REPRESENTATIONS

Covariance matrices may either be specified as representing uncertainties expressed in the above “Orbital Elements” types (e.g., COV_TYPE may be set to an ORB_TYPE such as CARTPVA) or may specify an event-based covariance type which includes event time uncertainties as enumerated in the *SANA Registry of Covariance Representations*, located at:

https://sanaregistry.org/r/orbital_covariance_matrix_types

B9 ATMOSPHERE MODELS

A set of allowed values for the ATMOSPHERIC_MODEL keyword is enumerated in the *SANA Registry of Atmosphere Models*, located at:

https://sanaregistry.org/r/atmosphere_models

B10 GRAVITY MODELS

A set of allowed values for the **GRAVITY_MODEL** keyword is enumerated in the *SANA Registry of Gravity Models*, located at:

https://sanaregistry.org/r/gravity_models

B11 OBJECT TYPES

A set of allowed values for the **OBJECT_TYPE** keyword is enumerated in the *SANA Registry of Object Types*, located at:

https://sanaregistry.org/r/object_types

B12 OPERATIONAL STATUS

A set of allowed values for the **OPS_STATUS** keyword is enumerated in the *SANA Registry of Operational Status of Space Object*, located at:

https://sanaregistry.org/r/operational_status

B13 ORBIT AVERAGING TECHNIQUES

A set of allowed values for the **ORB_AVERAGING** keyword is enumerated in the *SANA Registry of Orbit Averaging Techniques*, located at:

https://sanaregistry.org/r/orbit_averaging

B14 ORBIT TYPES

A set of allowed values for the **ORBIT_CATEGORY** keyword is enumerated in the *SANA Registry of Orbit Types*, located at:

https://sanaregistry.org/r/orbit_types

ANNEX C

TECHNICAL MATERIAL

(INFORMATIVE)

C1 SATELLITE PHYSICAL CHARACTERISTICS: OPTIMALLY-ENCOMPASSING BOX (OEB) (INFORMATIVE)

This section of the informative technical annex defines satellite dimensional and orientational parameters of the OCM's satellite physical characteristics specification.

To facilitate improved modeling of the physical space occupied by a space object, the space object's attitude/orientation, the probability of a hard body collision occurring, and drag and SRP acceleration forces, the OCM allows the specification of an “**Optimally-Encompassing Box**” (OEB). Note that the OEB describes the physical space occupied by the space object, which may or may not align with the inertia tensor for that object.

For a box-shaped satellite (e.g., a CubeSat) without appendages, the satellite and a corresponding OEB would be a one-to-one mapping.

For a satellite having solar arrays that extend from the spacecraft body structure, the OEB would extend from the main satellite body to encompass the deployed solar arrays as well.

The OEB shape is shown in Fig. C- 1 below. As illustrated, the OEB reference frame axes (depicted as **RED** dotted lines) are defined by convention as follows:

- The OEB x-axis is along the **longest** dimension of the box (\hat{X}_{OEB_MAX}). This is sometimes referred to the “span” of the space object.
- The OEB y-axis is along the **intermediate** orthonormal dimension (\hat{Y}_{OEB_INT})
- The OEB z-axis is along the **shortest** orthonormal dimension (\hat{Z}_{OEB_MIN}).

The BOX shape can easily represent a cube by setting all orthonormal dimensions equal. In the event that the longest two or three orthonormal dimensions are equivalent, \hat{X}_{OEB_MAX} is defined as the direction along one of those longest dimensions and the next as \hat{Y}_{OEB_INT} .

In the event that the longest two or three principal axis dimensions of the box are equivalent, \hat{X}_{OEB_MAX} is defined as the direction along one of those longest principal dimensions and the next as \hat{Y}_{OEB_MED} .

The OEB z-axis is always defined as: $\hat{Z}_{OEB_MIN} = \hat{X}_{OEB_MAX} \times \hat{Y}_{OEB_INT}$.

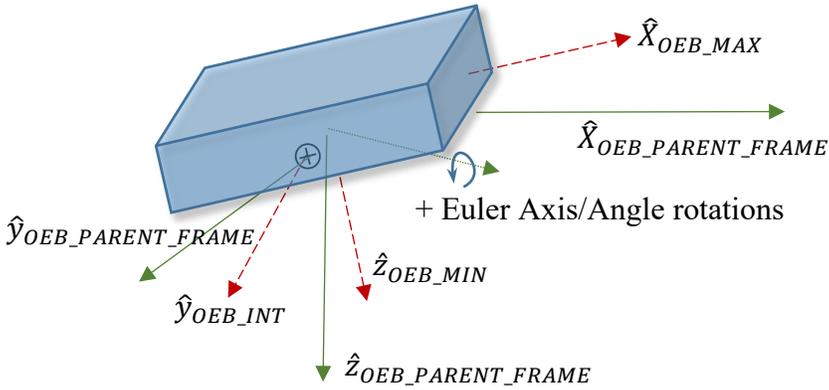


Fig. C- 1 Depiction of Optimally-Enclosing Box and definitions of MAX, INT, and MIN orientation vectors relative to OEB parent frame

NOTE: parent and body axis are shown in proximity to each other for display purposes only, but could generally be in any orientation as specified by the quaternion.

A fixed orientation of the Optimally-Encompassing Box with respect to the user-specified “OEB_PARENT_FRAME” is defined using a **quaternion** that maps from the user-specified OEB_PARENT_FRAME to the Optimally-Encompassing Box vector directions. The above figure shows the proper definitions and adopted sign conventions. The resulting transformation sequence is:

$$\begin{bmatrix} x \\ y \\ z \end{bmatrix}_{OEB} = [M] \begin{bmatrix} x \\ y \\ z \end{bmatrix}_{OEB_PARENT_FRAME}$$

Where the frame transformation matrix [M] is a function of the quaternion components

$$[M] = \begin{bmatrix} Q_1^2 - Q_2^2 - Q_3^2 + Q_c^2 & 2(Q_1Q_2 + Q_3Q_c) & 2(Q_1Q_3 - Q_2Q_c) \\ 2(Q_1Q_2 - Q_3Q_c) & -Q_1^2 + Q_2^2 - Q_3^2 + Q_c^2 & 2(Q_2Q_3 + Q_1Q_c) \\ 2(Q_1Q_3 + Q_2Q_c) & 2(Q_2Q_3 - Q_1Q_c) & -Q_1^2 - Q_2^2 + Q_3^2 + Q_c^2 \end{bmatrix}$$

The physical dimensions of the OEB (long, intermediate, and short dimensions) are specified via OEB_MAX, OEB_INT, and OEB_MIN respectively.

Cross-sectional area is modeled in the OCM as a combination of two parameter types:

- 1) an **attitude-independent**, constant cross-sectional area (e.g., **DRAG_CONST_AREA** or **SRP_CONST_AREA**)
- 2) **attitude-dependent** cross-sectional area as viewed along the OEB x, y and z axes (long, intermediate and short dimension directions) via **AREA_ALONG_OEB_MAX**, **AREA_ALONG_OEB_INT**, and **AREA_ALONG_OEB_MIN**, respectively.

The analyst may use one or a combination of both of these parameter types to best represent the total cross-sectional area profile of the space object to be used in drag, lift and SRP force estimates. The total cross-sectional area observed when viewed from an arbitrary unit-vector direction $[x\ y\ z]$ could be:

$$\text{TOTAL_AREA} = \text{DRAG_CONST_AREA} + \begin{bmatrix} \text{AREA_ALONG_OEB_MAX} \\ \text{AREA_ALONG_OEB_INT} \\ \text{AREA_ALONG_OEB_MIN} \end{bmatrix} \cdot [M] \begin{bmatrix} \widehat{x} \\ y \\ z \end{bmatrix}_{\text{OEB_PARENT_FRAME}}$$

For example, to model drag forces, a two-meter diameter spherical space object would be best modeled as a constant area ($\text{DRAG_CONST_AREA}=3.1415\text{m}^2$) with all three AREA_ALONG_OEB parameters defaulting to zero. Conversely, a ten-meter long, 10 cm box-crosssection gravity gradient boom would best be modeled by leaving DRAG_CONST_AREA defaulting to zero and setting $\text{AREA_ALONG_OEB_MAX} = \text{AREA_ALONG_OEB_INT} = 1\text{ m}^2$ and $\text{AREA_ALONG_OEB_MIN} = 0.01\text{ m}^2$. Finally, one can model a sphere encircling the gravity gradient boom's centroid using a combination of these approaches by setting $\text{DRAG_CONST_AREA}=3.1415\text{m}^2$, $\text{AREA_ALONG_OEB_MAX} = \text{AREA_ALONG_OEB_INT} = 1\text{ m}^2$ and $\text{AREA_ALONG_OEB_MIN} = 0.01\text{ m}^2$

As a second example, when modeling Solar Radiation Pressure (SRP) forces, note that many GEO spacecraft have very large solar arrays that dominate the total cross-sectional surface of the spacecraft. Importantly, these solar arrays are designed to remain as normal to the Sun as possible irrespective of current spacecraft bus orientation. As such, the cross-sectional area of these arrays is best portrayed in the OCM by setting SRP_CONST_AREA to the combined solar array surface area as observed along the Sun viewing vector, and the remaining bus cross-sectional areas can be set using the three attitude-dependent AREA_ALONG_OEB parameters.

C2 APPARENT-TO-ABSOLUTE VISUAL MAGNITUDE RELATIONSHIP (INFORMATIVE)

This section of the informative technical annex presents the relationships to be used to map apparent to absolute visual magnitude for inclusion in an OCM. These equations, based on ANNEX N, Reference [N-8], examine signal magnitude for reflected illumination by a Resident Space Object (RSO) that is exoatmospheric, meaning that its illumination by the Sun is not reduced or impeded by atmospheric transmission losses. The equations do not account for spatial distribution across multiple detectors, which involves characterizing the Point Spread Function of the system.

Definitions:

A_{Target}	Effective area of the target [m^2]
$E_{\text{EntranceAperture}}$	The point source irradiance reaching the sensor aperture [W/m^2]
$d_{\text{SunToTarget}}$	Distance from the sun to the target [m] (e.g. 1 AU = $1.4959787066 \times 10^{11}\text{ m}$)
$d_{\text{TargetToSensor}}$	Distance from target to sensor [m]
dia_{Target}	Effective diameter of the target, [m]

E_{Sun}	Exoatmospheric solar irradiance, nominally 1380 [W/m^2] at 1 AU
E_{Target}	Target Irradiance at Sensor without atmospheric loss [W/m^2]
E_0	Ref. Visual Magnitude (Vega) Irradiance [$2.77894 \times 10^{-8} W/m^2$]
F	General shadowing term accounting for the penumbra region's influence [unitless, $0 < F \leq 1$, 0 = umbra, and 1 = full Sun illumination]
I_{Sun}	Solar Intensity $\approx 3.088374161 \times 10^{25}$ [W/sr]
I_{Target}	Intensity of reflected energy from target treated as a point source [W/sr]
$Phase(\varphi)$	Geometric reflectance phase function [unitless, $0 < Phase(\varphi) \leq 1$]
φ	Critical Angle to the Sun (CATS) from sun to the sensor, as shown in Fig. C- 2 and referenced to the observed target [rad]
π	Pi constant
ρ	Reflectance of the target [between 0 (none) and 1 (perfect reflectance)]
$\tau_{Atmosphere}$	Effective transmission of the atmosphere [unitless, $0 < \tau \leq 1$]

Given an optical sensor's measured target entrance aperture radiance:

$$E_{target} = \frac{E_{EntranceAperture}}{\tau_{Atmosphere}} [W/m^2]$$

$$VM_{apparent} = -2.5 \log_{10} \frac{E_{target}}{E_0}, \text{ measured on the visual magnitude scale}$$

$$\text{or if } VM_{apparent} \text{ known: } E_{target} = E_0 10^{\left[-\frac{VM_{apparent}}{2.5}\right]}$$

$$I_{target} = E_{target} d_{TargetToSensor}^2 [W]$$

$$E_{Sun} = \frac{I_{Sun}}{d_{SunToTarget}^2} [W/m^2]$$

$$Phase(\varphi) = \frac{\sin \varphi + (\pi - \varphi) \cos \varphi}{\pi}$$

$$A_{Target} = \frac{\pi I_{Target}}{\rho F E_{Sun} Phase(\varphi)} [m^2] \text{ \{NOTE1: undefined in umbra (F=0=darkness), or no reflection (\rho = 0). NOTE2: If reflectance is unknown, one can assume a standard reference reflectance of fifteen percent}\}}$$

From which an effective diameter of the physical object can be roughly approximated as:

$$dia_{Target} \approx \sqrt{\frac{4 A_{Target}}{\pi}}$$

From the above equations, $VM_{absolute}$ “normalized” to a 1 AU Sun-to-target distance, a phase angle of 0° and an example reference 40,000 km target-to-sensor distance (equivalent to a GEO satellite tracked at 15.6° elevation above the optical site’s local horizon), is obtained as:

$$VM_{absolute} = -2.5 \log_{10} \left\{ \frac{E_{target}}{E_0} \right\}, \text{ from which:}$$

$$VM_{absolute} = -2.5 \log_{10} \left\{ \frac{[E_{Sun1AU}=1380 \text{ W/m}^2] [Phase(0 \text{ rad})=1.0] [\rho A_{Target} \text{ from above, in m}^2]}{\pi [E_0=2.77894 \times 10^{-8} \text{ W/m}^2] [(40,000,000)^2 \text{ m}^2]} \right\}$$

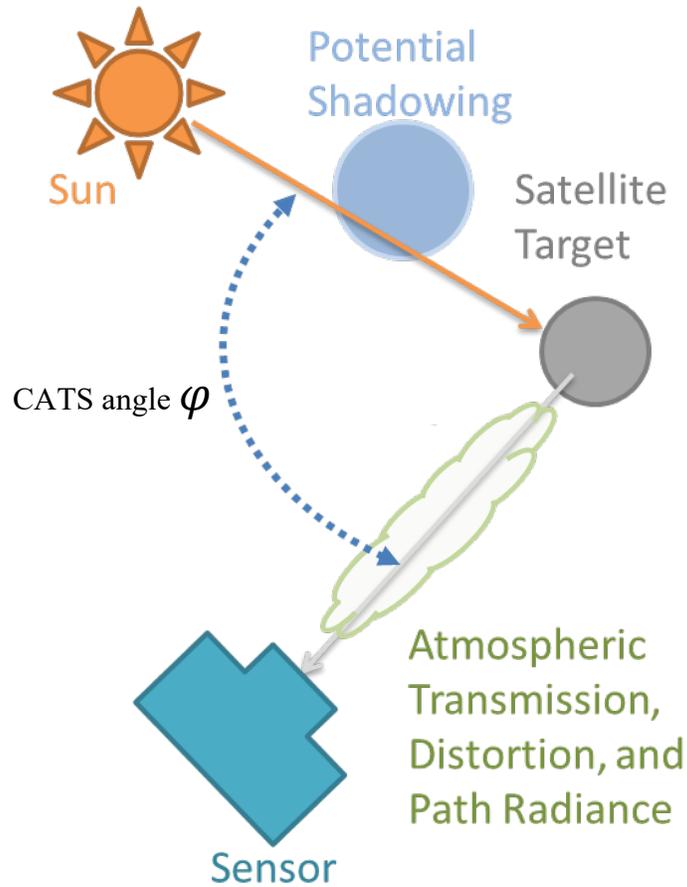


Fig. C- 2 Depiction of optical viewing Critical Angle to the Sun (CATS) angle geometry

C3 MANEUVER AND DUTY CYCLE DIAGRAMS (INFORMATIVE)

This section of the informative technical annex defines time-based and phase-angle-based duty cycle parameters.

A “duty cycle” is a cycle of thruster operation which operates intermittently rather than continuously, having an “on” interval followed by an “off” interval.

Time-based duty cycle parameters

Time-based duty cycle parameters define a window of duty cycle operations, the actual execution interval and “ON” and “OFF” intervals, as shown in Fig. C- 3.

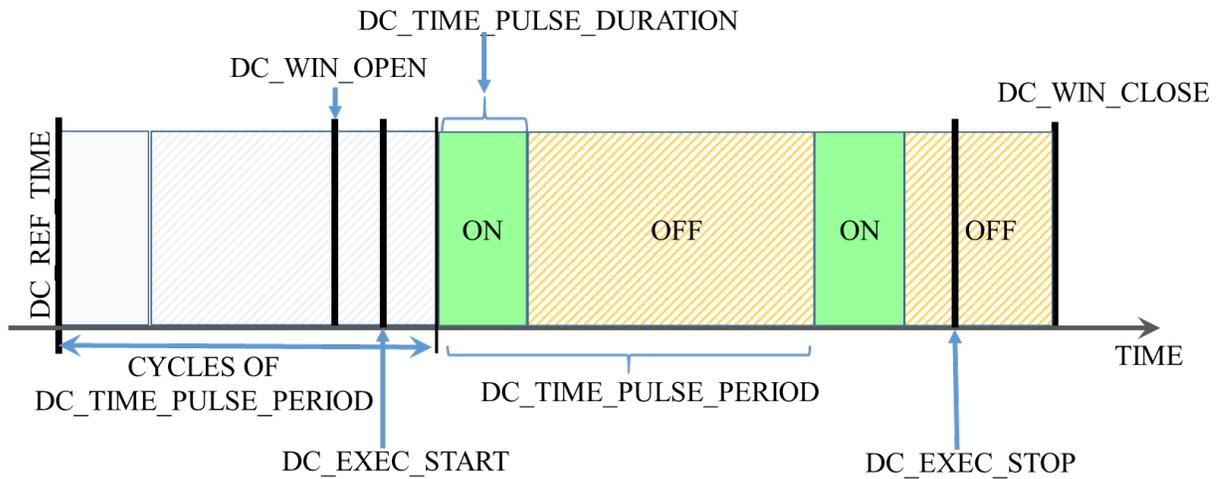


Fig. C- 3 Diagram of time-based duty cycle ($DC_TYPE = "TIME"$)

Angle-based duty cycle parameters

Angle-based duty cycle parameters also define a window of duty cycle operations and actual execution interval and “ON” and “OFF” intervals, but in this case the “ON” and “OFF” intervals are triggered by angular limits as shown in Fig. C- 4.

Where H represents the measurement matrix modeled in the state estimate, and W is a diagonal matrix of relative weights that represents the accuracy of the measurements. $\tilde{H}_0^T W \tilde{H}_0$ represents the information matrix, the inverse of which is the covariance matrix. By summing over time, one obtains an estimate of the state uncertainty from the time-derived measurement set.

C5 EULER AXIS/ANGLE INTERPOLATION (INFORMATIVE)

The Euler Axis and Angle representation of Euler's Theorem (see ANNEX N, Reference [N-17, pp. 10-14]) is an effective way to interpolate a series of covariance matrices, reference frames or maneuver, thrust, or acceleration vector directions.

Interpolation of a series of three-dimensional vectors:

As presented in [N-17], and consistent with the nomenclature of [N-1] where e_1 , e_2 , and e_3 represent the three vector components of axis of rotation \hat{e} and φ represents the angle of rotation, a time-based interpolation of adjacent unit vectors \hat{v}_A and \hat{v}_B in a reference frame can be undertaken as:

- (1) The axis of rotation \hat{e} can be obtained as: $\hat{e} = \frac{\hat{v}_B \times \hat{v}_A}{|\hat{v}_B \times \hat{v}_A|}$
- (2) Assuming a constant rotational rate during this interval, $\varphi(t) = \frac{(t-t_1) \cos^{-1}(\hat{v}_A \cdot \hat{v}_B)}{(t_2-t_1)}$
- (3) The orthonormal rotation matrix $[M(t)]$ is then

$$= \begin{pmatrix} (1 - \cos \varphi) \hat{e}_x^2 + \cos \varphi & (1 - \cos \varphi) \hat{e}_x \hat{e}_y + \hat{e}_z \sin \varphi & (1 - \cos \varphi) \hat{e}_x \hat{e}_z - \hat{e}_y \sin \varphi \\ (1 - \cos \varphi) \hat{e}_y \hat{e}_x - \hat{e}_z \sin \varphi & (1 - \cos \varphi) \hat{e}_y^2 + \cos \varphi & (1 - \cos \varphi) \hat{e}_y \hat{e}_z + \hat{e}_x \sin \varphi \\ (1 - \cos \varphi) \hat{e}_z \hat{e}_x + \hat{e}_y \sin \varphi & (1 - \cos \varphi) \hat{e}_z \hat{e}_y - \hat{e}_x \sin \varphi & (1 - \cos \varphi) \hat{e}_z^2 + \cos \varphi \end{pmatrix}$$

- (4) From which the interpolated vector at time t is then $\hat{v}(t) = [M(t)]\hat{v}_A$
- (5) The accompanying vector magnitudes (e.g. eigenvalues or thrust or acceleration magnitudes) may be interpolated using Lagrange polynomials or linear expressions.

Interpolation of a series of reference (or covariance eigenvector) frames:

The eigenvector matrix $[E(t)]$ contains the row-wise storage of the major, intermediate and minor eigenvectors at time t, taking care to ensure that this ordered "triad" of vectors adheres to the righthand rule. When interpolating between two eigenvector matrices $[E_1]$ and $[E_2]$ derived from two adjacent covariance matrices respectively, $[E(t)]$ can be evaluated as:

- (6) The rotation occurring between $[E_1]$ and $[E_2]$ is: $[M_{BA}] = [E_2][E_1]^T$
- (7) Compute $\sigma = (M_{BA11} + M_{BA22} + M_{BA33})$
- (8) The angle of rotation from A to B is: $\varphi_{BA} = \cos^{-1} \left[\frac{1}{2}(\sigma - 1) \right]$
- (9) Exercising caution to accommodate nonunique cases (when $\sin \varphi = 0$) as described in [N-17], the axis of rotation $\hat{e} = \left[\frac{(M_{BA23} - M_{BA32})}{2 \sin \varphi} \quad \frac{(M_{BA31} - M_{BA13})}{2 \sin \varphi} \quad \frac{(M_{BA12} - M_{BA21})}{2 \sin \varphi} \right]$
- (10) The angle of rotation at time t is $\varphi(t) = \frac{(t-t_1)\varphi_{BA}}{(t_2-t_1)}$
- (11) $[M(t)]$ can be computed using the above expression in step (3)
- (12) And finally, the eigenvector matrix $[E(t)] = [M(t)][E_1]$
- (13) When interpolating a series of covariance matrices, the accompanying eigenvalues may be interpolated using Lagrange polynomials or linear expressions.

C6 REGULAR EXPRESSIONS FOR VALIDATION AND INGEST OF ODMs (INFORMATIVE)

To accomplish validation and ingest of KVN versions of the ODM, the use of Regular Expressions (referred to as “Regex”) is strongly encouraged where possible. Regex offers a detailed and rigorous way to ensure proper validation, interpretation and conformance to Orbit Data Message content. Most programming languages support the Regex feature, including C#, C++, Delphi, HTML5, Java and Javascript, MySQL, Oracle, PCRE, Perl, PHP, PowerShell, Python, R, Ruby, Scala, TCL, VBscript, Visual Basic, and XML Schema.

Sample Regular Expression for “Keyword = CCSDS Date/Time Format”

The CCSDS Timecode format specified in Section 7.5.12 provides a convenient illustration of the power of using regular expressions. The color-coded Regex string below can be used to readily match any general ODM KVN line which sets a KVN keyword to a Timecode value. Of particular utility is the group naming capability (color-coded in green below) inherent with Regex, where the keyword, year, month, day, hour, minute, and second can be readily extracted and processed. This Regex sequence enforces the requirement that KVN keywords must be uppercase and can only consist of letters A-Z, digits 0-9, or underscores. Note that in this expression, the optional inclusion by the message creator of one or more white space characters with the “(\s*)?” sequences allows for maximum flexibility while still retaining a rigorous validation. NOTE: in some languages, you must explicitly enable “^” and “\$” matching at string line breaks to process a string containing a series of lines of the message.

```
^(?:\s*)?(?<keyword>[0-9A-Z_]*)(?:\s*)?=(?:\s*)?(?<yr>(?:\d{4}))-(?<mo>(?:\d{1,2}))-(?<dy>(?:\d{1,2}))T(?<hr>(?:\d{1,2})):(?<mn>(?:\d{1,2})):(?<sc>(?:\d{0,2})(?:\.\d*)?)?(?:\s*)?$
```

Regex pattern matching sequence:

Applying the above CCSDS Timecode Regex to a file containing the string “CREATION_DATE = 2020-09-13T00:09:47.059345” as an example, Figure C-5 illustrates how this Regex expression is used to rigorously validate and match the string and identify the specified group names.

```

Beginning match attempt at character 45
Match attempt failed after 1 steps
Beginning match attempt at character 46
ok
ok
ok
CREATION_DATE
CREATION_DATE
CREATION_DATE
CREATION_DATE
CREATION_DATE =
CREATION_DATE =
CREATION_DATE =
CREATION_DATE = 2020
CREATION_DATE = 2020
CREATION_DATE = 2020
CREATION_DATE = 2020-
CREATION_DATE = 2020-09
CREATION_DATE = 2020-09
CREATION_DATE = 2020-09
CREATION_DATE = 2020-09-
CREATION_DATE = 2020-09-13
CREATION_DATE = 2020-09-13
CREATION_DATE = 2020-09-13
CREATION_DATE = 2020-09-13T
CREATION_DATE = 2020-09-13T00
CREATION_DATE = 2020-09-13T00
CREATION_DATE = 2020-09-13T00:
CREATION_DATE = 2020-09-13T00:09
CREATION_DATE = 2020-09-13T00:09
CREATION_DATE = 2020-09-13T00:09
CREATION_DATE = 2020-09-13T00:09:
CREATION_DATE = 2020-09-13T00:09:47
CREATION_DATE = 2020-09-13T00:09:47.
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345
CREATION_DATE = 2020-09-13T00:09:47.059345n
CREATION_DATE = 2020-09-13T00:09:47.059345nbacktrac
CREATION_DATE = 2020-09-13T00:09:47.059345ok
CREATION_DATE = 2020-09-13T00:09:47.059345ok
Match found in 42 steps
Beginning match attempt at character 89
Match attempt failed after 1 steps
Beginning match attempt at character 90

```

Fig. C- 5 Regex pattern matching sequence for CCSDS Timecode

Sample Regular Expression for “Keyword = NonDecimalString”

Another common and useful Regex addresses the setting of keywords to a free text string. Consistent with requirements for KVN values as specified in Section 7.5, the following Regex matches this sequence:

```
^(?:\s*)(?<keyword>[0-9A-Z_]*)(?:\s*)=(?:\s*)(?<value>(?:(?:[0-9A-Z_\- ]*)|(?:[0-9a-z_\- ]*)))(?:\s*)$
```

Sample Regular Expression for “Keyword = FreeTextString”

Free text strings that are contain exclusively uppercase or lowercase alphabetical characters, numbers, and/or the decimal point, hyphen, space or underscore characters may be matched as follows. Note that the inclusion of a decimal point in this Regex sequence means that numerical KVN values can be matched with potentially unintended consequences; numbers should be specifically matched as a number (as shown below). Consistent with requirements for KVN values as specified in Section 7.5, the following Regex matches this sequence:

```
^(?:\s*)(?<keyword>[0-9A-Z_]*)(?:\s*)=(?:\s*)(?<value>(?:(?:[0-9A-Z_\- ]*)|(?:[0-9a-z_\- ]*)))(?:\s*)$
```

Sample Regular Expression for “Keyword = CCSDS Numerical Value with optional units”

Consistent with the requirements set forth in Sections 7.5.4 through 7.5.9, KVN values which are integers or non-integer fixed or floating-point numbers may be matched with this Regex sequence, with named groups “keyword,” “value,” and “units” set accordingly:

```
^(?:\s*)(?<keyword>[0-9A-Za-z_]*)(?:\s*)=(?:\s*)(?<value>(?:[+-]?(?:[0-9]+)(?:\.\d*)?(?:[eE][+-]?[0-9]+)?)(?:\s*)(?<unit>[0-9A-Za-z/_]*)(?:\s*)?$
```

Sample Regular Expression for “Keyword = Multipartite CCSDS Numerical Values”

For orbit, covariance and/or maneuver lines providing a multipartite sequence of numerical values, the following Regex pattern may be used (in this example, to capture and set value1, value2 and value3 for a 3-element set of numbers):

```
^(?:\s*)(?<value1>(?:[+-]?(?:[0-9]+)(?:\.\d*)?(?:[eE][+-]?[0-9]+)?)(?:\s+)(?<value2>(?:[+-]?[0-9]+)(?:\.\d*)?(?:[eE][+-]?[0-9]+)?)(?:\s+)(?<value3>(?:[+-]?[0-9]+)(?:\.\d*)?(?:[eE][+-]?[0-9]+)?)(?:\s*)$
```

ANNEX D

OPM EXAMPLES**(INFORMATIVE)**

The following are examples of Orbit Parameter Messages (OPMs).

OPM examples in KVN:

The following figures are examples of OPMs in Keyword Value Notation (KVN) format. The first has only a state; the second has state, Keplerian elements, and maneuvers; the third and fourth include the position/velocity covariance matrix.

```

CCSDS_OPM_VERS = 3.0
CREATION_DATE   = 1998-11-06T09:23:57
ORIGINATOR      = JAXA

COMMENT         GEOCENTRIC, CARTESIAN, EARTH FIXED
OBJECT_NAME     = OSPREY 5
OBJECT_ID       = 1998-999A
CENTER_NAME     = EARTH
REF_FRAME       = ITRF2000
TIME_SYSTEM     = UTC

EPOCH =          1998-12-18T14:28:15.1172
X =              6503.514000
Y =              1239.647000
Z =              -717.490000
X_DOT =          -0.873160
Y_DOT =           8.740420
Z_DOT =          -4.191076
MASS =           3000.000000
SOLAR_RAD_AREA =  18.770000
SOLAR_RAD_COEFF =  1.000000
DRAG_AREA =      18.770000
DRAG_COEFF =     2.500000

```

Fig. D-1: Simple OPM file example

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OPM_VERS      = 3.0

COMMENT  Generated by GSOC, R. Kiehling
COMMENT  Current intermediate orbit IO2 and maneuver planning data

CREATION_DATE       = 2000-06-03T05:33:00.000
ORIGINATOR          = GSOC

OBJECT_NAME         = EUTELSAT W4
OBJECT_ID           = 2000-028A
CENTER_NAME         = EARTH
REF_FRAME           = TOD
TIME_SYSTEM         = UTC

COMMENT  State Vector
EPOCH               = 2006-06-03T00:00:00.000
X                   = 6655.9942           [km]
Y                   = -40218.5751        [km]
Z                   = -82.9177           [km]
X_DOT               = 3.11548208        [km/s]
Y_DOT               = 0.47042605        [km/s]
Z_DOT               = -0.00101495       [km/s]

COMMENT  Keplerian elements
SEMI_MAJOR_AXIS     = 41399.5123         [km]
ECCENTRICITY        = 0.020842611
INCLINATION         = 0.117746          [deg]
RA_OF_ASC_NODE      = 17.604721         [deg]
ARG_OF_PERICENTER   = 218.242943       [deg]
TRUE_ANOMALY        = 41.922339        [deg]
GM                  = 398600.4415       [km**3/s**2]

COMMENT  Spacecraft parameters
MASS                 = 1913.000          [kg]
SOLAR_RAD_AREA       = 10.000           [m**2]
SOLAR_RAD_COEFF      = 1.300
DRAG_AREA            = 10.000           [m**2]
DRAG_COEFF           = 2.300

COMMENT  2 planned maneuvers

COMMENT  First maneuver: AMF-3
COMMENT  Non-impulsive, thrust direction fixed in inertial frame
MAN_EPOCH_IGNITION  = 2000-06-03T09:00:34.1
MAN_DURATION        = 132.60            [s]
MAN_DELTA_MASS      = -18.418           [kg]
MAN_REF_FRAME       = J2000
MAN_DV_1            = -0.02325700       [km/s]
MAN_DV_2            = 0.01683160        [km/s]
MAN_DV_3            = -0.00893444       [km/s]

COMMENT  Second maneuver: first station acquisition maneuver
COMMENT  impulsive, thrust direction fixed in RTN frame
MAN_EPOCH_IGNITION  = 2000-06-05T18:59:21.0
MAN_DURATION        = 0.00              [s]
MAN_DELTA_MASS      = -1.469            [kg]
MAN_REF_FRAME       = RTN
MAN_DV_1            = 0.00101500        [km/s]
MAN_DV_2            = -0.00187300       [km/s]
MAN_DV_3            = 0.00000000        [km/s]

```

Fig. D-2: OPM file example with optional Keplerian elements and two maneuvers

```

CCSDS_OPM_VERS = 3.0

CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR    = JAXA
MESSAGE_ID    = OPM 201113719185

COMMENT       GEOCENTRIC, CARTESIAN, EARTH FIXED
OBJECT_NAME   = OSPREY 5
OBJECT_ID     = 1998-999A
CENTER_NAME   = EARTH
REF_FRAME     = ITRF1997
TIME_SYSTEM   = UTC

EPOCH =      1998-12-18T14:28:15.1172
X =          6503.514000
Y =          1239.647000
Z =          -717.490000
X_DOT =      -0.873160
Y_DOT =       8.740420
Z_DOT =      -4.191076

MASS =        3000.000000
SOLAR_RAD_AREA = 18.770000
SOLAR_RAD_COEFF = 1.000000
DRAG_AREA =   18.770000
DRAG_COEFF =  2.500000

CX_X =  3.331349476038534e-04
CY_X =  4.618927349220216e-04
CY_Y =  6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z =  3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z =  2.484949578400095e-07
CX_DOT_X_DOT =  4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z =  1.798098699846038e-07
CY_DOT_X_DOT =  2.608899201686016e-10
CY_DOT_Y_DOT =  1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z =  3.540310904497689e-07
CZ_DOT_X_DOT =  1.869263192954590e-10
CZ_DOT_Y_DOT =  1.008862586240695e-10
CZ_DOT_Z_DOT =  6.224444338635500e-10
    
```

Fig. D-3: OPM file example with covariance matrix

```

CCSDS_OPM_VERS      = 3.0
COMMENT  Generated by GSOC, R. Kiehling
COMMENT  Current intermediate orbit IO2 and maneuver planning data
CREATION_DATE       = 2000-06-03T05:33:00.000
ORIGINATOR          = GSOC
OBJECT_NAME         = EUTELSAT W4
OBJECT_ID           = 2000-028A
CENTER_NAME         = EARTH
REF_FRAME           = TOD
TIME_SYSTEM         = UTC
COMMENT  State Vector
EPOCH               = 2006-06-03T00:00:00.000
X                   = 6655.9942           [km]
Y                   = -40218.5751        [km]
Z                   = -82.9177           [km]
X_DOT               = 3.11548208         [km/s]
Y_DOT               = 0.47042605         [km/s]
Z_DOT               = -0.00101495        [km/s]
COMMENT  Keplerian elements
SEMI_MAJOR_AXIS     = 41399.5123         [km]
ECCENTRICITY        = 0.020842611
INCLINATION         = 0.117746           [deg]
RA_OF_ASC_NODE      = 17.604721         [deg]
ARG_OF_PERICENTER   = 218.242943        [deg]
TRUE_ANOMALY        = 41.922339         [deg]
GM                  = 398600.4415        [km**3/s**2]
COMMENT  Spacecraft parameters
MASS                = 1913.000           [kg]
SOLAR_RAD_AREA      = 10.000            [m**2]
SOLAR_RAD_COEFF     = 1.300
DRAG_AREA           = 10.000            [m**2]
DRAG_COEFF          = 2.300
COV_REF_FRAME       = RTN
CX_X = 3.331349476038534e-04           [km**2]
CY_X = 4.618927349220216e-04           [km**2]
CY_Y = 6.782421679971363e-04           [km**2]
CZ_X = -3.070007847730449e-04          [km**2]
CZ_Y = -4.221234189514228e-04          [km**2]
CZ_Z = 3.231931992380369e-04           [km**2]
CX_DOT_X = -3.349365033922630e-07       [km**2/s]
CX_DOT_Y = -4.686084221046758e-07       [km**2/s]
CX_DOT_Z = 2.484949578400095e-07        [km**2/s]
CX_DOT_X_DOT = 4.296022805587290e-10     [km**2/s**2]
CY_DOT_X = -2.211832501084875e-07        [km**2/s]
CY_DOT_Y = -2.864186892102733e-07        [km**2/s]
CY_DOT_Z = 1.798098699846038e-07        [km**2/s]
CY_DOT_X_DOT = 2.608899201686016e-10     [km**2/s**2]
CY_DOT_Y_DOT = 1.767514756338532e-10     [km**2/s**2]
CZ_DOT_X = -3.041346050686871e-07        [km**2/s]
CZ_DOT_Y = -4.989496988610662e-07        [km**2/s]
CZ_DOT_Z = 3.540310904497689e-07        [km**2/s]
CZ_DOT_X_DOT = 1.869263192954590e-10     [km**2/s**2]
CZ_DOT_Y_DOT = 1.008862586240695e-10     [km**2/s**2]
CZ_DOT_Z_DOT = 6.224444338635500e-10     [km**2/s**2]
USER_DEFINED_EARTH_MODEL = WGS-84
    
```

Fig. D-4: OPM file example with optional Keplerian elements, covariance matrix, and a user-defined parameter

OPM example in XML:

Fig. D-5 contains an example of an OPM in Extensible Markup Language (XML) format.

```

<?xml version="1.0" encoding="UTF-8"?>
<opm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
  xsi:noNamespaceSchemaLocation="http://sanaregistry.org/r/ndmxml/ndmxml-1.0-master.xsd"
  id="CCSDS_OPM_VERS" version="3.0">

  <header>
    <COMMENT>THIS IS AN XML VERSION OF THE OPM</COMMENT>
    <CREATION_DATE>2001-11-06T09:23:57</CREATION_DATE>
    <ORIGINATOR>JAXA</ORIGINATOR>
    <MESSAGE_ID>OPM 201113719185</MESSAGE_ID>
  </header>
  <body>
    <segment>
      <metadata>
        <COMMENT>GEOCENTRIC, CARTESIAN, EARTH FIXED</COMMENT>
        <OBJECT_NAME>OSPREY 5</OBJECT_NAME>
        <OBJECT_ID>1998-999A</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>ITRF1997</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
      </metadata>
      <data>
        <stateVector>
          <EPOCH>1996-12-18T14:28:15.1172</EPOCH>
          <X>6503.514000</X>
          <Y>1239.647000</Y>
          <Z>-717.490000</Z>
          <X_DOT>-0.873160</X_DOT>
          <Y_DOT>8.740420</Y_DOT>
          <Z_DOT>-4.191076</Z_DOT>
        </stateVector>
        <spacecraftParameters>
          <MASS>3000.000000</MASS>
          <SOLAR_RAD_AREA>18.770000</SOLAR_RAD_AREA>
          <SOLAR_RAD_COEFF>1.000000</SOLAR_RAD_COEFF>
          <DRAG_AREA>18.770000</DRAG_AREA>
          <DRAG_COEFF>2.500000</DRAG_COEFF>
        </spacecraftParameters>
        <covarianceMatrix>
          <COV_REF_FRAME>ITRF1997</COV_REF_FRAME>
          <CX_X>0.316</CX_X>
          <CY_X>0.722</CY_X>
          <CY_Y>0.518</CY_Y>
          <CZ_X>0.202</CZ_X>
          <CZ_Y>0.715</CZ_Y>
          <CZ_Z>0.002</CZ_Z>
          <CX_DOT_X>0.912</CX_DOT_X>
          <CX_DOT_Y>0.306</CX_DOT_Y>
          <CX_DOT_Z>0.276</CX_DOT_Z>
          <CX_DOT_X_DOT>0.797</CX_DOT_X_DOT>
          <CY_DOT_X>0.562</CY_DOT_X>
          <CY_DOT_Y>0.899</CY_DOT_Y>
          <CY_DOT_Z>0.022</CY_DOT_Z>
          <CY_DOT_X_DOT>0.079</CY_DOT_X_DOT>
          <CY_DOT_Y_DOT>0.415</CY_DOT_Y_DOT>
          <CZ_DOT_X>0.245</CZ_DOT_X>
          <CZ_DOT_Y>0.965</CZ_DOT_Y>
          <CZ_DOT_Z>0.950</CZ_DOT_Z>
          <CZ_DOT_X_DOT>0.435</CZ_DOT_X_DOT>
          <CZ_DOT_Y_DOT>0.621</CZ_DOT_Y_DOT>
          <CZ_DOT_Z_DOT>0.991</CZ_DOT_Z_DOT>
        </covarianceMatrix>
      </data>
    </segment>
  </body>
</opm>

```

Fig. D-5: OPM file example in XML format

ANNEX E

OMM EXAMPLES

(INFORMATIVE)

The following are examples of Orbit Mean-Element Messages (OMMs). All of these examples are based on the TLE shown in Fig. E-1.

```
GOES 9 [P]
1 23581U 95025A 07064.44075725 -.00000113 00000-0 10000-3 0 9250
2 23581 3.0539 81.7939 0005013 249.2363 150.1602 1.00273272 43169
```

Fig. E-1: Example Two Line Element Set (TLE)

OMM examples in KVN:

The following figures are examples of OMMs in Keyword Value Notation (KVN) format .

```
CCSDS_OMM_VERS = 3.0
CREATION_DATE = 2007-065T16:00:00
ORIGINATOR = NOAA
MESSAGE_ID = OMM 201113719185

OBJECT_NAME = GOES 9
OBJECT_ID = 1995-025A
CENTER_NAME = EARTH
REF_FRAME = TEME
TIME_SYSTEM = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH = 2007-064T10:34:41.4264
MEAN_MOTION = 1.00273272
ECCENTRICITY = 0.0005013
INCLINATION = 3.0539
RA_OF_ASC_NODE = 81.7939
ARG_OF_PERICENTER = 249.2363
MEAN_ANOMALY = 150.1602
GM = 398600.8
EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH = 4316
BSTAR = 0.0001
MEAN_MOTION_DOT = -0.00000113
MEAN_MOTION_DDOT = 0.0
```

Fig. E-2: OMM file example without covariance matrix

```

CCSDS_OMM_VERS = 3.0
CREATION_DATE = 2007-065T16:00:00
ORIGINATOR = NOAA

OBJECT_NAME = GOES 9
OBJECT_ID = 1995-025A
CENTER_NAME = EARTH
REF_FRAME = TEME
TIME_SYSTEM = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH = 2007-064T10:34:41.4264
MEAN_MOTION = 1.00273272
ECCENTRICITY = 0.0005013
INCLINATION = 3.0539
RA_OF_ASC_NODE = 81.7939
ARG_OF_PERICENTER = 249.2363
MEAN_ANOMALY = 150.1602
GM = 398600.8

EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH = 4316
BSTAR = 0.0001
MEAN_MOTION_DOT = -0.00000113
MEAN_MOTION_DDOT = 0.0

COV_REF_FRAME = TEME
CX_X = 3.331349476038534e-04
CY_X = 4.618927349220216e-04
CY_Y = 6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z = 3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z = 2.484949578400095e-07
CX_DOT_X_DOT = 4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z = 1.798098699846038e-07
CY_DOT_X_DOT = 2.608899201686016e-10
CY_DOT_Y_DOT = 1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z = 3.540310904497689e-07
CZ_DOT_X_DOT = 1.869263192954590e-10
CZ_DOT_Y_DOT = 1.008862586240695e-10
CZ_DOT_Z_DOT = 6.224444338635500e-10
    
```

Fig. E-3: OMM file example with covariance matrix

```

CCSDS_OMM_VERS = 3.0
CREATION_DATE  = 2007-065T16:00:00
ORIGINATOR     = NOAA

OBJECT_NAME    = GOES 9
OBJECT_ID      = 1995-025A
CENTER_NAME    = EARTH
REF_FRAME      = TEME
TIME_SYSTEM    = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH          = 2007-064T10:34:41.4264
MEAN_MOTION    = 1.00273272      [rev/day]
ECCENTRICITY   = 0.0005013
INCLINATION    = 3.0539          [deg]
RA_OF_ASC_NODE = 81.7939         [deg]
ARG_OF_PERICENTER = 249.2363     [deg]
MEAN_ANOMALY   = 150.1602       [deg]
GM             = 398600.8        [km**3/s**2]
EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID   = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH   = 4316
BSTAR          = 0.0001          [1/ER]
MEAN_MOTION_DOT = -0.00000113   [rev/day**2]
MEAN_MOTION_DDOT = 0.0           [rev/day**3]

USER_DEFINED_EARTH_MODEL = WGS-84

```

Fig. E-4: OMM with units and a user-defined parameter

OMM example in XML:

Fig. E-5 contains an example of an OMM in Extensible Markup Language (XML) format.

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

<?xml version="1.0" encoding="UTF-8"?>
<omm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
      xsi:noNamespaceSchemaLocation="http://sanaregistry.org/r/ndmxml/ndmxml-1.0-master.xsd"
      id="CCSDS_OMM_VERS" version="3.0">
  <header>
    <COMMENT> THIS IS AN XML VERSION OF THE OMM </COMMENT>
    <CREATION_DATE>2007-065T16:00:00</CREATION_DATE>
    <ORIGINATOR>NOAA</ORIGINATOR>
    <MESSAGE_ID> OMM 201113719185</MESSAGE_ID>
  </header>

  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>GOES-9</OBJECT_NAME>
        <OBJECT_ID>1995-025A</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>TEME</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <MEAN_ELEMENT_THEORY>TLE</MEAN_ELEMENT_THEORY>
      </metadata>

      <data>
        <meanElements>
          <EPOCH>2007-064T10:34:41.4264</EPOCH>
          <MEAN_MOTION>1.00273272</MEAN_MOTION>
          <ECCENTRICITY>0.0005013</ECCENTRICITY>
          <INCLINATION>3.0539</INCLINATION>
          <RA_OF_ASC_NODE>81.7939</RA_OF_ASC_NODE>
          <ARG_OF_PERICENTER>249.2363</ARG_OF_PERICENTER>
          <MEAN_ANOMALY>150.1602</MEAN_ANOMALY>
          <GM>398600.8</GM>
        </meanElements>
        <tleParameters>
          <NORAD_CAT_ID>23581</NORAD_CAT_ID>
          <ELEMENT_SET_NO>0925</ELEMENT_SET_NO>
          <REV_AT_EPOCH>4316</REV_AT_EPOCH>
          <BSTAR>0.0001</BSTAR>
          <MEAN_MOTION_DOT>-0.00000113</MEAN_MOTION_DOT>
          <MEAN_MOTION_DDOT>0.0</MEAN_MOTION_DDOT>
        </tleParameters>
        <covarianceMatrix>
          <COV_REF_FRAME>TEME</COV_REF_FRAME>
          <CX_X>3.331349476038534e-04</CX_X>
          <CY_X>4.618927349220216e-04</CY_X>
          <CY_Y>6.782421679971363e-04</CY_Y>
          <CZ_X>-3.070007847730449e-04</CZ_X>
          <CZ_Y>-4.221234189514228e-04</CZ_Y>
          <CZ_Z>3.231931992380369e-04</CZ_Z>
          <CX_DOT_X>-3.349365033922630e-07</CX_DOT_X>
          <CX_DOT_Y>-4.686084221046758e-07</CX_DOT_Y>
          <CX_DOT_Z>2.484949578400095e-07</CX_DOT_Z>
          <CX_DOT_X_DOT>4.296022805587290e-10</CX_DOT_X_DOT>
          <CY_DOT_X>-2.211832501084875e-07</CY_DOT_X>
          <CY_DOT_Y>-2.864186892102733e-07</CY_DOT_Y>
          <CY_DOT_Z>1.798098699846038e-07</CY_DOT_Z>
          <CY_DOT_X_DOT>2.608899201686016e-10</CY_DOT_X_DOT>
          <CY_DOT_Y_DOT>1.767514756338532e-10</CY_DOT_Y_DOT>
          <CZ_DOT_X>-3.041346050686871e-07</CZ_DOT_X>
          <CZ_DOT_Y>-4.989496988610662e-07</CZ_DOT_Y>
          <CZ_DOT_Z>3.540310904497689e-07</CZ_DOT_Z>
          <CZ_DOT_X_DOT>1.869263192954590e-10</CZ_DOT_X_DOT>
          <CZ_DOT_Y_DOT>1.008862586240695e-10</CZ_DOT_Y_DOT>
          <CZ_DOT_Z_DOT>6.224444338635500e-10</CZ_DOT_Z_DOT>
        </covarianceMatrix>
      </data>
    </segment>
  </body>
</omm>

```

Fig. E-5: OMM file example in XML format

ANNEX F

OEM EXAMPLES

(INFORMATIVE)

The following are examples of Orbit Ephemeris Messages (OEMs).

OEM examples in KVN:

The following figures are examples of OEMs in Keyword Value Notation (KVN) format. Some ephemeris data lines have been omitted to save space.

```

CCSDS_OEM_VERS = 3.0
CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = J2000
TIME_SYSTEM      = UTC
START_TIME       = 1996-12-18T12:00:00.331
USEABLE_START_TIME = 1996-12-18T12:10:00.331
USEABLE_STOP_TIME  = 1996-12-28T21:23:00.331
STOP_TIME        = 1996-12-28T21:28:00.331
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This file was produced by M.R. Pigs, OSAR NAV/JPL, 1996NOV 04. It is
COMMENT to be used for DSN scheduling purposes only.

1996-12-18T12:00:00.331  2789.619 -280.045 -1746.755  4.73372 -2.49586 -1.04195
1996-12-18T12:01:00.331  2783.419 -308.143 -1877.071  5.18604 -2.42124 -1.99608
1996-12-18T12:02:00.331  2776.033 -336.859 -2008.682  5.63678 -2.33951 -1.94687

    < intervening data records omitted here >

1996-12-28T21:28:00.331 -3881.024  563.959 -682.773  -3.28827 -3.66735  1.63861

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = J2000
TIME_SYSTEM      = UTC
START_TIME       = 1996-12-28T21:29:07.267
USEABLE_START_TIME = 1996-12-28T22:08:02.5
USEABLE_STOP_TIME  = 1996-12-30T01:18:02.5
STOP_TIME        = 1996-12-30T01:28:02.267
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This block begins after trajectory correction maneuver TCM-3.

1996-12-28T21:29:07.267 -2432.166 -063.042 1742.754  7.33702 -3.495867 -1.041945
1996-12-28T21:59:02.267 -2445.234 -878.141 1873.073  1.86043 -3.421256 -0.996366
1996-12-28T22:00:02.267 -2458.079 -683.858 2007.684  6.36786 -3.339563 -0.946654

    < intervening data records omitted here >

1996-12-30T01:28:02.267 2164.375 1115.811 -688.131  -3.53328 -2.88452  0.88535

```

Fig. F-1: OEM Example with No Acceleration, No Covariance

```

CCSDS_OEM_VERS = 3.0

COMMENT  OEM WITH OPTIONAL ACCELERATIONS

CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 2000-028A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = J2000
TIME_SYSTEM      = UTC
START_TIME       = 1996-12-18T12:00:00.331
USEABLE_START_TIME = 1996-12-18T12:10:00.331
USEABLE_STOP_TIME  = 1996-12-28T21:23:00.331
STOP_TIME        = 1996-12-28T21:28:00.331
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT  This file was produced by M.R. Somebody, MSOO NAV/JPL, 2000 NOV 04. It is
COMMENT  to be used for DSN scheduling purposes only.

1996-12-18T12:00:00.331  2789.6 -280.0 -1746.8  4.73 -2.50 -1.04  0.008 0.001 -0.159
1996-12-18T12:01:00.331  2783.4 -308.1 -1877.1  5.19 -2.42 -2.00  0.008 0.001  0.001
1996-12-18T12:02:00.331  2776.0 -336.9 -2008.7  5.64 -2.34 -1.95  0.008 0.001  0.159

    < intervening data records omitted here >

1996-12-28T21:28:00.331 -3881.0  564.0 -682.8 -3.29 -3.67  1.64  -0.003 0.000  0.000
    
```

Fig. F-2: OEM Example with Optional Accelerations

```

CCSDS_OEM_VERS = 3.0
CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL
MESSAGE_ID = OEM 201113719185
META_START
OBJECT_NAME           = MARS GLOBAL SURVEYOR
OBJECT_ID             = 1996-062A
CENTER_NAME           = MARS BARYCENTER
REF_FRAME             = J2000
TIME_SYSTEM           = UTC
START_TIME            = 1996-12-28T21:29:07.267
USEABLE_START_TIME    = 1996-12-28T22:08:02.5
USEABLE_STOP_TIME     = 1996-12-30T01:18:02.5
STOP_TIME             = 1996-12-30T01:28:02.267
INTERPOLATION         = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This block begins after trajectory correction maneuver TCM-3.

1996-12-28T21:29:07.267 -2432.166 -063.042 1742.754 7.33702 -3.495867 -1.041945
1996-12-28T21:59:02.267 -2445.234 -878.141 1873.073 1.86043 -3.421256 -0.996366
1996-12-28T22:00:02.267 -2458.079 -683.858 2007.684 6.36786 -3.339563 -0.946654

    < intervening data records omitted here >

1996-12-30T01:28:02.267 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535

COVARIANCE_START
EPOCH = 1996-12-28T21:29:07.267
COV_REF_FRAME = J2000
 3.3313494e-04
 4.6189273e-04 6.7824216e-04
-3.0700078e-04 -4.2212341e-04 3.2319319e-04
-3.3493650e-07 -4.6860842e-07 2.4849495e-07 4.2960228e-10
-2.2118325e-07 -2.8641868e-07 1.7980986e-07 2.6088992e-10 1.7675147e-10
-3.0413460e-07 -4.9894969e-07 3.5403109e-07 1.8692631e-10 1.0088625e-10 6.2244443e-10

EPOCH = 1996-12-29T21:00:00
COV_REF_FRAME = J2000
 3.4424505e-04
 4.5078162e-04 6.8935327e-04
-3.0600067e-04 -4.1101230e-04 3.3420420e-04
-3.2382549e-07 -4.5750731e-07 2.3738384e-07 4.3071339e-10
-2.1007214e-07 -2.7530757e-07 1.6870875e-07 2.5077881e-10 1.8786258e-10
-3.0302350e-07 -4.8783858e-07 3.4302008e-07 1.7581520e-10 1.0077514e-10 6.2244443e-10
COVARIANCE_STOP
    
```

Fig. F-3: OEM Example with Optional Covariance Matrices

OEM example in XML:

Fig. F-4 contains an example of an Orbit Ephemeris Message in Extensible Markup Language (XML) format.

```

<?xml version="1.0" encoding="UTF-8"?>
<oem xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
     xsi:noNamespaceSchemaLocation="http://sanaregistry.org/r/ndmxml/ndmxml-1.0-master.xsd"
     id="CCSDS_OEM_VERS" version="3.0">

  <header>
    <COMMENT>OEM WITH OPTIONAL ACCELERATIONS</COMMENT>
    <CREATION_DATE>1996-11-04T17:22:31</CREATION_DATE>
    <ORIGINATOR>NASA/JPL</ORIGINATOR>
    <MESSAGE_ID>OEM 201113719185</MESSAGE_ID>
  </header>
    
```

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

<body>
  <segment>
    <metadata>
      <OBJECT_NAME>MARS GLOBAL SURVEYOR</OBJECT_NAME>
      <OBJECT_ID>2000-028A</OBJECT_ID>
      <CENTER_NAME>MARS BARYCENTER</CENTER_NAME>
      <REF_FRAME>J2000</REF_FRAME>
      <TIME_SYSTEM>UTC</TIME_SYSTEM>
      <START_TIME>1996-12-18T12:00:00.331</START_TIME>
      <USEABLE_START_TIME>1996-12-18T12:10:00.331</USEABLE_START_TIME>
      <USEABLE_STOP_TIME>1996-12-28T21:23:00.331</USEABLE_STOP_TIME>
      <STOP_TIME>1996-12-28T21:28:00.331</STOP_TIME>
      <INTERPOLATION>HERMITE</INTERPOLATION>
      <INTERPOLATION_DEGREE>7</INTERPOLATION_DEGREE>
    </metadata>
    <data>
      <COMMENT>Produced by M.R. Sombedody, MSO NAV/JPL, 1996 OCT 11. It is</COMMENT>
      <COMMENT>to be used for DSN scheduling purposes only.</COMMENT>
      <stateVector>
        <EPOCH>1996-12-18T12:00:00.331</EPOCH>
        <X>2789.6</X>
        <Y>-280.0</Y>
        <Z>-1746.8</Z>
        <X_DOT>4.73</X_DOT>
        <Y_DOT>-2.50</Y_DOT>
        <Z_DOT>-1.04</Z_DOT>
        <X_DDOT>0.008</X_DDOT>
        <Y_DDOT>0.001</Y_DDOT>
        <Z_DDOT>-0.159</Z_DDOT>
      </stateVector>
      <stateVector>
        <EPOCH>1996-12-18T12:01:00.331</EPOCH>
        <X>2783.4</X>
        <Y>-308.1</Y>
        <Z>-1877.1</Z>
        <X_DOT>5.19</X_DOT>
        <Y_DOT>-2.42</Y_DOT>
        <Z_DOT>-2.00</Z_DOT>
        <X_DDOT>0.008</X_DDOT>
        <Y_DDOT>0.001</Y_DDOT>
        <Z_DDOT>0.001</Z_DDOT>
      </stateVector>
      <stateVector>
        <EPOCH>1996-12-18T12:02:00.331</EPOCH>
        <X>2776.0</X>
        <Y>-336.9</Y>
        <Z>-2008.7</Z>
        <X_DOT>5.64</X_DOT>
        <Y_DOT>-2.34</Y_DOT>
        <Z_DOT>-1.95</Z_DOT>
        <X_DDOT>0.008</X_DDOT>
        <Y_DDOT>0.001</Y_DDOT>
        <Z_DDOT>0.159</Z_DDOT>
      </stateVector>
      <stateVector>
        <EPOCH>1996-12-28T21:28:00.331</EPOCH>
        <X>-3881.0</X>
        <Y>564.0</Y>
        <Z>-682.8</Z>
        <X_DOT>-3.29</X_DOT>
        <Y_DOT>-3.67</Y_DOT>
        <Z_DOT>1.64</Z_DOT>
        <X_DDOT>-0.003</X_DDOT>
        <Y_DDOT>0.000</Y_DDOT>
        <Z_DDOT>0.000</Z_DDOT>
      </stateVector>
      <covarianceMatrix>
        <EPOCH>1996-12-28T22:28:00.331</EPOCH>
        <COV_REF_FRAME>ITRF1997</COV_REF_FRAME>
        <CX_X>0.316</CX_X>
        <CY_X>0.722</CY_X>
    </data>
  </segment>
</body>

```

```
<CY_Y>0.518</CY_Y>
<CZ_X>0.202</CZ_X>
<CZ_Y>0.715</CZ_Y>
<CZ_Z>0.002</CZ_Z>
<CX_DOT_X>0.912</CX_DOT_X>
<CX_DOT_Y>0.306</CX_DOT_Y>
<CX_DOT_Z>0.276</CX_DOT_Z>
<CX_DOT_X_DOT>0.797</CX_DOT_X_DOT>
<CY_DOT_X>0.562</CY_DOT_X>
<CY_DOT_Y>0.899</CY_DOT_Y>
<CY_DOT_Z>0.022</CY_DOT_Z>
<CY_DOT_X_DOT>0.079</CY_DOT_X_DOT>
<CY_DOT_Y_DOT>0.415</CY_DOT_Y_DOT>
<CZ_DOT_X>0.245</CZ_DOT_X>
<CZ_DOT_Y>0.965</CZ_DOT_Y>
<CZ_DOT_Z>0.950</CZ_DOT_Z>
<CZ_DOT_X_DOT>0.435</CZ_DOT_X_DOT>
<CZ_DOT_Y_DOT>0.621</CZ_DOT_Y_DOT>
<CZ_DOT_Z_DOT>0.991</CZ_DOT_Z_DOT>
</covarianceMatrix>
</data>
</segment>
</body>
</oem>
```

Fig. F-4: OEM file example in XML format

ANNEX G

**OCM EXAMPLES AND ASSOCIATED SUPPLEMENTARY
INFORMATION****(INFORMATIVE)**

The following are examples of Orbit Comprehensive Messages (OCMs).

OCM examples in KVN:

The following figures are examples of OCMs in Keyword Value Notation (KVN) format. The first has only a time history of orbital states and constitutes a minimal content OCM. The second includes space object characteristics and perturbations specifications; the third includes a time series of maneuvers, a time history of Cartesian position and velocity orbit states, followed by a time history of Keplerian elements; and the fourth includes a time series of covariance matrices.

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR = JAXA
META_START
TIME_SYSTEM = UTC
EPOCH_TZERO = 1998-12-18T14:28:15.1172
META_STOP
ORB_START
ORB_BASIS=SIMULATED
ORB_REF_FRAME=ITRF2000
ORB_UNITS= [km, km, km, km/s, km/s, km/s]
  0.0  2854.5  -2916.2  -5360.7  5.90  4.86  0.52  0.0037  -0.0038  -0.0070
 120.0  5478.6   434.3  -3862.5  2.50  5.87  4.29  0.0072   0.0006  -0.0051
 240.0  4146.0  -1655.8  -5038.3  4.80  5.58  2.16  0.0054  -0.0022  -0.0066
< intervening data records omitted here >
86400  -1553.4  -4848.7  -4406.5  6.73  1.01  -3.53  -0.002  -0.0063  -0.0058
ORB_STOP

```

Fig. G-1: Simple/Succinct OCM File example with only Cartesian ephemeris. In this example, CENTER_NAME defaults to EARTH and orbit type (ORB_TYPE) to CARTPV. In this example, at the expense of readability, KVN values are unaligned to minimize message storage and transmission size.

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0
COMMENT This OCM reflects the latest conditions post-maneuver A67Z
COMMENT This example shows the specification of multiple comment lines
CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR = JAXA
MESSAGE_ID = OCM 201113719185

META_START
OBJECT_NAME = OSPREY 5
INTERNATIONAL_DESIGNATOR = 1998-999A

ORIGINATOR_POC = R. Rabbit
ORIGINATOR_POSITION = Flight Dynamics Mission Design Lead
ORIGINATOR_PHONE = (719)555-1234

TECH_POC = Mr. Rodgers
TECH_PHONE = (719)555-1234
TECH_ADDRESS = email@email.XXX

TIME_SYSTEM = UT1
EPOCH_TZERO = 1998-12-18T00:00:00.0000

TAIMUTC_AT_TZERO = 36 [s]
UT1MUTC_AT_TZERO = .357 [s]
META_STOP

ORB_START
COMMENT GEOCENTRIC, CARTESIAN, EARTH FIXED
COMMENT THIS IS MY SECOND COMMENT LINE
ORB_BASIS=PREDICTED
ORB_REF_FRAME = EFG
ORB_TYPE = CARTPV
ORB_UNITS = [km, km, km, km/s, km/s, km/s]
1998-12-18T14:28:25.1172 2854.533 -2916.187 -5360.774 5.688 4.652 0.520
ORB_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
WET_MASS = 100.0 [kg]
OEB_Q1 = 0.03123
OEB_Q2 = 0.78543
OEB_Q3 = 0.39158
OEB_QC = 0.47832
OEB_MAX = 2.0 [m]
OEB_INT = 1.0 [m]
OEB_MIN = 0.5 [m]
AREA_ALONG_OEB_MAX = 0.15 [m**2]
AREA_ALONG_OEB_INT = 0.3 [m**2]
AREA_ALONG_OEB_MIN = 0.5 [m**2]
PHYS_STOP

PERT_START
COMMENT Perturbations Specification:
ATMOSPHERIC_MODEL = NRLMSIS00
GRAVITY_MODEL = EGM-96: 36D 360
GM = 398600.4415 [km**3/s**2]
N_BODY_PERTURBATIONS = MOON, SUN
FIXED_GEOMAG_KP = 12.0
FIXED_F10P7 = 105.0
FIXED_F10P7_MEAN = 120.0
PERT_STOP

USER_START
USER_DEFINED_CONSOLE_POC = MAXWELL RAFERTY
USER_STOP

```

Fig. G-2: OCM example with space object characteristics and perturbations

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR = JAXA
META_START
EPOCH_TZERO = 1998-12-18T14:28:15.1172
META_STOP

ORB_START
COMMENT ORBIT EPHEMERIS INCORPORATING DEPLOYMENTS AND MANEUVERS (BELOW)
ORB_BASIS=PREDICTED
ORB_REF_FRAME = TOD_EARTH
ORB_FRAME_EPOCH = 1998-12-18T14:28:15.1172
ORB_TYPE = CARTPVA
ORB_UNITS = [km, km, km, km/s, km/s, km/s, km/s**2, km/s**2, km/s**2]
1998-12-18T14:36:05.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -0.0070
1998-12-18T14:38:05.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072 0.0006 -0.0051
1998-12-18T14:40:05.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -0.0022 -0.0066
< intervening data records omitted here >
1998-12-19T14:36:05.0 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -0.0063 -0.0058
ORB_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
DRAG_CONST_AREA = 10.00 [m**2]
DRAG_COEFF_NOM = 2.300
DRAG_UNCERTAINTY = 5.0
WET_MASS = 200.0 [kg]
SRP_CONST_AREA = 4.00
SOLAR_RAD_COEFF = 1.300
PHYS_STOP

MAN_START
COMMENT = Ten 1kg objects deployed at 1 m/s from 190kg host over 100 s time
COMMENT = 20 deg off of back-track direction
MAN_ID = E_W_20160305B
MAN_BASIS = CANDIDATE
MAN_DEVICE_ID = DEPLOY
MAN_PURPOSE = DEPLOY
MAN_REF_FRAME = RSW_ROTATING
MAN_COMPOSITION = TIME_RELATIVE, DEPLOY_ID, DEPLOY_DV_X, DEPLOY_DV_Y, DEPLOY_DV_Z,
<cont.>
DEPLOY_MASS, DEPLOY_DV_SIGMA, DEPLOY_DV_RATIO, DEPLOY_DV_CDA
MAN_UNITS = [n/a, km/s, km/s, km/s, kg, %, n/a, m**2]
500.0 CUBESAT_10 2.8773E-4 -9.3969E-4 1.8491E-4 -1.0 5.0 -0.005025 0.033
510.0 CUBESAT_11 1.4208E-4 -9.3969E-4 3.1111E-4 -1.0 5.0 -0.005051 0.033
520.0 CUBESAT_12 -4.8670E-5 -9.3969E-4 3.3854E-4 -1.0 5.0 -0.005076 0.033
530.0 CUBESAT_13 -2.2398E-4 -9.3969E-4 2.5848E-4 -1.0 5.0 -0.005102 0.033
540.0 CUBESAT_14 -3.2817E-4 -9.3969E-4 9.6360E-5 -1.0 5.0 -0.005128 0.033
550.0 CUBESAT_15 -3.2817E-4 -9.3969E-4 -9.6360E-5 -1.0 5.0 -0.005154 0.033
560.0 CUBESAT_16 -2.2398E-4 -9.3969E-4 -2.5848E-4 -1.0 5.0 -0.005181 0.033
570.0 CUBESAT_17 -4.8670E-5 -9.3969E-4 -3.3854E-4 -1.0 5.0 -0.005208 0.033
580.0 CUBESAT_18 1.4208E-4 -9.3969E-4 -3.1111E-4 -1.0 5.0 -0.005236 0.033
590.0 CUBESAT_19 2.8773E-4 -9.3969E-4 -1.8491E-4 -1.0 5.0 -0.005263 0.033
MAN_STOP

MAN_START
COMMENT = 100 s of 0.5N +in-track thrust w/effic η=0.95, Isp=300s, 5% 1-sigma
error
COMMENT = NOTE that this is OCM specifies a compound maneuver, with
COMMENT = occurring simultaneously with low-level host platform thrusting.
MAN_ID = E_W_20160305B
MAN_BASIS = CANDIDATE
MAN_DEVICE_ID = THR_01
MAN_PURPOSE = ORBIT
MAN_REF_FRAME = RSW_ROTATING
MAN_COMPOSITION = TIME_ABSOLUTE, TIME_RELATIVE, MAN_DURA, THR_X, THR_Y, THR_Z, THR_SIGMA,
THR_INTERP, THR_ISP, THR_EFFIC
MAN_UNITS = [s, N, N, N, %, n/a, s, n/a]
1998-12-18T14:36:35.1172 500.0 100.0 0.0 0.5 0.0 5.0 ON 300.0 0.95
MAN_STOP

```

```

PERT_START
COMMENT Perturbations specification
GM          = 398600.4415      [km**3/s**2]
PERT_STOP

OD_START
COMMENT Orbit Determination information
OD_ID       = OD #10059
OD_PREV_ID  = OD #10058
OD_METHOD   = SF: ODTK
OD_EPOCH   = 2001-11-06T11:17:33
OBS_USED    = 273
TRACKS_USED = 91
OD_STOP

```

Fig. G-3: OCM example with deployed objects and low-level thrusting maneuver during deployment to make “string-of-pearls” deployment

NOTE 1: This example is aligned with a multi-deployment simulation for a future mission, thus the specific mission is not identified.

NOTE 2: The demarcation “<cont.>” indicates that this single line is shown, for display purposes only, in multiple lines. This demarcation shall not be used in the actual OCM message, as all specified content shall be provided on the same line as its matching time tag.

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR = JAXA

META_START
OBJECT_NAME = OSPREY 5
OBJECT_DESIGNATOR= 98765
INTERNATIONAL_DESIGNATOR = 1998-999A
CATALOG_NAME = CSPOC

TIME_SYSTEM = UTC
EPOCH_TZERO = 1998-12-18T14:28:15.1172
META_STOP

ORB_START
ORB_BASIS = PREDICTED
ORB_REF_FRAME = TOD_EARTH
ORB_FRAME_EPOCH = 1998-12-18T14:28:15.1172
USEABLE_START_TIME= 1998-12-18T14:28:15.1172
USEABLE_STOP_TIME= 1998-12-18T14:28:45.1172
ORB_TYPE = CARTPVA
ORB_UNITS = [km, km, km, km/s, km/s, km/s, km/s**2, km/s**2, km/s**2]
0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -0.0070
120.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072 0.0006 -0.0051
240.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -0.0022 -0.0066
< intervening data records omitted here >
86400 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -0.0063 -0.0058
ORB_STOP

ORB_START
ORB_BASIS = DETERMINED_OD
ORB_REF_FRAME = J2000
ORB_TYPE = KEPLERIAN
ORB_UNITS = [km, n/a, deg, deg, deg, deg]
0.000000 6600.0 .03 28.5 50.0 30.0 10.0
10.000000 6600.0 .03 28.5 50.0 30.0 10.1
20.000000 6600.0 .03 28.5 50.0 30.0 10.2
< intervening data records omitted here >
500.000000 6600.0 .03 28.5 50.0 30.0 35.0
ORB_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
DRAG_CONST_AREA = 10.00 [m**2]
DRAG_COEFF_NOM = 2.300
WET_MASS = 100.0 [kg]
SRP_CONST_AREA = 4.00
SOLAR_RAD_COEFF = 1.300
PHYS_STOP

MAN_START
COMMENT = 200 s of 10N thrust (in-track transitioning to radial)
COMMENT = w/effic η=0.95, Isp=300s, 5% 1-sigma error
MAN_ID = E_W_20160305B
MAN_BASIS = CANDIDATE
MAN_DEVICE_ID = THR_01
MAN_PURPOSE = ORBIT
MAN_REF_FRAME = RSW_ROTATING
MAN_COMPOSITION = TIME_RELATIVE, MAN_DURA, THR_X, THR_Y, THR_Z, THR_SIGMA, THR_INTERP,
THR_ISP, THR_EFFIC
MAN_UNITS = [s, N, N, N, %, n/a, s, n/a]
500.0 100.0 0.0 10.0 0.0 5.0 ON 300.0 0.95
600.0 100.0 10.0 0.0 0.0 5.0 OFF 300.0 0.95
MAN_STOP

PERT_START
COMMENT Perturbations specification
GM = 398600.4415 [km**3/s**2]
PERT_STOP

OD_START

```

COMMENT	Orbit Determination information
OD_ID	= OD #10059
OD_PREV_ID	= OD #10058
OD_METHOD	= BWLS: BAHN
OD_EPOCH	= 2001-11-06T11:17:33
OBS_USED	= 273
TRACKS_USED	= 91
OD_STOP	

Fig. G-4: OCM example with multiple orbit time histories, a maneuver, OD, Cartesian & Keplerian ephemeris

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS = 3.0

CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR = JAXA

META_START
OBJECT_NAME = OSPREY 5
INTERNATIONAL_DESIGNATOR = 1998-999A
TIME_SYSTEM = UTC
EPOCH_TZERO = 1998-12-18T14:28:15.1172
META_STOP

ORB_START
COMMENT GEOCENTRIC, CARTESIAN, EARTH FIXED
ORB_BASIS = PREDICTED
CENTER_NAME = EARTH
ORB_REF_FRAME = TOD_EARTH
ORB_FRAME_EPOCH = 1998-12-18T14:28:15.1172
ORB_TYPE = CARTPVA

    0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -0.0070
    120.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072 0.0006 -0.0051
    240.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -0.0022 -0.0066
< intervening data records omitted here >
86400 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -0.0063 -0.0058
ORB_STOP

PHYS_START
COMMENT Spacecraft Physical Characteristics:
DRAG_CONST_AREA = 10.000 [m**2]
DRAG_COEFF_NOM = 2.300
WET_MASS = 1913.000 [kg]
SRP_CONST_AREA = 10.000 [m**2]
SOLAR_RAD_COEFF = 1.300
PHYS_STOP

COV_START
COV_BASIS = PREDICTED
COV_REF_FRAME = J2000
COV_TYPE = ADBARV
COV_UNITS = [deg, deg, deg, deg, km, km/s]
10.00 3.331349e-04 4.618927e-04 6.782421e-04 -3.070007e-04 -4.221234e-04 <cont.>
3.231931e-04 -3.349365e-07 -4.686084e-07 2.484949e-07 4.296022e-10 <cont.>
-2.211832e-07 -2.864186e-07 1.798098e-07 2.608899e-10 1.767514e-10 <cont.>
-3.041346e-07 -4.989496e-07 3.540310e-07 1.869263e-10 1.008862e-10 6.224444e-10
< ... intervening data records omitted here ... >
20.0 3.442450e-04 4.507816e-04 6.893532e-04 -3.060006e-04 -4.110123e-04 <cont.>
3.342042e-04 -3.238254e-07 -4.575073e-07 2.373838e-07 4.307133e-10 <cont.>
-2.100721e-07 -2.753075e-07 1.687087e-07 2.507788e-10 1.878625e-10 <cont.>
-3.030235e-07 -4.878385e-07 3.430200e-07 1.758152e-10 1.007751e-10 6.224444e-10
COV_STOP

COV_START
COV_BASIS = PREDICTED
COV_REF_FRAME = FIXED_EARTH
COV_TYPE = CARTP
COV_UNITS = [km, km, km]
1998-12-18T14:31:35.1172 3.331349e-04 4.618927e-04 6.782421e-04 -3.070007e-04 <cont.>
-4.221234e-04 3.231931e-04
COV_STOP

PERT_START
COMMENT Perturbations specification
GM = 398600.4415 [km**3/s**2]
PERT_STOP
    
```

Fig. G-5: OCM example with Covariance Matrix

NOTE: The demarcation “<cont.>” indicates that this single line is shown, for display purposes only, in multiple lines. This demarcation shall not be used in the actual OCM message, as all specified content shall be provided on the same line as its matching time tag.

OCM example in XML:

The following is an example of an Orbit Comprehensive Message in Extensible Markup Language (XML) format.

```

<?xml version="1.0" encoding="UTF-8"?>
<ocm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
  xsi:noNamespaceSchemaLocation="file:///Y:/Desktop/ccsds-ndmxml/ndmxml-
  1.0.x-schemas-unqualified/ndmxml-1.0-master.xsd"
  id="CCSDS_OCM_VERS" version="3.0">
  <header>
  <COMMENT>ODM V.3 Example G-2</COMMENT>
  <COMMENT>OCM example with space object characteristics and
  perturbations.</COMMENT>
  <COMMENT>This OCM reflects the latest conditions post-maneuver A67Z</COMMENT>
  <COMMENT>This example shows the specification of multiple comment
  lines</COMMENT>
  <CREATION_DATE>1998-11-06T09:23:57</CREATION_DATE>
  <ORIGINATOR>JAXA</ORIGINATOR>
  <MESSAGE_ID> OCM 201113719185</MESSAGE_ID> </header>
  <body>
  <segment>
  <metadata>
  <OBJECT_NAME>OSPREY 5</OBJECT_NAME>
  <INTERNATIONAL_DESIGNATOR>1998-999A</INTERNATIONAL_DESIGNATOR>
  <ORIGINATOR_POC>R. Rabbit</ORIGINATOR_POC>
  <ORIGINATOR_POSITION>Flight Dynamics Mission Design
  Lead</ORIGINATOR_POSITION>
  <ORIGINATOR_PHONE>(719)555-1234</ORIGINATOR_PHONE>
  <TECH_POC>Mr. Rodgers</TECH_POC>
  <TECH_PHONE>(719)555-1234</TECH_PHONE>
  <TECH_ADDRESS>email@email.XXX</TECH_ADDRESS>
  <TIME_SYSTEM>UT1</TIME_SYSTEM>
  <EPOCH_TZERO>1998-12-18T00:00:00.0000</EPOCH_TZERO>
  <TAIMUTC_AT_TZERO units="s">36</TAIMUTC_AT_TZERO>
  <UT1MUTC_AT_TZERO units="s">.357</UT1MUTC_AT_TZERO>
  </metadata>
  <data>
  <orb>
  <COMMENT>GEOCENTRIC, CARTESIAN, EARTH FIXED</COMMENT>
  <COMMENT>THIS IS MY SECOND COMMENT LINE</COMMENT>
  <ORB_BASIS>PREDICTED</ORB_BASIS >
  <ORB_REF_FRAME>EFG</ORB_REF_FRAME>
  <ORB_TYPE>CARTPVA</ORB_TYPE>
  <orbLine>0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -0.0038 -
  0.0070</orbLine>
  </orb>
  <phys>
  <COMMENT>Spacecraft Physical Characteristics</COMMENT>
  <WET_MASS units="kg">100.0</WET_MASS>
  <OEB_Q1>0.03123</OEB_Q1>
  <OEB_Q2>0.78543</OEB_Q2>
  <OEB_Q3>0.39158</OEB_Q3>
  <OEB_QC>0.47832</OEB_QC>
  <OEB_MAX units="m">2.0</OEB_MAX>
  <OEB_INT units="m">1.0</OEB_INT>
  <OEB_MIN units="m">0.5</OEB_MIN>
  <AREA_ALONG_OEB_MAX units="m**2">0.15</AREA_ALONG_OEB_MAX>
  <AREA_ALONG_OEB_INT units="m**2">0.3</AREA_ALONG_OEB_INT>
  <AREA_ALONG_OEB_MIN units="m**2">0.5</AREA_ALONG_OEB_MIN>
  </phys>
  <pert>
  <COMMENT>Perturbations Specification</COMMENT>
  <ATMOSPHERIC_MODEL>NRLMSIS00</ATMOSPHERIC_MODEL>
  <GRAVITY_MODEL>EGM-96: 36D 360</GRAVITY_MODEL>
  <GM units="km**3/s**2">398600.4415</GM>
  <N_BODY_PERTURBATIONS>MOON, SUN</N_BODY_PERTURBATIONS>
  <FIXED_GEOG_KP>12.0</FIXED_GEOG_KP>
  <FIXED_F10P7>105.0</FIXED_F10P7>
  <FIXED_F10P7_MEAN>120.0</FIXED_F10P7_MEAN>
  </pert>
  <userDef>
  <USER_DEFINED parameter="EARTH_MODEL">WGS-84</USER_DEFINED>
  </userDef>
  
```

```
</data>  
</segment>  
</body>  
</ocm>
```

Fig. G-6: OCM example in Extensible Markup Language (XML) format.

ANNEX H

AGGREGATING MULTIPLE ODMS IN A SINGLE NDM XML FILE

(INFORMATIVE)

The following example illustrates how multiple Orbit Data Messages can be aggregated in a single XML file using the Navigation Data Message (NDM) "combined instantiation" schema.

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

<?xml version="1.0" encoding="UTF-8"?>
<ndm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml/ndmxml-1.0-
master.xsd">
  <omm id="CCSDS_OMM_VERS" version="2.0">
    <header>
      <COMMENT>GENERATED VIA SPACE-TRACK.ORG API</COMMENT>
      <CREATION_DATE>2020-05-16T14:00:01</CREATION_DATE>
      <ORIGINATOR>18 SPCS</ORIGINATOR>
    </header>
    <body>
      <segment>
        <metadata>
          <OBJECT_NAME>STARLINK-1073</OBJECT_NAME>
          <OBJECT_ID>2020-001A</OBJECT_ID>
          <CENTER_NAME>EARTH</CENTER_NAME>
          <REF_FRAME>TEME</REF_FRAME>
          <TIME_SYSTEM>UTC</TIME_SYSTEM>
          <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
        </metadata>
        <data>
          <meanElements>
            <EPOCH>2020-05-16T14:00:01</EPOCH>
            <MEAN_MOTION>15.05566242</MEAN_MOTION>
            <ECCENTRICITY>0.0001225</ECCENTRICITY>
            <INCLINATION>52.9981</INCLINATION>
            <RA_OF_ASC_NODE>157.6133</RA_OF_ASC_NODE>
            <ARG_OF_PERICENTER>93.35</ARG_OF_PERICENTER>
            <MEAN_ANOMALY>295.8599</MEAN_ANOMALY>
          </meanElements>
          <tleParameters>
            <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
            <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
            <NORAD_CAT_ID>44914</NORAD_CAT_ID>
            <ELEMENT_SET_NO>999</ELEMENT_SET_NO>
            <REV_AT_EPOCH>176</REV_AT_EPOCH>
            <BSTAR>0.00057678</BSTAR>
            <MEAN_MOTION_DOT>0.00008131</MEAN_MOTION_DOT>
            <MEAN_MOTION_DDOT>0</MEAN_MOTION_DDOT>
          </tleParameters>
          <userDefinedParameters>
            <USER_DEFINED parameter="TLE_LINE0">0 STARLINK-1073</USER_DEFINED>
            <USER_DEFINED parameter="TLE_LINE1">
              1 44914U 20001A 20137.58334491 +.00008131 +00000-0 +57678-3 0 9994
            </USER_DEFINED>
            <USER_DEFINED parameter="TLE_LINE2">
              2 44914 052.9981 157.6133 0001225 093.3500 295.8599
              15.05566242001761
            </USER_DEFINED>
          </userDefinedParameters>
        </data>
      </segment>
    </body>
  </omm>
  <omm id="CCSDS_OMM_VERS" version="2.0">
    <header>
      <COMMENT>GENERATED VIA SPACE-TRACK.ORG API</COMMENT>
      <CREATION_DATE>2020-05-16T14:00:01</CREATION_DATE>
      <ORIGINATOR>18 SPCS</ORIGINATOR>
    </header>
    <body>
      <segment>
        <metadata>
          <OBJECT_NAME>STARLINK-1084</OBJECT_NAME>
          <OBJECT_ID>2020-001B</OBJECT_ID>
          <CENTER_NAME>EARTH</CENTER_NAME>
          <REF_FRAME>TEME</REF_FRAME>
          <TIME_SYSTEM>UTC</TIME_SYSTEM>
          <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>

```

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

</metadata>
<data>
  <meanElements>
    <EPOCH>2020-05-16T14:00:01</EPOCH>
    <MEAN_MOTION>15.05603711</MEAN_MOTION>
    <ECCENTRICITY>0.000126</ECCENTRICITY>
    <INCLINATION>52.9991</INCLINATION>
    <RA_OF_ASC_NODE>157.611</RA_OF_ASC_NODE>
    <ARG_OF_PERICENTER>75.9191</ARG_OF_PERICENTER>
    <MEAN_ANOMALY>151.2104</MEAN_ANOMALY>
  </meanElements>
  <tleParameters>
    <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
    <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
    <NORAD_CAT_ID>44915</NORAD_CAT_ID>
    <ELEMENT_SET_NO>999</ELEMENT_SET_NO>
    <REV_AT_EPOCH>1986</REV_AT_EPOCH>
    <BSTAR>0.0006442</BSTAR>
    <MEAN_MOTION_DOT>0.00009125</MEAN_MOTION_DOT>
    <MEAN_MOTION_DDOT>0</MEAN_MOTION_DDOT>
  </tleParameters>
  <userDefinedParameters>
    <USER_DEFINED parameter="TLE_LINE0">0 STARLINK-1084</USER_DEFINED>
    <USER_DEFINED parameter="TLE_LINE1">
      1 44915U 20001B 20137.58334491 +.00009125 +00000-0 +64420-3 0 9992
    </USER_DEFINED>
    <USER_DEFINED parameter="TLE_LINE2">
      2 44915 052.9991 157.6110 0001260 075.9191 151.2104
15.05603711019869
    </USER_DEFINED>
  </userDefinedParameters>
</data>
</segment>
</body>
</omml>
<omml id="CCSDS_OMM_VERS" version="2.0">
  <header>
    <COMMENT>GENERATED VIA SPACE-TRACK.ORG API</COMMENT>
    <CREATION_DATE>2020-05-16T14:00:01</CREATION_DATE>
    <ORIGINATOR>18 SPCS</ORIGINATOR>
  </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>STARLINK-1097</OBJECT_NAME>
        <OBJECT_ID>2020-001C</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>TEME</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
      </metadata>
      <data>
        <meanElements>
          <EPOCH>2020-05-16T14:00:01</EPOCH>
          <MEAN_MOTION>15.0559315</MEAN_MOTION>
          <ECCENTRICITY>0.0001361</ECCENTRICITY>
          <INCLINATION>52.999</INCLINATION>
          <RA_OF_ASC_NODE>157.6123</RA_OF_ASC_NODE>
          <ARG_OF_PERICENTER>94.2334</ARG_OF_PERICENTER>
          <MEAN_ANOMALY>78.9025</MEAN_ANOMALY>
        </meanElements>
        <tleParameters>
          <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
          <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
          <NORAD_CAT_ID>44916</NORAD_CAT_ID>
          <ELEMENT_SET_NO>999</ELEMENT_SET_NO>
          <REV_AT_EPOCH>1986</REV_AT_EPOCH>
          <BSTAR>0.00072742</BSTAR>
          <MEAN_MOTION_DOT>0.00010329</MEAN_MOTION_DOT>
          <MEAN_MOTION_DDOT>0</MEAN_MOTION_DDOT>
        </tleParameters>

```

```

    <userDefinedParameters>
      <USER_DEFINED parameter="TLE_LINE0">0 STARLINK-1097</USER_DEFINED>
      <USER_DEFINED parameter="TLE_LINE1">
        1 44916U 20001C 20137.58334491 +.00010329 +00000-0 +72742-3 0 9997
      </USER_DEFINED>
      <USER_DEFINED parameter="TLE_LINE2">
        2 44916 052.9990 157.6123 0001361 094.2334 078.9025
15.05559315019865
      </USER_DEFINED>
    </userDefinedParameters>
  </data>
</segment>
</body>
</omdm>
</ndm>

```

Fig. H-1: Aggregating multiple ODMs into a single NDM file

```

<?xml version="1.0" encoding="UTF-8"?>
<ndm xmlns:xsi="http://www.w3.org/2001/XMLSchema-instance"
  xsi:noNamespaceSchemaLocation="https://sanaregistry.org/r/ndmxml/ndmxml-1.0-
master.xsd">

<COMMENT>This example combines an OPM, OMM, OEM, and OCM in a single Navigation
Data Message XML.</COMMENT>
<COMMENT>NOTE: In this case the messages are unrelated, but show how, in
principle, related </COMMENT>
<COMMENT>messages could be combined to satisfy a specific use case.</COMMENT>

<opm id="CCSDS_OPM_VERS" version="3.0">
  <header>
    <COMMENT>Example Orbit Parameter Message</COMMENT>
    <COMMENT>The OPM includes a set of orbital elements at time t, and an
impulsive maneuver design with MAN_EPOCH_IGNITION = t.</COMMENT>
    <CREATION_DATE>2009-05-18T13:06:00</CREATION_DATE>
    <ORIGINATOR>NASA</ORIGINATOR>
  </header>
  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>SOHO</OBJECT_NAME>
        <OBJECT_ID>2009-000A</OBJECT_ID>
        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>EME2000</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
      </metadata>
      <data>
        <stateVector>
          <EPOCH>2009-05-12T15:30:00</EPOCH>
          <X>687754.36358524448</X>
          <Y>941287.85553999904</Y>
          <Z>520080.81101286016</Z>
          <X_DOT>-0.41396551860286032</X_DOT>
          <Y_DOT>0.29174462217893128</Y_DOT>
          <Z_DOT>0.11756781919443198</Z_DOT>
        </stateVector>
        <maneuverParameters>
          <COMMENT>The below contains information for the impulsive
maneuver.</COMMENT>
          <MAN_EPOCH_IGNITION>2009-05-12T15:30:00</MAN_EPOCH_IGNITION>
          <MAN_DURATION>0</MAN_DURATION>
          <MAN_DELTA_MASS>-0.0252069575402913408</MAN_DELTA_MASS>
          <MAN_REF_FRAME>EME2000</MAN_REF_FRAME>
          <MAN_DV_1>0.000028562811624</MAN_DV_1>
          <MAN_DV_2>3.0883529021E-7</MAN_DV_2>
          <MAN_DV_3>1.4646782842E-8</MAN_DV_3>
        </maneuverParameters>
      </data>
    </segment>
  </body>
</opm>

<omm id="CCSDS_OMM_VERS" version="3.0">
  <header>
    <COMMENT> Example Orbit Mean-Elements Message</COMMENT>
    <COMMENT> ODM PROTOTYPING FOR CCSDS 502.0-P-1.1 - TEST NUMBER 4</COMMENT>
    <CREATION_DATE>2009-091T16:00:00</CREATION_DATE>
    <ORIGINATOR>CNES</ORIGINATOR>
  </header>

  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>ISS-ZARYA</OBJECT_NAME>
        <OBJECT_ID>1998-067A</OBJECT_ID>

```

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

        <CENTER_NAME>EARTH</CENTER_NAME>
        <REF_FRAME>TEME</REF_FRAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <MEAN_ELEMENT_THEORY>SGP4</MEAN_ELEMENT_THEORY>
    </metadata>
    <data>
        <COMMENT>USAF SGP4 IS THE ONLY PROPAGATOR THAT SHOULD BE USED FOR
THIS DATA</COMMENT>
        <meanElements>
            <EPOCH>2009-087T11:58:23.211264</EPOCH>
            <MEAN_MOTION>15.71784080</MEAN_MOTION>
            <ECCENTRICITY>0.0009738</ECCENTRICITY>
            <INCLINATION>51.6429</INCLINATION>
            <RA_OF_ASC_NODE>0.4159</RA_OF_ASC_NODE>
            <ARG_OF_PERICENTER>166.5533</ARG_OF_PERICENTER>
            <MEAN_ANOMALY>354.9076</MEAN_ANOMALY>
        </meanElements>
        <tleParameters>
            <EPHEMERIS_TYPE>0</EPHEMERIS_TYPE>
            <CLASSIFICATION_TYPE>U</CLASSIFICATION_TYPE>
            <NORAD_CAT_ID>25544</NORAD_CAT_ID>
            <ELEMENT_SET_NO>557</ELEMENT_SET_NO>
            <REV_AT_EPOCH>59325</REV_AT_EPOCH>
            <BSTAR>-0.21414E-3</BSTAR>
            <MEAN_MOTION_DOT>0.00027894</MEAN_MOTION_DOT>
            <MEAN_MOTION_DDOT>0.0</MEAN_MOTION_DDOT>
        </tleParameters>
    </data>
</segment>
</body>
</omm>

<oem id="CCSDS_OEM_VERS" version="3.0">
    <header>
        <COMMENT>Example Orbit Ephemeris Message</COMMENT>
        <CREATION_DATE>2009-02-05T10:09:12</CREATION_DATE>
        <ORIGINATOR>JAXA</ORIGINATOR>
    </header>
    <body>
        <segment>
            <metadata>
                <OBJECT_NAME>SELENE</OBJECT_NAME>
                <OBJECT_ID>131</OBJECT_ID>
                <CENTER_NAME>EARTH</CENTER_NAME>
                <REF_FRAME>EME2000</REF_FRAME>
                <TIME_SYSTEM>UTC</TIME_SYSTEM>
                <START_TIME>2007-09-14T10:43:00.000000</START_TIME>
                <STOP_TIME>2007-09-14T10:47:00.000000</STOP_TIME>
                <INTERPOLATION>Hermite</INTERPOLATION>
                <INTERPOLATION_DEGREE>7</INTERPOLATION_DEGREE>
            </metadata>
            <data>
                <stateVector>
                    <EPOCH>2007-09-14T10:43:00.000000</EPOCH>
                    <X>1.336886031675458E+03</X>
                    <Y>9.480146566872711E+04</Y>
                    <Z>3.266088528177066E+04</Z>
                    <X_DOT>-6.305293974707967E-01</X_DOT>
                    <Y_DOT>1.848864859994888E+00</Y_DOT>
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                </stateVector>
                <stateVector>
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                    <X>1.299053338219702E+03</X>
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                    <Z>3.271712789822781E+04</Z>
                    <X_DOT>-6.305602161673365E-01</X_DOT>
                    <Y_DOT>1.846618917569059E+00</Y_DOT>
                    <Z_DOT>9.369900978163187E-01</Z_DOT>
                </stateVector>
            </data>
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    </body>
</oem>

```

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

<stateVector>
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  <Y>9.502306005072314E+04</Y>
  <Z>3.277332412355368E+04</Z>
  <X_DOT>-6.305900282238581E-01</X_DOT>
  <Y_DOT>1.844378570838882E+00</Y_DOT>
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  <Y>9.513365569386665E+04</Y>
  <Z>3.282947404774766E+04</Z>
  <X_DOT>-6.306188407720047E-01</X_DOT>
  <Y_DOT>1.842143793832282E+00</Y_DOT>
  <Z_DOT>9.354467257772656E-01</Z_DOT>
</stateVector>
<stateVector>
  <EPOCH>2007-09-14T10:47:00.000000</EPOCH>
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  <Y>9.524411741683590E+04</Y>
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</stateVector>
</data>
</segment>
</body>
</oem>

<ocm id="CCSDS_OCM_VERS" version="3.0">

  <header>
    <COMMENT>Example G-4 from OCM P2.40</COMMENT>
    <COMMENT>OCM example with multiple orbit time histories, a maneuver, OD,
    Cartesian and Keplerian ephemeris</COMMENT>
    <CREATION_DATE>1998-11-06T09:23:57</CREATION_DATE>
    <ORIGINATOR>JAXA</ORIGINATOR>
  </header>

  <body>
    <segment>
      <metadata>
        <OBJECT_NAME>OSPREY 5</OBJECT_NAME>
        <OBJECT_DESIGNATOR>98765</OBJECT_DESIGNATOR>
        <INTERNATIONAL_DESIGNATOR>1998-999A</INTERNATIONAL_DESIGNATOR>
        <CATALOG_NAME>CSPOC</CATALOG_NAME>
        <TIME_SYSTEM>UTC</TIME_SYSTEM>
        <EPOCH_TZERO>1998-12-18T14:28:15.1172</EPOCH_TZERO>
      </metadata>

      <data>
        <orb>
          <ORB_BASIS>PREDICTED</ORB_BASIS>
          <ORB_REF_FRAME>TOD EARTH</ORB_REF_FRAME>
          <ORB_FRAME_EPOCH>1998-12-18T14:28:15.1172</ORB_FRAME_EPOCH>
          <USEABLE_START_TIME>1998-12-18T14:28:15.1172</USEABLE_START_TIME>
          <USEABLE_STOP_TIME>1998-12-18T14:28:45.1172</USEABLE_STOP_TIME>
          <ORB_TYPE>CARTPVA</ORB_TYPE>
          <ORB_UNITS>[km, km, km, km/s, km/s, km/s, km/s**2, km/s**2,
          km/s**2]</ORB_UNITS>
          <orbLine> 0.0 2854.5 -2916.2 -5360.7 5.90 4.86 0.52 0.0037 -
          0.0038 -0.0070</orbLine>
          <orbLine>120.0 5478.6 434.3 -3862.5 2.50 5.87 4.29 0.0072
          0.0006 -0.0051</orbLine>
          <orbLine>240.0 4146.0 -1655.8 -5038.3 4.80 5.58 2.16 0.0054 -
          0.0022 -0.0066</orbLine>
        </orb>
      </data>
    </segment>
  </body>
</oem>

```

```

        <orbLine>500.0 -1553.4 -4848.7 -4406.5 6.73 1.01 -3.53 -0.002 -
0.0063 -0.0058</orbLine>
    </orb>

    <orb>
    <ORB_BASIS>DETERMINED_OD</ORB_BASIS>
    <ORB_REF_FRAME>J2000</ORB_REF_FRAME>
    <ORB_TYPE>KEPLERIAN</ORB_TYPE>
    <ORB_UNITS>[km, n/a, deg, deg, deg, deg]</ORB_UNITS>
    <orbLine>0.000000 6600.0 .03 28.5 50.0 30.0 10.0</orbLine>
    <orbLine>120.000000 6600.0 .03 28.5 50.0 30.0 10.1</orbLine>
    <orbLine>240.000000 6600.0 .03 28.5 50.0 30.0 10.2</orbLine>
    <orbLine>500.000000 6600.0 .03 28.5 50.0 30.0 35.0</orbLine>
    </orb>

    <phys>
    <COMMENT>Spacecraft Physical Characteristics</COMMENT>
    <DRAG_CONST_AREA units="m**2">10.00</DRAG_CONST_AREA>
    <DRAG_COEFF_NOM>2.3</DRAG_COEFF_NOM>
    <MASS units="kg">100.0</MASS>
    <SRP_CONST_AREA units="m**2">4</SRP_CONST_AREA>
    <SOLAR_RAD_COEFF >1.3</SOLAR_RAD_COEFF >
    </phys>

    <man>
    <COMMENT>200 s of 10N thrust (in-track transitioning to
radial)</COMMENT>
    <COMMENT>w/effic η=0.95, Isp=300s, 5% 1-sigma error</COMMENT>
    <MAN_ID>E_W_20160305B</MAN_ID>
    <MAN_BASIS>CANDIDATE</MAN_BASIS>
    <MAN_DEVICE_ID>THR_01</MAN_DEVICE_ID>
    <MAN_PURPOSE>ORBIT</MAN_PURPOSE>
    <MAN_REF_FRAME>RSW_ROTATING</MAN_REF_FRAME>
    <MAN_COMPOSITION>TIME_RELATIVE MAN_DURA, THR_X, THR_Y, THR_Z,
THR_SIGMA, THR_INTERP, THR_ISP, THR_EFFIC</MAN_COMPOSITION>
    <MAN_UNITS>[s, N, N, N, %, n/a, s, n/a]</MAN_UNITS>
    <manLine>500.0 100.0 0.0 10.0 0.0 5.0 ON 300.0 0.95</manLine>
    <manLine>600.0 100.0 10.0 0.0 0.0 5.0 OFF 300.0 0.95</manLine>
    </man>

    <pert>
    <COMMENT>Perturbations specification</COMMENT>
    <GM>398600.4415</GM>
    </pert>

    <od>
    <COMMENT> Orbit Determination information</COMMENT>
    <OD_ID>OD #10059</OD_ID>
    <OD_PREV_ID>OD #10058</OD_PREV_ID>
    <OD_METHOD>BWLS: BAHN</OD_METHOD>
    <OD_EPOCH>2001-11-06T11:17:33</OD_EPOCH>
    <OBS_USED>273</OBS_USED>
    <TRACKS_USED>91</TRACKS_USED>
    </od>

    </data>
  </segment>
</body>
</ocm>
</ndm>

```

Fig. H-2: Aggregating OPM, OMM, OEM, and OCM in a single Navigation Data Message XML.

ANNEX I

ABBREVIATIONS AND ACRONYMS**(INFORMATIVE)**

ASCII	American Standard Code for Information Interchange
CCSDS	Consultative Committee for Space Data Systems
DSST	Draper Semi-Analytic Satellite Theory
EGM	Earth Gravitational Model, Earth Geopotential Model
EOP	Earth Orientation Parameters
GPS	Global Positioning System
IAU	International Astronomical Union
ICD	Interface Control Document
ICRF	International Celestial Reference Frame
IEC	International Electro-technical Commission
IERS	International Earth Rotation and Reference Systems Service
IIRV	Improved Inter-Range Vector
ISO	International Standards Organization
ITRF	International Terrestrial Reference Frame
GRC	Greenwich Rotating Coordinate Frame
J2000	Earth Mean Equator and Equinox of J2000 (Julian Date 2000)
KVN	Keyword = Value Notation
LTM	Lower Triangular Matrix
NORAD	North American Aerospace Defense Command
OD	Orbit Determination
ODM	Orbit Data Message
OEB	Optimally-Encompassing Box

OEM	Orbit Ephemeris Message
OCM	Orbit Comprehensive Message
OMM	Orbit Mean-Elements Message
OPM	Orbit Parameter Message
RTN	Radial, Transverse (along-track), and Normal
S/C	Spacecraft
SGP4	US Air Force Simplified General Perturbations No. 4
SPK	Satellite, Planetary Kernel
TAI	International Atomic Time
TCB	Barycentric Coordinate Time
TDB	Barycentric Dynamical Time
TDR	True of Date Rotating
TDT	Terrestrial Dynamical Time (see also ‘TT’)
TEME	True Equator Mean Equinox
TLE	Two Line Element
TOD	True Equator and Equinox of Date
TT	Terrestrial Dynamical Time (see also ‘TDT’)
USM	Universal Semi-analytical Method
UTC	Coordinated Universal Time
W3C	World Wide Web Consortium
WGS	World Geodetic System
XML	Extensible Markup Language

ANNEX J

RATIONALE FOR ORBIT DATA MESSAGES

(INFORMATIVE)

J1 OVERVIEW

This annex presents the rationale behind the design of each message. It may help the application engineer to select a suitable message.

A specification of requirements agreed to by all parties is essential to focus design and to ensure the product meets the needs of the Member Agencies and satellite operators. There are many ways of organizing requirements, but the categorization of requirements is not as important as the agreement to a sufficiently comprehensive set. In this section the requirements are organized into three categories:

- a) **Primary Requirements:** These are the most elementary and necessary requirements. They would exist no matter the context in which the CCSDS is operating, i.e., regardless of pre-existing conditions within the CCSDS, its Member Agencies, or other independent users.
- b) **Heritage Requirements:** These are additional requirements that derive from pre-existing Member Agency or other independent user requirements, conditions or needs. Ultimately these carry the same weight as the Primary Requirements. This Recommended Standard reflects heritage requirements pertaining to some of the CCSDS Areas' home institutions collected during the preparation of the document; it does not speculate on heritage requirements that could arise from other sources. Corrections and/or additions to these requirements are expected during future updates.
- c) **Desirable Characteristics:** These are not requirements, but they are felt to be important or useful features of the Recommended Standard.

J2 PRIMARY REQUIREMENTS ACCEPTED BY THE ORBIT DATA MESSAGES

J2.1 PRIMARY REQUIREMENTS

#	Requirement	OPM?	OMM?	OEM?	OCM?
P1	Data must be provided in digital form (computer file).	Y	Y	Y	Y
P2	The file specification must not require of the receiving exchange partner the separate application of, or modeling of, spacecraft dynamics or gravitational force models, or integration or propagation.	N	N	Y	Y
P3	The interface must facilitate the receiver of the message to generate a six-component Cartesian state vector (position and velocity) at any required epoch.	Y	Y	Y	Y
P4	State vector information must be provided in a reference frame that is clearly identified and unambiguous.	Y	Y	Y	Y
P5	Identification of the object and the center(s) of motion must be clearly identified and unambiguous.	Y	Y	Y	Y
P6	Time measurements (time stamps, or epochs) must be provided in a commonly used, clearly specified system.	Y	Y	Y	Y
P7	The time bounds of the ephemeris must be unambiguously specified.	N/A	N/A	Y	Y
P8	The Recommended Standard must provide for clear specification of units of measure.	Y	Y	Y	Y
P9	Files must be readily ported between, and useable within, 'all' computational environments in use by Member Agencies.	Y	Y	Y	Y
P10	Files must have means of being uniquely identified and clearly annotated. The file name alone is considered insufficient for this purpose.	Y	Y	Y	Y
P11	File name syntax and length must not violate computer constraints for those computing environments in use by Member Agencies.	Y	Y	Y	Y
P12	A means to convey information about the uncertainty of the state shall be provided.	Y	Y	Y	Y

J2.2 HERITAGE REQUIREMENTS

#	Requirement	OPM?	OMM?	OEM?	OCM?
H1	Ephemeris data is reliably convertible into the SPICE SPK (NASA) format (reference [N-4]) and IIRV (NASA) format (reference [N-5]) using a standard, multi-mission, unsupervised pipeline process. A complete ephemeris, not subject to integration or propagation by the customer, must be provided.	N	N	Y	Y
H2	Ephemeris data provided for scheduling or operations (metric predicts) is to be certified by the providing Agency as correct and complete for the intended purpose. The receiving Agency cannot provide evaluation, trajectory propagation or other usability services.	N	N	Y	Y

#	Requirement	OPM?	OMM?	OEM?	OCM?
H3	The ODM shall provide a mechanism by which messages may be uniquely identified and clearly annotated. Facilitates discussion between the recipient and the message originator, should that be necessary.	Y	Y	Y	Y
H4	The ODM shall provide a mechanism by which maneuvers may be uniquely identified and clearly annotated. Facilitates discussion between the recipient and the message originator, should that be necessary.	N	N	N	Y
H5	The Recommended Standard is, or includes, an ASCII format.	Y	Y	Y	Y
H6	The Recommended Standard does not require software supplied by other Agencies.	Y	N	Y	Y

J2.3 DESIRABLE CHARACTERISTICS

#	Requirement	OPM?	OMM?	OEM?	OCM?
DC1	The Recommended Standard applies to non-traditional objects, such as landers, rovers, balloons, and natural bodies (asteroids, comets).	Y	N	Y	Y
DC2	The Recommended Standard allows state vectors to be provided in other than the traditional J2000 inertial reference frame; one example is the International Astronomical Union (IAU) Mars body-fixed frame. (In such a case, provision or ready availability of supplemental information needed to transform data into a standard frame must be arranged.)	Y	Y	Y	Y
DC3	The Recommended Standard is extensible with no disruption to existing users/uses.	Y	Y	Y	Y
DC4	The Recommended Standard is consistent with, and ideally a part of, ephemeris products and processes used for other space science purposes.	Y	Y	Y	Y
DC5	The Recommended Standard is as consistent as reasonable with any related CCSDS ephemeris Recommended Standards used for earth-to-spacecraft or spacecraft-to-spacecraft applications.	Y	Y	Y	Y

J2.4 APPLICABILITY OF CRITERIA TO MESSAGE OPTIONS

The selection of one particular message will depend on the optimization criteria in the given application. Section J2.5 compares the four recommended messages in terms of the relevant selection criteria identified by the CCSDS:

J2.5 APPLICABILITY OF THE CRITERIA TO ORBIT DATA MESSAGES

Criteria	Definition	Applicable to OPM?	Applicable to OMM?	Applicable to OEM?	Applicable to OCM?
Modeling Fidelity	Permits modeling of any dynamic perturbation to the trajectory.	N	N	Y	Y
Human Readability	Provides easily readable message corresponding to widely used orbit representation.	Y	Y	Y	Y
Remote Body Extensibility	Permits use for assets on remote solar system bodies.	Y	N	Y	Y
Lander/Rover Compatibility	Permits exchange of non-orbit trajectories.	N	N	Y	Y

J3 INCREASING ORBIT PROPAGATION FIDELITY OF AN OPM OR OMM

Some OPM or OMM users may desire/require a higher fidelity propagation of the state vector or Keplerian elements. A higher fidelity technique may be desired/required to minimize inconsistencies in predictions generated by diverse, often operator-unique propagation schemes. Nominally the OPM and OMM are engineered only for low- to medium-fidelity orbit propagation. However, with the inclusion of additional context information, it is possible for users to provide data that could be used to provide a relatively higher fidelity orbit propagation. For this relatively higher fidelity orbit propagation, a much greater amount of ancillary information regarding spacecraft properties and dynamical models should be provided. Higher fidelity orbit propagations may be useful in special studies such as orbit conjunction studies.

Spacecraft orbit determination is a stochastic estimation problem; observations are inherently uncertain, and not all of the phenomena that influence satellite motion are clearly discernible. State vectors and Keplerian elements with their respective covariances are best propagated with models that include the same forces and phenomena that were used for determining the orbit. Including this information in an OPM or OMM allows exchange partners to compare the results of their respective orbit propagations.

With additional context information, the OPM and OMM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and propagating the orbits of active satellites, inactive man-made objects, and near-Earth debris fragments. The additional information facilitates dynamic modeling of any user's approach to conservative and non-conservative phenomena.

The primary vehicle for the provision of additional optional ancillary information to be used when propagating an OPM or OMM is the COMMENT mechanism. Alternatively, the 'USER_DEFINED_' keyword prefix may be used, though this usage is not encouraged.

J4 SERVICES RELATED TO THE DIFFERENT ORBIT DATA MESSAGE FORMATS

The different orbit data messages have been distinguished by the self-interpretability of the messages. The different services that can be achieved without special arrangements between users of the CCSDS orbit data messages are listed in Table J4.1

J4.1 SERVICES AVAILABLE WITH ORBIT DATA MESSAGES

Service	Definition	Applicable to OPM?	Applicable to OMM?	Applicable to OEM?	Applicable to OCM?
Absolute Orbit Interpretation	State availability at specific times for use in additional computations (geometry, event detection, etc.).	Y	Y	Y	Y
Relative Orbit Interpretation	Trajectory comparison and differencing for events based on the same time source.	Only at time specified at Epoch	Only at time specified at Epoch	Y	Y

ANNEX K

ITEMS FOR AN INTERFACE CONTROL DOCUMENT (ICD)**(INFORMATIVE)****K1 STANDARD ICD ITEMS**

In several places in this document there are references to items which should be specified in an ICD between participants that supplements an exchange of ephemeris data. The ICD should be jointly produced by both participants in a cross-support involving the transfer of ephemeris data. This annex compiles those recommendations into a single section. Although the Orbit Data Messages described in this document may at times be used in situations in which participants have not negotiated ICDs, they should be developed and negotiated whenever specified in this Recommended Standard.

Item	Section
1) Detailed description of any user-defined OPM, OMM, OEM, and OCM parameters used.	3.2, 4.2, 5.2, 6.2
2) Detailed description of any exceptions for keyword values not drawn from the SANA registry (sanaregistry.org).	Annex B, Section B6
3) Specific information security interoperability provisions that apply between agencies.	Annex M

ANNEX L

CHANGES IN ODM VERSION 3**(INFORMATIVE)**

This annex lists the differences between ODM 2.0 and ODM 3.0. The differences are divided into those which affect the content of one or more of the orbit data messages, and those which only affect the document.

L1 CHANGES IN THE MESSAGES

1. The Orbit Comprehensive Message (OCM) was added to provide better support for ISO Technical Committee 20, Subcommittee 14 objectives (see Section 4).
2. MESSAGE_ID was added to the OPM, OMM, and OEM to provide better satisfaction of requirement P10 (identification and annotation of messages).

L2 CHANGES IN THE DOCUMENT

1. A new CCSDS repository for normative keyword values for navigation messages has been created at the SANA Registry, accessible on the Internet at: https://sanaregistry.org/r/navigation_standard_normative_annexes/. See Annex B for details on the affected keywords and links to the content.
2. Several annexes were added. Some are required by CCSDS rule changes, and some are for the provision of supplementary material.
3. Examples for OPM, OMM, and OEM that formerly appeared in Sections 3, 4, and 5 respectively have been moved to Informative Annexes.
4. The "Checklist ICD" that was added in ODM Version 2 has been greatly reduced. This Checklist ICD, intended to convey information that the OPM, OEM and OMM did not address such as third-body perturbations, solar pressure model, solid tides, ocean tides, Earth albedo and polar motion, has now been replaced by the material that can be specified in the Orbit Comprehensive Message.

ANNEX M

SECURITY, SANA, AND PATENT CONSIDERATIONS**M1 SECURITY CONSIDERATIONS****M1.1 ANALYSIS OF SECURITY CONSIDERATIONS**

This Section presents the results of an analysis of security considerations applied to the technologies specified in this Recommended Standard.

M1.2 CONSEQUENCES OF NOT APPLYING SECURITY TO THE TECHNOLOGY

The consequences of not applying security to the systems and networks on which this Recommended Standard is implemented could include potential loss, corruption, and theft of data. Because these messages are used in preparing pointing and frequency predicts used during spacecraft commanding, and may also be used in collision avoidance analyses, the consequences of not applying security to the systems and networks on which this Recommended Standard is implemented could include compromise or loss of the mission if malicious tampering of a particularly severe nature occurs.

M1.3 POTENTIAL THREATS AND ATTACK SCENARIOS

Potential threats or attack scenarios include, but are not limited to, (a) unauthorized access to the programs/processes that generate and interpret the messages, (b) unauthorized access to the messages during transmission between exchange partners and (c) modification of the messages between partners. Protection from unauthorized access during transmission is especially important if the mission utilizes open ground networks, such as the Internet, to provide ground-station connectivity for the exchange of data formatted in compliance with this Recommended Standard. It is strongly recommended that potential threats or attack scenarios applicable to the systems and networks on which this Recommended Standard is implemented be addressed by the management of those systems and networks.

M1.4 DATA PRIVACY

Privacy of data formatted in compliance with the specifications of this Recommended Standard should be assured by the systems and networks on which this Recommended Standard is implemented.

M1.5 DATA INTEGRITY

Integrity of data formatted in compliance with the specifications of this Recommended Standard should be assured by the systems and networks on which this Recommended Standard is implemented.

M1.6 AUTHENTICATION OF COMMUNICATING ENTITIES

Authentication of communicating entities involved in the transport of data which complies with the specifications of this Recommended Standard should be provided by the systems and networks on which this Recommended Standard is implemented.

M1.7 DATA TRANSFER BETWEEN COMMUNICATING ENTITIES

The transfer of data formatted in compliance with this Recommended Standard between communicating entities should be accomplished via secure mechanisms approved by the Information Technology Security functionaries of exchange participants.

M1.8 CONTROL OF ACCESS TO RESOURCES

Control of access to resources should be managed by the systems upon which originator formatting and recipient processing are performed.

M1.9 AUDITING OF RESOURCE USAGE

Auditing of resource usage should be handled by the management of systems and networks on which this Recommended Standard is implemented.

M1.10 UNAUTHORIZED ACCESS

Unauthorized access to the programs/processes that generate and interpret the messages should be prohibited in order to minimize potential threats and attack scenarios.

M1.11 DATA SECURITY IMPLEMENTATION SPECIFICS

Specific information-security interoperability provisions that may apply between agencies and other independent users involved in an exchange of data formatted in compliance with this Recommended Standard could be specified in an ICD.

M2 SANA CONSIDERATIONS

The following ODM-related items have been registered with the SANA Operator.

- The ODM/XML schema (see Section 8).

The following normative ODM elements shall be selected from the SANA registry (see Annex B):

- ODM ORIGINATORS;

- Spacecraft identifiers;
- Reference Frame Center and Third-Body Perturbations;
- Time Systems;
- Reference Frames (inertial, quasi-inertial, orbit-relative, spacecraft & attitude frames);
- Orbital element set and covariance matrix composition definitions.
- Atmosphere models
- Gravity models
- Object types
- Operational status
- Orbit averaging techniques
- Orbit types

The registration rule for new entries in the SANA registry is the approval of new requests by the CCSDS Area or Working Group responsible for the maintenance of the ODM at the time of the request. New requests for this registry should be sent to SANA (<mailto:info@sanaregistry.org>).

M3 PATENT CONSIDERATIONS

The recommendations of this document have no patent issues.

ANNEX N

INFORMATIVE REFERENCES

(INFORMATIVE)

- [N-1] *Navigation Data—Definitions and Conventions*. Report Concerning Space Data System Standards, CCSDS 500.0-G-4. Green Book. Issue 4. Washington, D.C.: CCSDS, November 2019
- [N-2] “CelesTrak.” Center for Space Standards & Innovation (CSSI). <<http://celestrak.com/>>
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- [N-5] *Ground Network Tracking and Acquisition Data Handbook*. 453-HNDK-GN. Greenbelt, Maryland: Goddard Space Flight Center, May 2007, <http://imbrium.mit.edu/LRORS/DOCUMENT/453_HDBK_GN.PDF> .
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- [N-7] David A. Vallado, et al. *Fundamentals of Astrodynamics and Applications, 4th Ed.*, Microcosm Press and Springer, ISBN 978-1881883180.
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