

**Recommendation for Space Data System Standards**

**ORBIT DATA  
MESSAGES**

PROPOSED STANDARD

**CCSDS 502.0.P-2.36**

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6 Oct1 2016 DRAFT

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This document has been approved for publication by the Management Council of the Consultative Committee for Space Data Systems (CCSDS) and represents the consensus technical agreement of the participating CCSDS Member Agencies. The procedure for review and authorization of CCSDS documents is detailed in the *Procedures Manual for the Consultative Committee for Space Data Systems*, and the record of Agency participation in the authorization of this document can be obtained from the CCSDS Secretariat at the address below.

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## FOREWORD

This document is a Recommended Standard for Orbit Data Messages (ODMs) and has been prepared by the Consultative Committee for Space Data Systems (CCSDS). The set of orbit data messages described in this Recommended Standard is the baseline concept for trajectory representation in data interchange applications that are cross-supported between Agencies of the CCSDS.

This Recommended Standard establishes a common framework and provides a common basis for the interchange of orbit and orbit-relevant data. It allows implementing organizations within each Agency to proceed coherently with the development of compatible derived standards for the flight and ground systems that are within their cognizance. Derived Agency standards may implement only a subset of the optional features allowed by the Recommended Standard and may incorporate features not addressed by this Recommended Standard.

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- Space and Upper Atmosphere Research Commission (SUPARCO)/Pakistan.
- Swedish Space Corporation (SSC)/Sweden.
- United States Geological Survey (USGS)/USA.

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<b>CCSDS 502.0.P-2.36</b>	Orbit Comprehensive Message		Added Orbit Comprehensive Message (OCM)

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# 1 INTRODUCTION

## 1.1 PURPOSE AND SCOPE

This Orbit Data Message (ODM) Recommended Standard specifies four standard message formats for use in transferring spacecraft orbit information between space agencies and commercial or governmental spacecraft operators: the Orbit Parameter Message (OPM), the Orbit Mean-Elements Message (OMM), the Orbit Ephemeris Message (OEM) and the Orbit Comprehensive Message (OCM). Such exchanges are used for:

- a) pre-flight planning for tracking or navigation support;
- b) scheduling tracking support;
- c) carrying out tracking operations (sometimes called metric predicts);
- d) performing orbit comparisons;
- e) carrying out navigation operations such as orbit propagation and orbit reconstruction;
- f) assessing mutual physical and electromagnetic interference among satellites orbiting the same celestial body (currently primarily Earth, Moon, and Mars);
- g) performing orbit conjunction (collision avoidance) studies; and
- h) developing and executing collaborative maneuvers to mitigate interference or enhance mutual operations.

This Recommended Standard includes sets of requirements and criteria that the message formats have been designed to meet. For exchanges where these requirements do not capture the needs of the participating agencies and satellite operators, another mechanism may be selected.

Four CCSDS-recommended ODMs are described in this Recommended Standard: the Orbit Parameter Message (OPM), the Orbit Mean-Elements Message (OMM), the Orbit Ephemeris Message (OEM), and the Orbit Comprehensive Message (OCM).

The Orbit Data Message (ODM) is an international standard published under the auspices of CCSDS and International Standards Organization (ISO) Subcommittee 13, developed jointly and in concert with the ISO SC14. As such, this CCSDS standard is also properly labeled as ISO 26900.

The recommended orbit data messages are ASCII text format (reference [3]). This ODM document describes 'keyword = value notation' formatted messages, while reference [4] describes XML formatted messages (the ICD should specify which of these formats will be exchanged).

NOTE – As currently specified, an OPM, OMM, or OEM file is to represent orbit data for a single spacecraft and the OCM is to represent orbit data for either a single spacecraft or single parent spacecraft of a parent/child spacecraft deployment scenario. It is possible that the architecture may support multiple spacecraft per file; this could be considered in the future.

## 1.2 APPLICABILITY

The rationale behind the design of each orbit data message is described in ANNEX F and may help the application engineer to select a suitable message. Definition of the orbit accuracy underlying a particular orbit message is outside of the scope of this Recommended Standard and should be specified via Interface Control Document (ICD) between data exchange participants (or specified via COMMENT sections in the message itself). Applicability information specific to each orbit data message format appears in sections 3, 4, and 5, as well as in F2.4.

This Recommended Standard is applicable only to the message format and content, but not to its transmission. The transmission of the message between agencies and operators is outside the scope of this document and should be specified in the ICD.

Description of the message formats based on the use of Extensible Markup Language (XML) is detailed in an integrated XML schema document for all Navigation Data Message Recommended Standards. (See reference [4].)

## 1.3 RATIONALE

This update to version 2 of the Orbit Data Messages adds a fourth message type (OCM) based on collaboration of the CCSDS Navigation Working Group and the ISO Technical Committee 20, Subcommittee 14, Working Group 3 (ISO TC20/SC14/WG3). A full list of the changes in this document is located in ANNEX H.

## 1.4 DOCUMENT STRUCTURE

Section 2 provides a brief overview of the CCSDS-recommended Orbit Data Message types, the Orbit Parameter Message (OPM), Orbit Mean-Elements Message (OMM), Orbit Ephemeris Message (OEM) and the Orbit Comprehensive Message (OCM).

Section 3 provides details about the structure and content of the OPM.

Section 4 provides details about the structure and content of the OMM.

Section 5 provides details about the structure and content of the OEM.

Section 6 provides details about the structure and content of the OCM.

Section 7 discusses the syntax considerations of the set of Orbit Data Messages (OPM, OMM, OEM and OCM).

Section 0 discusses security requirements for the Orbit Data Messages.

Following the principal content of the document, there are a number of annexes, both normative and informative, to guide the ODM user.

## 1.5 DEFINITIONS

For the purposes of this document, the following definitions apply:

- a) the word ‘agencies’ may also be construed as meaning ‘satellite operators’ or ‘satellite service providers’;
- b) the word ‘participant’ denotes an entity that has the ability to acquire or broadcast navigation messages and/or radio frequencies, for example, a spacecraft, a tracking station, a tracking instrument, or an agency/operator;
- c) the notation ‘n/a’ signifies ‘not applicable’;
- d) depending on context, the term ‘ODM’ may be used to refer to this document, or may be used to refer collectively to the OPM, OMM, OEM and OCM messages.

## 1.6 NOMENCLATURE

The following conventions apply for the normative specifications in this Manual:

- a) the words ‘shall’ and ‘must’ imply a binding and verifiable specification;
- b) the word ‘should’ implies an optional, but desirable, specification;
- c) the word ‘may’ implies an optional specification;
- d) the words ‘is’, ‘are’, and ‘will’ imply statements of fact.

NOTE – These conventions do not imply constraints on diction in text that is clearly informative in nature.

## 1.7 REFERENCES

The following documents contain provisions which, through reference in this text, constitute provisions of this Recommended Standard. At the time of publication, the editions indicated were valid. All documents are subject to revision, and users of this Recommended Standard are encouraged to investigate the possibility of applying the most recent editions of the documents indicated below. The CCSDS Secretariat maintains a register of currently valid CCSDS Recommended Standards.

- [1] *Time Code Formats*. Recommendation for Space Data System Standards, CCSDS 301.0-B-4. Blue Book. Issue 4. Washington, D.C.: CCSDS, November 2010.
- [2] *United Nations Office of Outer Space Affairs satellite designator/index, searchable at* <<http://www.unoosa.org/oosa/osoindex> >
- [3] *Information Technology—8-Bit Single-Byte Coded Graphic Character Sets—Part 1: Latin Alphabet No. 1*. International Standard, ISO/IEC 8859-1:1998. Geneva: ISO, 1998.
- [4] *XML Specification for Navigation Data Messages*. Recommendation for Space Data System Standards, CCSDS 505.0-B-1. Blue Book. Issue 1. Washington, D.C.: CCSDS, December 2010.
- [5] “JPL Solar System Dynamics.” Solar System Dynamics Group. <<http://ssd.jpl.nasa.gov/>>
- [6] Paul V. Biron and Ashok Malhotra, eds. *XML Schema Part 2: Datatypes*. 2nd Edition. W3C Recommendation. N.p.: W3C, October 2004. <<http://www.w3.org/TR/2001/REC-xmlschema-2-20010502/>>
- [7] *IEEE Standard for Binary Floating-Point Arithmetic*. IEEE Std 754-1985. New York: IEEE, 1985.

## **2 OVERVIEW**

### **2.1 ORBIT DATA MESSAGE TYPES**

#### **2.2 ORBIT PARAMETER MESSAGE (OPM)**

An OPM specifies the position and velocity of a single object at a specified epoch. Optionally, osculating Keplerian elements may be provided. This message is suited to exchanges that (1) involve automated interaction and/or human interaction, and (2) do not require high-fidelity dynamic modeling.

The OPM requires the use of a propagation technique to determine the position and velocity at times different from the specified epoch, leading to a higher level of effort for software implementation than for the OEM. A 6x6 position/velocity covariance matrix that may be used in the propagation process is optional.

The OPM allows for modeling of any number of maneuvers (as both finite and instantaneous events) and simple modeling of solar radiation pressure and atmospheric drag.

The OPM also contains an optional covariance matrix which reflects the uncertainty of the orbit state.

Though primarily intended for use by computers, the attributes of the OPM also make it suitable for applications such as exchanges by email, FAX or voice, or applications where the message is to be frequently interpreted by humans.

#### **2.3 ORBIT MEAN-ELEMENTS MESSAGE (OMM)**

An OMM specifies the orbital characteristics of a single object at a specified epoch, expressed in mean Keplerian elements. This message is suited to exchanges that (1) involve automated interaction and/or human interaction, and (2) do not require high-fidelity dynamic modeling. Such exchanges may be inter-agency exchanges, or ad hoc exchanges among satellite operators when interface control documents have not been negotiated. Ad hoc interactions usually involve more than one satellite, each satellite controlled and operated by a different operating authority.

The OMM includes keywords and values that can be used to generate canonical NORAD Two Line Element Sets (TLEs) to accommodate the needs of heritage users (see reference [L4]).

The OMM also contains an optional covariance matrix which reflects the uncertainty of the mean Keplerian elements. This information may be used to determine contact parameters that encompass uncertainties in predicted future states of orbiting objects of interest.

This message is suited for directing antennas and planning contacts with satellites. It is not recommended for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, or propagating precisely the orbits of

active satellites, inactive man-made objects, and near-Earth debris fragments. It is not suitable for numerical integration of the governing equations.

Though primarily intended for use by computers, the attributes of the OMM also make it suitable for applications such as exchanges by email, FAX or voice, or applications where the message is to be frequently interpreted by humans.

## 2.4 ORBIT EPHEMERIS MESSAGE (OEM)

An OEM specifies the position and velocity of a single object at multiple epochs contained within a specified time range. The OEM is suited to exchanges that (1) involve automated interaction (e.g., computer-to-computer communication where frequent, fast automated time interpretation and processing is required), and (2) require higher fidelity or higher precision dynamic modeling than is possible with the OPM.

The OEM allows for dynamic modeling of any number of gravitational and non-gravitational accelerations. The OEM requires the use of an interpolation technique to interpret the position and velocity at times different from the tabular epochs.

The OEM also contains an optional covariance matrix which reflects the uncertainty of the orbit solution used to generate states in the ephemeris.

## 2.5 ORBIT COMPREHENSIVE MESSAGE (OCM)

An OCM specifies position and velocity of either a single object or an en masse parent/child deployment scenario stemming from a single object. The OCM aggregates and extends OPM, OEM and OMM content in a single comprehensive hybrid message (file) and offers the following additional capabilities:

- Optional Earth Orientation (UT1 and UTC) at a nearby (relevant) reference epoch;
- Optional Leap second specification
- Optional area cross-sections for drag, SRP perturbations modeling.
- Optional spacecraft dimensions and orientation information for collision probability estimation
- Optional perturbations model specification;
- Optional maneuver specification (impulsive or finite burn);
- Optional orbit states (specified using one or more of Cartesian and orbit elements and reference frames) for a single or parent object at either a single epoch or as a time history (ephemeris);
- Optional orbit determination data and metrics;
- Optional covariance matrix of selectable/arbitrary order for a single or parent object at either a single epoch or as a time history (ephemeris) which reflects the uncertainty of the orbit solution or Monte Carlo simulation used to obtain the nominal states in the orbit state(s);
- Optional covariance content options (e.g. Cartesian 3x3, 6x6, 7x7, or any combination of order, reference frame and orbit elements)

- Optional State Transition Matrix specification;
- Optional Ephemeris Compression (EC) specification via polynomials;

The OCM simultaneously emphasizes flexibility and message conciseness by offering extensive optional content while minimizing mandatory content. The OCM is well-suited for exchanges that (1) involve automated interaction (e.g., computer-to-computer communication where frequent, fast automated time interpretation and processing is required), and (2) involve regular orbit data transfer for numerous objects (e.g. 200,000) using minimal network bandwidth, disk storage and quantity of files. The OCM allows the user, in a single message/file, to either embed high-fidelity orbit propagation into an ephemeris time history (akin to the OEM ephemeris), or specify orbital states which can be propagated with supplied perturbations model parameters (akin to OPM content), or both.

## **2.6 EXCHANGE OF MULTIPLE MESSAGES**

For a given object, multiple OPM, OMM, or OEM messages may be provided in a message exchange session to achieve ephemeris fidelity requirements, whereas a single, self-contained OCM is typically sufficient. If ephemeris information for multiple objects is to be exchanged, then multiple OPM, OMM, OEM or OCM files must be used, with the exception that the OCM supports parent/child deployment scenario specifications in a single message.

## **2.7 DEFINITIONS**

Definitions of time systems, reference frames, planetary models, maneuvers and other fundamental topics related to the interpretation and processing of state vectors and spacecraft ephemerides are provided in reference [L1].



### **3 ORBIT PARAMETER MESSAGE (OPM)**

#### **3.1 GENERAL**

**3.1.1** Orbit information may be exchanged between two participants by sending a state vector (see reference [L1]) for a specified epoch using an Orbit Parameter Message (OPM). The message recipient must have an orbit propagator available that is able to propagate the OPM state vector to compute the orbit at other desired epochs. For this propagation, additional ancillary information (spacecraft properties such as mass, area, and maneuver planning data, if applicable) may be included with the message.

**3.1.2** Osculating Keplerian elements and Gravitational Coefficient may be included in the OPM in addition to the Cartesian state to aid the message recipient in performing consistency checks. If any Keplerian element is included, the entire set of elements must be provided.

**3.1.3** If participants wish to exchange mean element information, then the Orbit Mean-Elements Message (OMM) or Orbit Comprehensive Message (OCM) should be the selected message type. (See sections 4 and 6.)

**3.1.4** The use of the OPM is best applicable under the following conditions:

- a) an orbit propagator consistent with the models used to develop the orbit data should be available at the receiver's site;
- b) the receiver's modeling of gravitational forces, solar radiation pressure, atmospheric drag, and thrust phases (see reference [L1]) should fulfill accuracy requirements established between the exchange partners.

**3.1.5** The OPM shall be a plain text file consisting of orbit data for a single object. It shall be easily readable by both humans and computers.

**3.1.6** The OPM file-naming scheme should be agreed to on a case-by-case basis between the exchange partners, and should be documented in an ICD. The method of exchanging OPMs should be decided on a case-by-case basis by the exchange partners and documented in an ICD.

NOTE – Detailed syntax rules for the OPM are specified in section 7.

#### **3.2 OPM CONTENT/STRUCTURE**

##### **3.2.1 GENERAL**

The OPM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) data; and

- d) optional comments (explanatory information).

### 3.2.2 OPM HEADER

3.2.2.1 Table 3-1 specifies for each header item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is mandatory or optional.

3.2.2.2 Only those keywords shown in table 3-1 shall be used in an OPM header.

**Table 3-1: OPM Header**

Keyword	Description	Examples of Values	Mandatory
CCSDS_OPM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	2.0	Yes
COMMENT	Comments (allowed in the OPM Header only immediately after the OPM version number). (See 7.7 for formatting rules.)	COMMENT This is a comment	No
CREATION_DATE	File creation date/time in UTC. (For format specification, see 7.5.9)	2001-11-06T11:17:33 2002-204T15:56:23Z	Yes
ORIGINATOR	Creating agency or operator (value should be drawn from the SANA "Organizations" registry). The country of origin should also be provided where the originator is not a national space agency.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT/USA, USAF, INMARSAT/UK	Yes

### 3.2.3 OPM METADATA

3.2.3.1 Table 3-2 specifies for each metadata item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is mandatory or optional.

3.2.3.2 Only those keywords shown in table 3-2 shall be used in OPM metadata.

NOTE – For some keywords (OBJECT\_NAME, OBJECT\_ID, ORB\_CENTER\_NAME) there are no definitive lists of authorized values maintained by a control authority; the references listed in 0 are the best known sources for authorized values to date. For the TIME\_SYSTEM and REF\_FRAME keywords, the approved values are listed in ANNEX B.

**Table 3-2: OPM Metadata**

Keyword	Description	Examples of Values	Mandatory
COMMENT	Comments (allowed at the beginning of the OPM Metadata). (See 7.7 for formatting rules.)	COMMENT This is a comment	No
OBJECT_NAME	Spacecraft name for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [2]), which include Object name and international designator of the participant.	EUTELSAT W1 MARS PATHFINDER STS 106 NEAR	Yes
OBJECT_ID	Object identifier of the object for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended that values be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [2]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch.  In cases where the asset is not listed in the UN Office of Outer Space Affairs designator index or that index format is not used, the value should be provided in an ICD.	2000-052A 1996-068A 2000-053A 1996-008A	Yes
CENTER_NAME	Origin of reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, or another spacecraft (in this case the value for 'CENTER_NAME' is subject to the same rules as for 'OBJECT_NAME'). There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).	EARTH EARTH BARYCENTER MOON SOLAR SYSTEM BARYCENTER SUN JUPITER BARYCENTER STS 106 EROS	Yes

Keyword	Description	Examples of Values	Mandatory
REF_FRAME	Name of the reference frame in which the state vector and optional Keplerian element data are given. Use of values other than those in ANNEX B, subsection B2 must be documented and conveyed in an ICD. The reference frame must be the same for all data elements, with the exception of the maneuvers and covariance matrix, for which applicable different reference frames may be specified.	ICRF ITRF-93 ITRF-97 ITRF2000 ITRFXXXX (TEMPLATE FOR A FUTURE VERSION) TOD (TRUE EQUATOR/EQUINOX OF DATE) EME2000 (EARTH MEAN EQUATOR AND EQUINOX OF J2000) TDR (TRUE OF DATE ROTATING) GRC (GREENWICH ROTATING COORDINATE FRAME)	Yes
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	No
TIME_SYSTEM	Time system used for state vector, maneuver, and covariance data (also see table 3-3). Use of values other than those in ANNEX B, subsection B1 must be documented and conveyed in an ICD.	UTC, TAI, TT, GPS, TDB, TCB	Yes

### 3.2.4 OPM DATA

3.2.4.1 Table 3-3 provides an overview of the six logical blocks in the OPM Data section (State Vector, Osculating Keplerian Elements, Spacecraft Parameters, Position/Velocity Covariance Matrix, Maneuver Parameters, and User Defined Parameters), and specifies for each data item:

- a) the keyword to be used;
- b) a short description of the item;
- c) the units to be used;
- d) whether the item is mandatory or optional.

**3.2.4.2** Only those keywords shown in table 3-3 shall be used in OPM data.

NOTE – Requirements relating to the keywords in table 3-3 appear after the table.

**Table 3-3: OPM Data**

Keyword	Description	Units	Mandatory
<b>State Vector Components in the Specified Coordinate System</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
EPOCH	Epoch of state vector & optional Keplerian elements. (See 7.5.9 for formatting rules.)	n/a	Yes
X	Position vector X-component	km	Yes
Y	Position vector Y-component	km	Yes
Z	Position vector Z-component	km	Yes
X_DOT	Velocity vector X-component	km/s	Yes
Y_DOT	Velocity vector Y-component	km/s	Yes
Z_DOT	Velocity vector Z-component	km/s	Yes
<b>Osculating Keplerian Elements in the Specified Reference Frame (none or all parameters of this block must be given.)</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
SEMI_MAJOR_AXIS	Semi-major axis	km	No
ECCENTRICITY	Eccentricity	n/a	No
INCLINATION	Inclination	deg	No
RA_OF_ASC_NODE	Right ascension of ascending node	deg	No
ARG_OF_PERICENTER	Argument of pericenter	deg	No
TRUE_ANOMALY or MEAN_ANOMALY	True anomaly or mean anomaly	deg	No
GM	Gravitational Coefficient (Gravitational Constant x Central Mass)	km**3/s**2	No
<b>Spacecraft Parameters</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
MASS	S/C Mass	kg	No
SOLAR_RAD_AREA	Solar Radiation Pressure Area ( $A_R$ )	m**2	No
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient ( $C_R$ )	n/a	No
DRAG_AREA	Drag Area ( $A_D$ )	m**2	No
DRAG_COEFF	Drag Coefficient ( $C_D$ )	n/a	No
<b>Position/Velocity Covariance Matrix (6x6 Lower Triangular Form. None or all parameters of the matrix must be given. COV_REF_FRAME may be omitted if it is the same as the metadata REF_FRAME.)</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
COV_REF_FRAME	Coordinate system for covariance matrix (value must be selected from ANNEX B, subsections B2 and B3)	n/a	No
CX_X	Covariance matrix [1,1]	km**2	No
CY_X	Covariance matrix [2,1]	km**2	No
CY_Y	Covariance matrix [2,2]	km**2	No
CZ_X	Covariance matrix [3,1]	km**2	No
CZ_Y	Covariance matrix [3,2]	km**2	No
CZ_Z	Covariance matrix [3,3]	km**2	No
CX_DOT_X	Covariance matrix [4,1]	km**2/s	No
CX_DOT_Y	Covariance matrix [4,2]	km**2/s	No
CX_DOT_Z	Covariance matrix [4,3]	km**2/s	No
CX_DOT_X_DOT	Covariance matrix [4,4]	km**2/s**2	No
CY_DOT_X	Covariance matrix [5,1]	km**2/s	No
CY_DOT_Y	Covariance matrix [5,2]	km**2/s	No

Keyword	Description	Units	Mandatory
CY_DOT_Z	Covariance matrix [5,3]	km**2/s	No
CY_DOT_X_DOT	Covariance matrix [5,4]	km**2/s**2	No
CY_DOT_Y_DOT	Covariance matrix [5,5]	km**2/s**2	No
CZ_DOT_X	Covariance matrix [6,1]	km**2/s	No
CZ_DOT_Y	Covariance matrix [6,2]	km**2/s	No
CZ_DOT_Z	Covariance matrix [6,3]	km**2/s	No
CZ_DOT_X_DOT	Covariance matrix [6,4]	km**2/s**2	No
CZ_DOT_Y_DOT	Covariance matrix [6,5]	km**2/s**2	No
CZ_DOT_Z_DOT	Covariance matrix [6,6]	km**2/s**2	No
Maneuver Parameters (Repeat for each maneuver. None or all parameters of this block must be given.)			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
MAN_EPOCH_IGNITION	Epoch of ignition. (See 7.5.9 for formatting rules.)	n/a	No
MAN_DURATION	Maneuver duration (If = 0, impulsive maneuver)	s	No
MAN_DELTA_MASS	Mass change during maneuver (value is < 0)	kg	No
MAN_REF_FRAME	Coordinate system for velocity increment vector (value must be selected from ANNEX B, subsection B2 and B3)	n/a	No
MAN_DV_1	1 <sup>st</sup> component of the velocity increment	km/s	No
MAN_DV_2	2 <sup>nd</sup> component of the velocity increment	km/s	No
MAN_DV_3	3 <sup>rd</sup> component of the velocity increment	km/s	No
User Defined Parameters (all parameters in this section must be described in an ICD).			
USER_DEFINED_x	User defined parameter, where 'x' is replaced by a variable length user specified character string. Any number of user defined parameters may be included, if necessary to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED_EARTH_MODEL = WGS-84	n/a	No

**3.2.4.3** All values except Maneuver Parameters in the OPM data are ‘at epoch’, i.e., the value of the parameter at the time specified in the EPOCH keyword.

**3.2.4.4** Table 3-3 is broken into six logical blocks, each of which has a descriptive heading. These descriptive headings shall not be included in an OPM, unless they appear in a properly formatted COMMENT statement.

**3.2.4.5** If the solar radiation coefficient,  $C_R$ , is set to zero, no solar radiation pressure shall be taken into account.

**3.2.4.6** If the atmospheric drag coefficient,  $C_D$ , is set to zero, no atmospheric drag shall be taken into account.

**3.2.4.7** Parameters for thrust phases may be optionally given for the computation of the trajectory during or after maneuver execution (see reference [L1] for the simplified modeling of such maneuvers). For impulsive maneuvers, MAN\_DURATION must be set to zero. MAN\_DELTA\_MASS may be used for both finite and impulsive maneuvers; the value must be a negative number. Permissible reference frames for the velocity increment vector shall be those specified in ANNEX B, subsection B2 and B3.

**3.2.4.8** Multiple sets of maneuver parameters may appear. For each maneuver, all the maneuver parameters shall be repeated in the order shown in table 3-3.

**3.2.4.9** If the OPM contains a maneuver definition, the Spacecraft Parameters section must be included.

**3.2.4.10** Values in the covariance matrix shall be expressed in the applicable reference frame (COV\_REF\_FRAME keyword if used, or REF\_FRAME keyword if not), and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row left to right. Variance and covariance values shall be expressed in standard double precision as related in 7.5. This logical block of the OPM may be useful for risk assessment and establishing maneuver and mission margins. The intent is to provide causal connections between output orbit data and both physical hypotheses and measurement uncertainties. These causal relationships guide operators' corrective actions and mitigations.

**3.2.4.11** A section of User Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User Defined Parameters, if included in an OPM, should be used as sparingly as possible; their use is not encouraged.

### **3.3 OPM EXAMPLES**

Figure 3-1 through figure 3-4 are examples of Orbit Parameter Messages. The first has only a state; the second has state, Keplerian elements, and maneuvers; the third and fourth include the position/velocity covariance matrix.

Figure 3-1 and figure 3-2 are compatible with the ODM version 1.0 processing because they do not contain any of the unique features of the ODM version 2.0. Thus for these examples a value of 1.0 could be specified for the 'CCSDS\_OPM\_VERS' keyword. (See ANNEX I.)

Figure 3-3 and figure 3-4 include unique features of ODM version 2.0, and thus 'CCSDS\_OPM\_VERS = 2.0' must be specified.



```
CCSDS_OPM_VERS = 2.0
CREATION_DATE  = 1998-11-06T09:23:57
ORIGINATOR     = JAXA

COMMENT        GEOCENTRIC, CARTESIAN, EARTH FIXED
OBJECT_NAME    = GODZILLA 5
OBJECT_ID      = 1998-057A
CENTER_NAME    = EARTH
REF_FRAME      = ITRF-97
TIME_SYSTEM    = UTC

EPOCH =          1998-12-18T14:28:15.1172
X =             6503.514000
Y =             1239.647000
Z =             -717.490000
X_DOT =         -0.873160
Y_DOT =          8.740420
Z_DOT =         -4.191076
MASS =           3000.000000
SOLAR_RAD_AREA =  18.770000
SOLAR_RAD_COEFF =  1.000000
DRAG_AREA =      18.770000
DRAG_COEFF =     2.500000
```

**Figure 3-1: Simple OPM File Example**

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OPM_VERS      = 2.0

COMMENT  Generated by GSOC, R. Kiehling
COMMENT  Current intermediate orbit IO2 and maneuver planning data

CREATION_DATE       = 2000-06-03T05:33:00.000
ORIGINATOR          = GSOC

OBJECT_NAME         = EUTELSAT W4
OBJECT_ID           = 2000-028A
CENTER_NAME         = EARTH
REF_FRAME           = TOD
TIME_SYSTEM         = UTC

COMMENT  State Vector
EPOCH               = 2006-06-03T00:00:00.000
X                   = 6655.9942           [km]
Y                   = -40218.5751        [km]
Z                   = -82.9177           [km]
X_DOT               = 3.11548208         [km/s]
Y_DOT               = 0.47042605         [km/s]
Z_DOT               = -0.00101495        [km/s]

COMMENT  Keplerian elements
SEMI_MAJOR_AXIS     = 41399.5123         [km]
ECCENTRICITY        = 0.020842611
INCLINATION         = 0.117746           [deg]
RA_OF_ASC_NODE      = 17.604721         [deg]
ARG_OF_PERICENTER   = 218.242943        [deg]
TRUE_ANOMALY        = 41.922339         [deg]
GM                  = 398600.4415        [km**3/s**2]

COMMENT  Spacecraft parameters
MASS                 = 1913.000           [kg]
SOLAR_RAD_AREA      = 10.000             [m**2]
SOLAR_RAD_COEFF     = 1.300
DRAG_AREA           = 10.000             [m**2]
DRAG_COEFF          = 2.300

COMMENT  2 planned maneuvers

COMMENT  First maneuver: AMF-3
COMMENT  Non-impulsive, thrust direction fixed in inertial frame
MAN_EPOCH_IGNITION = 2000-06-03T09:00:34.1
MAN_DURATION        = 132.60             [s]
MAN_DELTA_MASS      = -18.418            [kg]
MAN_REF_FRAME       = EME2000
MAN_DV_1            = -0.02325700        [km/s]
MAN_DV_2            = 0.01683160         [km/s]
MAN_DV_3            = -0.00893444        [km/s]

COMMENT  Second maneuver: first station acquisition maneuver
COMMENT  impulsive, thrust direction fixed in RTN frame
MAN_EPOCH_IGNITION = 2000-06-05T18:59:21.0
MAN_DURATION        = 0.00               [s]
MAN_DELTA_MASS      = -1.469             [kg]
MAN_REF_FRAME       = RTN
MAN_DV_1            = 0.00101500         [km/s]
MAN_DV_2            = -0.00187300        [km/s]
MAN_DV_3            = 0.00000000         [km/s]

```

**Figure 3-2: OPM File Example with Optional Keplerian Elements and Two Maneuvers**

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OPM_VERS = 2.0

CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR    = JAXA

COMMENT       GEOCENTRIC, CARTESIAN, EARTH FIXED
OBJECT_NAME   = GODZILLA 5
OBJECT_ID     = 1998-057A
CENTER_NAME   = EARTH
REF_FRAME     = ITRF-97
TIME_SYSTEM   = UTC

EPOCH =      1998-12-18T14:28:15.1172
X =          6503.514000
Y =          1239.647000
Z =          -717.490000
X_DOT =      -0.873160
Y_DOT =          8.740420
Z_DOT =      -4.191076

MASS =          3000.000000
SOLAR_RAD_AREA = 18.770000
SOLAR_RAD_COEFF = 1.000000
DRAG_AREA =    18.770000
DRAG_COEFF =   2.500000

CX_X = 3.331349476038534e-04
CY_X = 4.618927349220216e-04
CY_Y = 6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z = 3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z = 2.484949578400095e-07
CX_DOT_X_DOT = 4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z = 1.798098699846038e-07
CY_DOT_X_DOT = 2.608899201686016e-10
CY_DOT_Y_DOT = 1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z = 3.540310904497689e-07
CZ_DOT_X_DOT = 1.869263192954590e-10
CZ_DOT_Y_DOT = 1.008862586240695e-10
CZ_DOT_Z_DOT = 6.224444338635500e-10
    
```

**Figure 3-3: OPM File Example with Covariance Matrix**

```

CCSDS_OPM_VERS = 2.0
COMMENT Generated by GSOC, R. Kiehling
COMMENT Current intermediate orbit IO2 and maneuver planning data
CREATION_DATE = 2000-06-03T05:33:00.000
ORIGINATOR = GSOC
OBJECT_NAME = EUTELSAT W4
OBJECT_ID = 2000-028A
CENTER_NAME = EARTH
REF_FRAME = TOD
TIME_SYSTEM = UTC
COMMENT State Vector
EPOCH = 2006-06-03T00:00:00.000
X = 6655.9942 [km]
Y = -40218.5751 [km]
Z = -82.9177 [km]
X_DOT = 3.11548208 [km/s]
Y_DOT = 0.47042605 [km/s]
Z_DOT = -0.00101495 [km/s]
COMMENT Keplerian elements
SEMI_MAJOR_AXIS = 41399.5123 [km]
ECCENTRICITY = 0.020842611
INCLINATION = 0.117746 [deg]
RA_OF_ASC_NODE = 17.604721 [deg]
ARG_OF_PERICENTER = 218.242943 [deg]
TRUE_ANOMALY = 41.922339 [deg]
GM = 398600.4415 [km**3/s**2]
COMMENT Spacecraft parameters
MASS = 1913.000 [kg]
SOLAR_RAD_AREA = 10.000 [m**2]
SOLAR_RAD_COEFF = 1.300
DRAG_AREA = 10.000 [m**2]
DRAG_COEFF = 2.300
COV_REF_FRAME = RTN
CX_X = 3.331349476038534e-04
CY_X = 4.618927349220216e-04
CY_Y = 6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z = 3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z = 2.484949578400095e-07
CX_DOT_X_DOT = 4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z = 1.798098699846038e-07
CY_DOT_X_DOT = 2.608899201686016e-10
CY_DOT_Y_DOT = 1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z = 3.540310904497689e-07
CZ_DOT_X_DOT = 1.869263192954590e-10
CZ_DOT_Y_DOT = 1.008862586240695e-10
CZ_DOT_Z_DOT = 6.224444338635500e-10
USER_DEFINED_EARTH_MODEL = WGS-84

```

**Figure 3-4: OPM File Example with Optional Keplerian Elements, Covariance Matrix, and a User Defined Parameter**

## 4 ORBIT MEAN-ELEMENTS MESSAGE (OMM)

### 4.1 GENERAL

**4.1.1** Orbit information may be exchanged between two participants by sending an orbital state based on mean Keplerian elements (see reference [L1]) for a specified epoch using an Orbit Mean-Elements Message (OMM). The message recipient must use appropriate orbit propagator algorithms in order to correctly propagate the OMM state to compute the orbit at other desired epochs.

**4.1.2** The OMM is intended to allow replication of the data content of an existing TLE in a CCSDS standard format, but the message can also accommodate other implementations of mean elements. All essential fields of the ‘de facto standard’ TLE are included in the OMM in a style that is consistent with that of the other ODMs (i.e., the OPM and OEM). From the fields in the OMM, it is possible to generate a TLE (see reference [L3]). Programs that convert OMMs to TLEs must be aware of the structural requirements of the TLE, including the checksum algorithm and the formatting requirements for the values in the TLE. The checksum and formatting requirements of the TLE do not apply to the values in an OMM.

**4.1.3** If participants wish to exchange osculating element information, then the Orbit Parameter Message (OPM) or the Orbit Comprehensive Message (OCM) should be the selected message type. (See sections 3 and 6.)

**4.1.4** The use of the OMM is best applicable under the following conditions:

- a) an orbit propagator consistent with the models used to develop the orbit data should be run at the receiver’s site;
- b) the receiver’s modeling of gravitational forces, solar radiation pressure, atmospheric drag, etc. (see reference [L1]), should fulfill accuracy requirements established between the exchange partners.

**4.1.5** The OMM shall be a plain text file consisting of orbit data for a single object. It shall be easily readable by both humans and computers.

**4.1.6** The OMM file-naming scheme should be agreed to on a case-by-case basis between the exchange partners, and should be documented in an ICD. The method of exchanging OMMs should be decided on a case-by-case basis by the exchange partners and documented in an ICD.

NOTE – Detailed syntax rules for the OMM are specified in section 7.

## 4.2 OMM CONTENT/STRUCTURE

### 4.2.1 GENERAL

The OMM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) data; and
- d) optional comments (explanatory information).

### 4.2.2 OMM HEADER

4.2.2.1 Table 4-1 specifies for each header item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is mandatory or optional.

4.2.2.2 Only those keywords shown in table 4-1 shall be used in an OMM header.

**Table 4-1: OMM Header**

Keyword	Description	Examples of Values	Mandatory
CCSDS_OMM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	2.0	Yes
COMMENT	Comments (allowed in the OMM Header only immediately after the OMM version number). (See 7.7 for formatting rules.)	COMMENT This is a comment	No
CREATION_DATE	File creation date/time in UTC. ( For format specification, see 7.5.9.)	2001-11-06T11:17:33 2002-204T15:56:23Z	Yes
ORIGINATOR	Creating agency or operator (value should be drawn from the SANA "Organizations" registry). The country of origin should also be provided where the originator is not a national space agency.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT/USA, USAF, INMARSAT/UK	Yes

### **4.2.3 OMM METADATA**

**4.2.3.1** Table 4-2 specifies for each metadata item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is mandatory or optional.

**4.2.3.2** Only those keywords shown in table 4-2 shall be used in OMM metadata.

NOTE – For some keywords (OBJECT\_NAME, OBJECT\_ID, CENTER\_NAME) there are no definitive lists of authorized values maintained by a control authority; the references listed in 0 are the best known sources for authorized values to date. For the TIME\_SYSTEM and REF\_FRAME keywords, the approved values are shown in ANNEX B, subsections B1 and B2.

**Table 4-2: OMM Metadata**

Keyword	Description	Examples of Values	Mandatory
COMMENT	Comments (allowed at the beginning of the OMM Metadata). (See 7.7 for formatting rules.)	COMMENT This is a comment	No
OBJECT_NAME	Spacecraft name for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [2]), which include Object name and international designator of the participant.	TELCOM 2 SPACEWAY 2 INMARSAT 4-F2	Yes
OBJECT_ID	Object identifier of the object for which the orbit state is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended that values be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [2]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch.  In cases where the asset is not listed in the bulletin, or the UN Office of Outer Space Affairs designator index format is not used, the value should be provided in an ICD.	2005-046B 2005-046A 2003-022A 2005-044A	Yes
CENTER_NAME	Origin of reference frame. There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).	EARTH MARS MOON	Yes
REF_FRAME	Name of the reference frame in which the Keplerian element data are given. Use of values other than those in ANNEX B, subsection B2 must be documented and conveyed in an ICD. The reference frame must be the same for all data elements, with the exception of the covariance matrix, for which an applicable different reference frame may be specified.	TEME EME2000	Yes
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	No
TIME_SYSTEM	Time system used for the orbit state and covariance matrix. Use of values other than those in ANNEX B, subsection B1 must be documented and conveyed in an ICD.	UTC	Yes
MEAN_ELEMENT_THEORY	Description of the Mean Element Theory. Indicates the proper method to employ to propagate the state.	SGP4 DSST USM	Yes



#### 4.2.4 OMM DATA

4.2.4.1 Table 4-3 provides an overview of the five logical blocks in the OMM Data section (Mean Keplerian Elements, Spacecraft Parameters, TLE Related Parameters, Position/Velocity Covariance Matrix, and User Defined Parameters), and specifies for each data item:

- a) the keyword to be used;
- b) a short description of the item;
- c) the units to be used;
- d) whether the item is mandatory or optional.

4.2.4.2 Only those keywords shown in table 4-3 shall be used in OMM data.

NOTE – Requirements relating to the keywords in table 4-3 appear after the table.

**Table 4-3: OMM Data**

Keyword	Description	Units	Mandatory
<b>Mean Keplerian Elements in the Specified Reference Frame</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
EPOCH	Epoch of Mean Keplerian elements. (See 7.5.9 for formatting rules.)	n/a	Yes
SEMI_MAJOR_AXIS or MEAN_MOTION	Semi-major axis in kilometers (preferred), or, if MEAN_ELEMENT_THEORY = SGP/SGP4, the Keplerian Mean motion in revolutions per day	km rev/day	Yes
ECCENTRICITY	Eccentricity	n/a	Yes
INCLINATION	Inclination	deg	Yes
RA_OF_ASC_NODE	Right ascension of ascending node	deg	Yes
ARG_OF_PERICENTER	Argument of pericenter	deg	Yes
MEAN_ANOMALY	Mean anomaly	deg	Yes
GM	Gravitational Coefficient (Gravitational Constant x Central Mass)	km**3/s**2	No
<b>Spacecraft Parameters</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
MASS	S/C Mass	kg	No
SOLAR_RAD_AREA	Solar Radiation Pressure Area ( $A_R$ )	m**2	No
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient ( $C_R$ )	n/a	No
DRAG_AREA	Drag Area ( $A_D$ )	m**2	No
DRAG_COEFF	Drag Coefficient ( $C_D$ )	n/a	No
<b>TLE Related Parameters (This section is only required if MEAN_ELEMENT_THEORY=SGP/SGP4)</b>			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
EPHEMERIS_TYPE	Default value = 0. (See 4.2.4.7.)	n/a	No
CLASSIFICATION_TYPE	Default value = U. (See 4.2.4.7.)	n/a	No
NORAD_CAT_ID	NORAD Catalog Number ('Satellite Number') an integer of up to nine digits. This keyword is only required if MEAN_ELEMENT_THEORY=SGP/SGP4.	n/a	No

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Keyword	Description	Units	Mandatory
ELEMENT_SET_NO	Element set number for this satellite. Normally incremented sequentially, but may be out of sync if it is generated from a backup source. Used to distinguish different TLEs, and therefore only meaningful if TLE-based data is being exchanged (i.e., MEAN_ELEMENT_THEORY = SGP/SGP4).	n/a	No
REV_AT_EPOCH	Revolution Number	n/a	No
BSTAR	SGP/SGP4 drag-like coefficient (in units 1/[Earth radii]). Only required if MEAN_ELEMENT_THEORY=SGP/SGP4	1/ER	No
MEAN_MOTION_DOT	First Time Derivative of the Mean Motion (only required if MEAN_ELEMENT_THEORY = SGP)	rev/day**2	No
MEAN_MOTION_DDOT	Second Time Derivative of Mean Motion (only required if MEAN_ELEMENT_THEORY = SGP)	rev/day**3	No
Position/Velocity Covariance Matrix (6x6 Lower Triangular Form. None or all parameters of the matrix must be given. COV_REF_FRAME may be omitted if it is the same as the metadata REF_FRAME.)			
COMMENT	(See 7.7 for formatting rules.)	n/a	No
COV_REF_FRAME	Reference frame for the covariance matrix. The value must be selected from ANNEX B, subsections B2 and B3.	n/a	No
CX_X	Covariance matrix [1,1]	km**2	No
CY_X	Covariance matrix [2,1]	km**2	No
CY_Y	Covariance matrix [2,2]	km**2	No
CZ_X	Covariance matrix [3,1]	km**2	No
CZ_Y	Covariance matrix [3,2]	km**2	No
CZ_Z	Covariance matrix [3,3]	km**2	No
CX_DOT_X	Covariance matrix [4,1]	km**2/s	No
CX_DOT_Y	Covariance matrix [4,2]	km**2/s	No
CX_DOT_Z	Covariance matrix [4,3]	km**2/s	No
CX_DOT_X_DOT	Covariance matrix [4,4]	km**2/s**2	No
CY_DOT_X	Covariance matrix [5,1]	km**2/s	No
CY_DOT_Y	Covariance matrix [5,2]	km**2/s	No
CY_DOT_Z	Covariance matrix [5,3]	km**2/s	No
CY_DOT_X_DOT	Covariance matrix [5,4]	km**2/s**2	No
CY_DOT_Y_DOT	Covariance matrix [5,5]	km**2/s**2	No
CZ_DOT_X	Covariance matrix [6,1]	km**2/s	No
CZ_DOT_Y	Covariance matrix [6,2]	km**2/s	No
CZ_DOT_Z	Covariance matrix [6,3]	km**2/s	No
CZ_DOT_X_DOT	Covariance matrix [6,4]	km**2/s**2	No
CZ_DOT_Y_DOT	Covariance matrix [6,5]	km**2/s**2	No
CZ_DOT_Z_DOT	Covariance matrix [6,6]	km**2/s**2	No
User Defined Parameters (all parameters in this section must be described in an ICD).			
USER_DEFINED_x	User defined parameter, where 'x' is replaced by a variable length user specified character string. Any number of user defined parameters may be included, if necessary to provide essential information that cannot be conveyed in COMMENT statements. Example: USER_DEFINED_EARTH_MODEL = WGS-84	n/a	No

**4.2.4.3** All values in the OMM are ‘at epoch’, i.e., the value of the parameter at the time specified in the EPOCH keyword.

**4.2.4.4** Table 4-3 is broken into five logical blocks, each of which has a descriptive heading. These descriptive headings shall not be included in an OMM, unless they appear in a properly formatted COMMENT statement.

**4.2.4.5** Values in the covariance matrix shall be expressed in the applicable reference frame (COV\_REF\_FRAME keyword if used, or REF\_FRAME keyword if not), and shall be presented sequentially from upper left [1,1] to lower right [6,6], lower triangular form, row by row left to right. Variance and covariance values shall be expressed in standard double precision as related in 6.4. This logical block of the OMM may be useful for risk assessment and establishing maneuver and mission margins.

**4.2.4.6** For operations in Earth orbit with a TLE-based OMM, some special conventions must be observed, as follows:

- The value associated with the CENTER\_NAME keyword shall be ‘EARTH’.
- The value associated with the REF\_FRAME keyword shall be ‘TEME’ (see ANNEX B, subsection B2).
- The value associated with the TIME\_SYSTEM keyword shall be ‘UTC’.
- The format of the OBJECT\_NAME and OBJECT\_ID keywords shall be that of the UN Office of Outer Space Affairs designator index (reference [2]).
- The MEAN\_MOTION keyword must be used instead of SEMI\_MAJOR\_AXIS.

**4.2.4.7** For those who wish to use the OMM to represent a TLE, there are a number of considerations that apply with respect to precision of angle representation, use of certain fields by the propagator, reference frame, etc. Some sources suggest the coding for the EPHEMERIS\_TYPE keyword: 1=SGP, 2=SGP4, 3=SDP4, 4=SGP8, 5=SDP8. Some sources suggest the following coding for the CLASSIFICATION\_TYPE keyword: U=unclassified, S=secret. (For further information see references [L3] and [L4])

**4.2.4.8** Maneuvers are not accommodated in the OMM. Users of the OMM who wish to model maneuvers may use several OMM files to describe the orbit at applicable epochs.

**4.2.4.9** A section of User Defined Parameters is allowed. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their meanings must be described in an ICD. User Defined Parameters, if included in an OMM, should be used as sparingly as possible; their use is not encouraged.

### 4.3 OMM EXAMPLES

Figure 4-2 and figure 4-3 are examples of OMMs based on the TLE shown in figure 4-1.

```

GOES 9 [P]
1 23581U 95025A 07064.44075725 -.00000113 00000-0 10000-3 0 9250
2 23581 3.0539 81.7939 0005013 249.2363 150.1602 1.00273272 43169
    
```

**Figure 4-1: Example Two Line Element Set (TLE)**

```

CCSDS_OMM_VERS = 2.0
CREATION_DATE  = 2007-065T16:00:00
ORIGINATOR     = NOAA/USA

OBJECT_NAME    = GOES 9
OBJECT_ID      = 1995-025A
CENTER_NAME    = EARTH
REF_FRAME      = TEME
TIME_SYSTEM    = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH          = 2007-064T10:34:41.4264
MEAN_MOTION    = 1.00273272
ECCENTRICITY    = 0.0005013
INCLINATION     = 3.0539
RA_OF_ASC_NODE = 81.7939
ARG_OF_PERICENTER = 249.2363
MEAN_ANOMALY   = 150.1602
GM              = 398600.8
EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID   = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH   = 4316
BSTAR          = 0.0001
MEAN_MOTION_DOT = -0.00000113
MEAN_MOTION_DDOT = 0.0
    
```

**Figure 4-2: OMM File Example without Covariance Matrix**

```

CCSDS_OMM_VERS = 2.0
CREATION_DATE = 2007-065T16:00:00
ORIGINATOR = NOAA/USA

OBJECT_NAME = GOES 9
OBJECT_ID = 1995-025A
CENTER_NAME = EARTH
REF_FRAME = TEME
TIME_SYSTEM = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH = 2007-064T10:34:41.4264
MEAN_MOTION = 1.00273272
ECCENTRICITY = 0.0005013
INCLINATION = 3.0539
RA_OF_ASC_NODE = 81.7939
ARG_OF_PERICENTER = 249.2363
MEAN_ANOMALY = 150.1602
GM = 398600.8

EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH = 4316
BSTAR = 0.0001
MEAN_MOTION_DOT = -0.00000113
MEAN_MOTION_DDOT = 0.0

COV_REF_FRAME = TEME
CX_X = 3.331349476038534e-04
CY_X = 4.618927349220216e-04
CY_Y = 6.782421679971363e-04
CZ_X = -3.070007847730449e-04
CZ_Y = -4.221234189514228e-04
CZ_Z = 3.231931992380369e-04
CX_DOT_X = -3.349365033922630e-07
CX_DOT_Y = -4.686084221046758e-07
CX_DOT_Z = 2.484949578400095e-07
CX_DOT_X_DOT = 4.296022805587290e-10
CY_DOT_X = -2.211832501084875e-07
CY_DOT_Y = -2.864186892102733e-07
CY_DOT_Z = 1.798098699846038e-07
CY_DOT_X_DOT = 2.608899201686016e-10
CY_DOT_Y_DOT = 1.767514756338532e-10
CZ_DOT_X = -3.041346050686871e-07
CZ_DOT_Y = -4.989496988610662e-07
CZ_DOT_Z = 3.540310904497689e-07
CZ_DOT_X_DOT = 1.869263192954590e-10
CZ_DOT_Y_DOT = 1.008862586240695e-10
CZ_DOT_Z_DOT = 6.224444338635500e-10
    
```

**Figure 4-3: OMM File Example with Covariance Matrix**

```

CCSDS_OMM_VERS = 2.0
CREATION_DATE  = 2007-065T16:00:00
ORIGINATOR     = NOAA/USA

OBJECT_NAME    = GOES 9
OBJECT_ID      = 1995-025A
CENTER_NAME    = EARTH
REF_FRAME      = TEME
TIME_SYSTEM    = UTC
MEAN_ELEMENT_THEORY = SGP/SGP4

EPOCH          = 2007-064T10:34:41.4264
MEAN_MOTION    = 1.00273272      [rev/day]
ECCENTRICITY   = 0.0005013
INCLINATION    = 3.0539          [deg]
RA_OF_ASC_NODE = 81.7939         [deg]
ARG_OF_PERICENTER = 249.2363     [deg]
MEAN_ANOMALY   = 150.1602       [deg]
GM             = 398600.8        [km**3/s**2]
EPHEMERIS_TYPE = 0
CLASSIFICATION_TYPE = U
NORAD_CAT_ID   = 23581
ELEMENT_SET_NO = 0925
REV_AT_EPOCH   = 4316
BSTAR          = 0.0001          [1/ER]
MEAN_MOTION_DOT = -0.00000113   [rev/day**2]
MEAN_MOTION_DDOT = 0.0           [rev/day**3]

USER_DEFINED_EARTH_MODEL = WGS-84

```

**Figure 4-4: OMM with Units and a User Defined Parameter**

## 5 ORBIT EPHEMERIS MESSAGE (OEM)

### 5.1 GENERAL

**5.1.1** Orbit information may be exchanged between two participants by sending an ephemeris in the form of a series of state vectors (Cartesian vectors providing position and velocity, and optionally accelerations) using an Orbit Ephemeris Message (OEM). The message recipient must have a means of interpolating across these state vectors to obtain the state at an arbitrary time contained within the span of the ephemeris.

**5.1.2** The OEM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and representing the orbits of active satellites, inactive man-made objects, near-Earth debris fragments, etc. The OEM reflects the dynamic modeling of any users' approach to conservative and non-conservative phenomena.

**5.1.3** The OEM shall be a plain text file consisting of orbit data for a single object. It shall be easily readable by both humans and computers.

**5.1.4** The OEM file-naming scheme should be agreed to on a case-by-case basis between the participants, typically using an ICD. The method of exchanging OEMs should be decided on a case-by-case basis by the participants and documented in an ICD.

NOTE – Detailed syntax rules for the OEM are specified in section 7.

### 5.2 OEM CONTENT/STRUCTURE

#### 5.2.1 GENERAL

**5.2.1.1** The OEM shall be represented as a combination of the following:

- a) a header;
- b) metadata (data about data);
- c) ephemeris data;
- d) optional covariance matrix data; and
- e) optional comments (explanatory information).

**5.2.1.2** OEM files must have a set of minimum required sections; some may be repeated. Table 5-1 outlines the contents of an OEM.

**Table 5-1: OEM File Layout Specifications**

<b>Required Sections</b>	Header Metadata Ephemeris Data (Appropriate comments should also be included, although they are not required.)
<b>Allowable Repetitions of Sections</b>	Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) Metadata Ephemeris Data Covariance Matrix (optional) ...etc. (Appropriate comments should also be included.)

**5.2.2 OEM HEADER**

**5.2.2.1** The OEM header assignments are shown in table 5-2, which specifies for each item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is mandatory or optional.

**5.2.2.2** Only those keywords shown in table 5-2 shall be used in an OEM header.



**Table 5-2: OEM Header**

<b>Keyword</b>	<b>Description</b>	<b>Examples of Values</b>	<b>Mandatory</b>
CCSDS_OEM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	2.0	Yes
COMMENT	Comments (allowed in the OEM Header only immediately after the OEM version number). (See 7.7 for formatting rules.)	COMMENT This is a comment	No
CREATION_DATE	File creation date and time in UTC. (For format specification, see 7.5.9.)	2001-11-06T11:17:33 2002-204T15:56:23	Yes
ORIGINATOR	Creating agency or operator (value should be drawn from the SANA "Organizations" registry). The country of origin should also be provided where the originator is not a national space agency.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT/USA, USAF, INMARSAT/UK	Yes

### 5.2.3 OEM METADATA

**5.2.3.1** The OEM metadata assignments are shown in table 5-3, which specifies for each item:

- a) the keyword to be used;
- b) a short description of the item;
- c) examples of allowed values; and
- d) whether the item is mandatory or optional.

**5.2.3.2** Only those keywords shown in table 5-3 shall be used in OEM metadata.

NOTE – For some keywords (OBJECT\_NAME, OBJECT\_ID, CENTER\_NAME) there are no definitive lists of authorized values maintained by a control authority; the references listed in 0 are the best known sources for authorized values to date. For the TIME\_SYSTEM and REF\_FRAME keywords, the approved values are listed in ANNEX B, subsection B1.

**5.2.3.3** A single metadata group shall precede each ephemeris data block. Multiple occurrences of a metadata group followed by an ephemeris data block may be used. Before each metadata group the string 'META\_START' shall appear on a separate line and after each metadata group (and before the associated ephemeris data block) the string 'META\_STOP' shall appear on a separate line.

**Table 5-3: OEM Metadata**

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Keyword	Description	Examples of Values	Mandatory
META_START	The OEM message contains metadata, ephemeris data, and covariance data; this keyword is used to delineate the start of a metadata block within the message (metadata are provided in a block, surrounded by 'META_START' and 'META_STOP' markers to facilitate file parsing). This keyword must appear on a line by itself.	n/a	Yes
COMMENT	Comments allowed only immediately after the META_START keyword. (See 7.7 for formatting rules.)	COMMENT This is a comment.	No
OBJECT_NAME	The name of the object for which the ephemeris is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended to use names from the UN Office of Outer Space Affairs designator index (reference [2]), which include Object name and international designator of the participant.	EUTELSAT W1 MARS PATHFINDER STS 106 NEAR	Yes
OBJECT_ID	Object identifier of the object for which the ephemeris is provided. There is no CCSDS-based restriction on the value for this keyword, but it is recommended that values be the international spacecraft designator as published in the UN Office of Outer Space Affairs designator index (reference [2]). Recommended values have the format YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three-digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch. In cases where the asset is not listed in reference [2], or the UN Office of Outer Space Affairs designator index format is not used, the value should be provided in an ICD.	2000-052A 1996-068A 2000-053A 1996-008A	Yes
CENTER_NAME	Origin of reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, or another spacecraft (in this case the value for 'CENTER_NAME' is subject to the same rules as for 'OBJECT_NAME'). There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).	EARTH EARTH BARYCENTER MOON SOLAR SYSTEM BARYCENTER SUN JUPITER BARYCENTER STS 106 EROS	Yes

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Keyword	Description	Examples of Values	Mandatory
REF_FRAME	Name of the reference frame in which the ephemeris data are given. Use of values other than those in ANNEX B, subsections B2 must be documented and conveyed in an ICD. The reference frame must be the same for all data elements, with the exception of the covariance matrix, for which an applicable different reference frame may be specified.	ICRF ITRF-93 ITRF-97 ITRF2000 ITRFxxxx (template for future versions) TOD (True Equator and Equinox of Date) EME2000 (Earth Mean Equator and Equinox of J2000) TDR (true of date rotating) GRC (Greenwich rotating coordinate frame, another name for TDR)	Yes
REF_FRAME_EPOCH	Epoch of reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)	2001-11-06T11:17:33 2002-204T15:56:23Z	No
TIME_SYSTEM	Time system used for metadata, ephemeris data, and covariance data. Use of values other than those in ANNEX B, subsection B1 must be documented and conveyed in an ICD.	UTC, TAI, TT, GPS, TDB, TCB	Yes
START_TIME	Start of TOTAL time span covered by ephemeris data and covariance data immediately following this metadata block. (For format specification, see 7.5.9.)	1996-12-18T14:28:15.1172 1996-277T07:22:54	Yes
USEABLE_START_TIME USEABLE_STOP_TIME	Optional start and end of USEABLE time span covered by ephemeris data immediately following this metadata block. To allow for proper interpolation near the ends of the ephemeris data block it may be necessary, depending upon the interpolation method to be used, to utilize these keywords with values within the time span covered by the ephemeris data records as denoted by the START/STOP_TIME time tags. (For format specification, see 7.5.9.) These keywords are optional items, and thus may not be necessary, depending on the recommended interpolation method. However, it is recommended to use the USEABLE_START_TIME and USEABLE_STOP_TIME capability in all cases. The USEABLE_START_TIME time tag at a new block of ephemeris data must be greater than or equal to the USEABLE_STOP_TIME time tag of the previous block.	1996-12-18T14:28:15.1172 1996-277T07:22:54	No
STOP_TIME	End of TOTAL time span covered by ephemeris data and covariance data immediately following this metadata block. (For format specification, see 7.5.9.)	1996-12-18T14:28:15.1172 1996-277T07:22:54	Yes
INTERPOLATION	This keyword may be used to specify the recommended interpolation method for ephemeris data in the immediately following set of ephemeris lines.	Hermite Linear Lagrange	No

Keyword	Description	Examples of Values	Mandatory
INTERPOLATION_DEGREE	Recommended interpolation degree for ephemeris data in the immediately following set of ephemeris lines. Must be an integer value. This keyword must be used if the 'INTERPOLATION' keyword is used.	5 1	No
META_STOP	The OEM message contains metadata, ephemeris data, and covariance data; this keyword is used to delineate the end of a metadata block within the message (metadata are provided in a block, surrounded by 'META_START' and 'META_STOP' markers to facilitate file parsing). This keyword must appear on a line by itself.	n/a	Yes

## 5.2.4 OEM DATA: EPHEMERIS DATA LINES

**5.2.4.1** Each set of ephemeris data, including the time tag, must be provided on a single line. The order in which data items are given shall be fixed: **Epoch, X, Y, Z, X\_DOT, Y\_DOT, Z\_DOT, X\_DDOT, Y\_DDOT, Z\_DDOT**.

**5.2.4.2** The position and velocity terms shall be mandatory; acceleration terms may be provided.

**5.2.4.3** At least one space character must be used to separate the items in each ephemeris data line.

**5.2.4.4** Repeated time tags may occur in consecutive ephemeris data blocks if the STOP\_TIME of the first ephemeris data block is greater than the START\_TIME of the second ephemeris data block. Although the USEABLE\_STOP\_TIME and USEABLE\_START\_TIME of the consecutive ephemeris data blocks must not overlap (except for a possibly shared endpoint), the STOP\_TIME of the first ephemeris data block may be greater than the START\_TIME of the second ephemeris data block if the extra data is required for interpolation purposes.

**5.2.4.5** The TIME\_SYSTEM value must remain fixed within an OEM.

**5.2.4.6** The occurrence of a second (or greater) metadata block after some ephemeris data indicates that interpolation using succeeding ephemeris data with ephemeris data occurring prior to that metadata block shall not be done. This method may be used for proper modeling of propulsive maneuvers or any other source of a discontinuity such as eclipse entry or exit.

**5.2.4.7** Details about interpolation method should be specified using the INTERPOLATION and INTERPOLATION\_DEGREE keywords within the OEM. All data blocks must contain a sufficient number of ephemeris data records to allow the recommended interpolation method to be carried out consistently throughout the OEM.

### 5.2.5 OEM DATA: COVARIANCE MATRIX LINES

**5.2.5.1** A single covariance matrix data section may optionally follow each ephemeris data block.

**5.2.5.2** If present, the covariance matrix data lines in the OEM are separated from the ephemeris data by means of two new keywords: `COV_START` and `COVARIANCE_STOP`. The '`COV_START`' keyword must appear before the first line of the covariance matrix data. The '`COVARIANCE_STOP`' keyword must appear after the last line of covariance data. Each of these keywords shall appear on a line by itself with no time tags or values.

**5.2.5.3** The epoch of the navigation solution related to the covariance matrix must be provided via the '`EPOCH`' keyword. The reference frame of the covariance matrix, if different from that of the states in the ephemeris, must be provided via the '`COV_REF_FRAME`' keyword.

**5.2.5.4** Each row of the 6x6 lower triangular covariance matrix must be provided on a single line. The order in which data items are given shall be fixed. The elements in each row of covariates shall be defined by the order in the ephemeris data line (i.e., **X, Y, Z, X\_DOT, Y\_DOT, Z\_DOT**). The six rows of the covariance matrix contain from one to six numbers depending on what row of the matrix is being represented (first row has one element, second row has two, continuing in this fashion until the sixth row has six elements).

**5.2.5.5** At least one space character must be used to separate the items in each covariance matrix data line.

**5.2.5.6** Multiple covariance matrices may appear in the covariance matrix section; they may appear with any desired frequency (one for each navigation solution that makes up the overall ephemeris is recommended). The OEM may also contain propagated covariances, not just individual covariances associated with navigation solutions.

**5.2.5.7** If there are multiple covariance matrices in the data section, they must be ordered by increasing time tag.

### 5.3 OEM EXAMPLES

Figure 5-1, figure 5-2, and figure 5-3 are example OEMs. Some ephemeris data lines have been omitted to save space.

Figure 5-1 is compatible with ODM version 1, and thus could use either '`CCSDS_OEM_VERS = 1.0`' (since it does not contain any of the unique features of the ODM version 2), or '`CCSDS_OEM_VERS = 2.0`' (as shown). Figure 5-2 and figure 5-3 contain features unique to the ODM version 2, and thus '`CCSDS_OEM_VERS = 2.0`' must be specified.

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OEM_VERS = 2.0
CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME     = MARS BARYCENTER
REF_FRAME       = EME2000
TIME_SYSTEM     = UTC
START_TIME      = 1996-12-18T12:00:00.331
USEABLE_START_TIME = 1996-12-18T12:10:00.331
USEABLE_STOP_TIME  = 1996-12-28T21:23:00.331
STOP_TIME       = 1996-12-28T21:28:00.331
INTERPOLATION   = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This file was produced by M.R. Somebody, MSOO NAV/JPL, 1996NOV 04. It is
COMMENT to be used for DSN scheduling purposes only.

1996-12-18T12:00:00.331  2789.619 -280.045 -1746.755  4.73372 -2.49586 -1.04195
1996-12-18T12:01:00.331  2783.419 -308.143 -1877.071  5.18604 -2.42124 -1.99608
1996-12-18T12:02:00.331  2776.033 -336.859 -2008.682  5.63678 -2.33951 -1.94687

    < intervening data records omitted here >

1996-12-28T21:28:00.331 -3881.024  563.959 -682.773  -3.28827 -3.66735  1.63861

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME     = MARS BARYCENTER
REF_FRAME       = EME2000
TIME_SYSTEM     = UTC
START_TIME      = 1996-12-28T21:29:07.267
USEABLE_START_TIME = 1996-12-28T22:08:02.5
USEABLE_STOP_TIME  = 1996-12-30T01:18:02.5
STOP_TIME       = 1996-12-30T01:28:02.267
INTERPOLATION   = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This block begins after trajectory correction maneuver TCM-3.

1996-12-28T21:29:07.267 -2432.166 -063.042 1742.754  7.33702 -3.495867 -1.041945
1996-12-28T21:59:02.267 -2445.234 -878.141 1873.073  1.86043 -3.421256 -0.996366
1996-12-28T22:00:02.267 -2458.079 -683.858 2007.684  6.36786 -3.339563 -0.946654

    < intervening data records omitted here >

1996-12-30T01:28:02.267 2164.375 1115.811 -688.131  -3.53328 -2.88452  0.88535

```

**Figure 5-1: Version 1 OEM Compatible Example (No Acceleration, No Covariance)**

## CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```
CCSDS_OEM_VERS = 2.0

COMMENT  OEM WITH OPTIONAL ACCELERATIONS MUST BE OEM VERSION 2.0

CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME      = MARS GLOBAL SURVEYOR
OBJECT_ID        = 1996-062A
CENTER_NAME      = MARS BARYCENTER
REF_FRAME        = EME2000
TIME_SYSTEM      = UTC
START_TIME       = 1996-12-18T12:00:00.331
USEABLE_START_TIME = 1996-12-18T12:10:00.331
USEABLE_STOP_TIME  = 1996-12-28T21:23:00.331
STOP_TIME        = 1996-12-28T21:28:00.331
INTERPOLATION    = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT  This file was produced by M.R. Somebody, MSO NAV/JPL, 2000 NOV 04. It is
COMMENT  to be used for DSN scheduling purposes only.

1996-12-18T12:00:00.331  2789.6 -280.0 -1746.8  4.73 -2.50 -1.04  0.008 0.001 -0.159
1996-12-18T12:01:00.331  2783.4 -308.1 -1877.1  5.19 -2.42 -2.00  0.008 0.001  0.001
1996-12-18T12:02:00.331  2776.0 -336.9 -2008.7  5.64 -2.34 -1.95  0.008 0.001  0.159

  < intervening data records omitted here >

1996-12-28T21:28:00.331 -3881.0  564.0 -682.8 -3.29 -3.67  1.64 -0.003 0.000  0.000
```

**Figure 5-2: Version 2 OEM Example with Optional Accelerations**



CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OEM_VERS = 2.0
CREATION_DATE = 1996-11-04T17:22:31
ORIGINATOR = NASA/JPL

META_START
OBJECT_NAME           = MARS GLOBAL SURVEYOR
OBJECT_ID             = 1996-062A
CENTER_NAME           = MARS BARYCENTER
REF_FRAME             = EME2000
TIME_SYSTEM           = UTC
START_TIME            = 1996-12-28T21:29:07.267
USEABLE_START_TIME    = 1996-12-28T22:08:02.5
USEABLE_STOP_TIME     = 1996-12-30T01:18:02.5
STOP_TIME             = 1996-12-30T01:28:02.267
INTERPOLATION         = HERMITE
INTERPOLATION_DEGREE = 7
META_STOP

COMMENT This block begins after trajectory correction maneuver TCM-3.

1996-12-28T21:29:07.267 -2432.166 -063.042 1742.754 7.33702 -3.495867 -1.041945
1996-12-28T21:59:02.267 -2445.234 -878.141 1873.073 1.86043 -3.421256 -0.996366
1996-12-28T22:00:02.267 -2458.079 -683.858 2007.684 6.36786 -3.339563 -0.946654

    < intervening data records omitted here >

1996-12-30T01:28:02.267 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535

COV_START
EPOCH = 1996-12-28T21:29:07.267
COV_REF_FRAME = EME2000
 3.3313494e-04
 4.6189273e-04 6.7824216e-04
-3.0700078e-04 -4.2212341e-04 3.2319319e-04
-3.3493650e-07 -4.6860842e-07 2.4849495e-07 4.2960228e-10
-2.2118325e-07 -2.8641868e-07 1.7980986e-07 2.6088992e-10 1.7675147e-10
-3.0413460e-07 -4.9894969e-07 3.5403109e-07 1.8692631e-10 1.0088625e-10 6.2244443e-10

EPOCH = 1996-12-29T21:00:00
COV_REF_FRAME = EME2000
 3.4424505e-04
 4.5078162e-04 6.8935327e-04
-3.0600067e-04 -4.1101230e-04 3.3420420e-04
-3.2382549e-07 -4.5750731e-07 2.3738384e-07 4.3071339e-10
-2.1007214e-07 -2.7530757e-07 1.6870875e-07 2.5077881e-10 1.8786258e-10
-3.0302350e-07 -4.8783858e-07 3.4302008e-07 1.7581520e-10 1.0077514e-10 6.2244443e-10
COVARIANCE_STOP
    
```

**Figure 5-3: Version 2 OEM Example with Optional Covariance Matrices**

## **6 ORBIT COMPREHENSIVE MESSAGE (OCM)**

### **6.1 GENERAL**

**6.1.1** Comprehensive orbit information may be exchanged between two participants by sending orbit data/content for one or more epochs using an Orbit Comprehensive Message (OCM). The OCM aggregates and extends OMM, OPM and OEM content in a single hybrid message. The OCM simultaneously emphasizes flexibility and message conciseness by offering extensive optional standardized content while minimizing mandatory content.

**6.1.2** The OCM shall be a plain text file consisting of orbit data for a single space object, or in the case of a parent/child satellite deployment scenario, a single parent object. It shall be easily readable by both humans and computers.

**6.1.3** The OCM file-naming scheme should be agreed to on a case-by-case basis between the exchange partners, and should be documented in an ICD. The method of exchanging OCMs should be decided on a case-by-case basis by the exchange partners and documented in an ICD.

**6.1.4** Orbit information may be exchanged between two participants by sending an ephemeris in the form of one or more time series of orbital states (selectable as orbital elements and/or Cartesian vectors providing position and optionally velocity and accelerations) using an Orbit Comprehensive Message (OCM). If orbital states are desired at arbitrary time(s) contained within the span of the ephemeris, the message recipient is encouraged to use a suitable interpolation method. For times outside of supplied orbit state time spans or if the step size between orbit states is too large to support interpolation [L8], optional perturbations parameters should be included with this message and the recipient must have a suitably-compatible orbit propagator.

**6.1.5** The OCM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and representing the orbits of active satellites, inactive man-made objects, near-Earth debris fragments, etc. The OCM reflects the dynamic modeling of any users' approach to conservative and non-conservative phenomena.

NOTE – Detailed syntax rules for the OCM are specified in section 7.

### **6.2 OCM CONTENT/STRUCTURE**

#### **6.2.1 GENERAL**

The OCM shall be represented as a combination of the following as shown in Table 6-1. The ordering of these sections is mandatory. The order of occurrence of the OCM sections shall be fixed as shown in table 6-1.

- 1) a single mandatory header;
- 2) a single mandatory metadata section (data about data);

- 3) optional data section(s), comprised of one or more data constituent types:
  - a. a single, optional space object physical characteristics section
  - b. a single, optional perturbations section
  - c. optional maneuver data section(s)
  - d. optional orbit state time histories
  - e. a single, optional orbit determination data section
  - f. optional covariance time histories
  - g. optional State Transition Matrix (STM) time histories
  - h. one or more optional Ephemeris Compression (EC) time segments
  - i. a single, optional, user-defined data and supplemental comments (explanatory information).

**Table 6-1: OCM File Layout and Ordering Specification**

<b>Section</b>	<b>Content</b>
Mandatory Header	Header of message
Mandatory Metadata	Metadata (Informational comments recommended but not required.)
Optional Space Object Physical Description	Optional space object physical characteristics, if known.
Optional Perturbations Section	Optional perturbations parameters (Accompanying informational comment(s) encouraged.)
Optional Maneuver Section(s)	Optional maneuver specifications for either impulsive or finite burns or acceleration profiles
Optional Orbit Data Section(s)	Optional: One or more orbit state time histories (each consisting of one or more orbit states)
Optional Orbit Determination Section	Optional orbit determination data section
Optional Covariance Data Section(s)	Optional: One or more covariance time histories (each consisting of one or more covariance matrices)
Optional State Transition Matrix Data Section(s)	Optional: One or more state transition matrix time histories (each consisting of one or more state transition matrices)
Optional Ephemeris Compression Data Section(s)	Optional: One or more ephemeris compression sections (each consisting of one or more ephemeris compression segments)
Optional Attitude section(s)	Optional: One or more attitude time histories (each consisting of one or more attitude states)

## 6.2.2 OCM HEADER

6.2.2.1 Table 6-2 specifies the keywords for each header item.

6.2.2.2 Only those keywords shown in table 3-1 shall be used in an OCM header.

6.2.2.3 The order of occurrence of these OCM header keywords shall be fixed as shown in table 6-2.

**Table 6-2: OCM Header**

Keyword	Description	Examples of Values	Mandatory
CCSDS_OCM_VERS	Format version in the form of 'x.y', where 'y' is incremented for corrections and minor changes, and 'x' is incremented for major changes.	1.0	Yes
COMMENT	Comments (allowed in the OCM Header only immediately after the OCM version number). (See 7.7 for formatting rules.)	COMMENT This is a comment	No
CREATION_DATE	File creation date/time in UTC. (For format specification, see 7.5.9.)	2001-11-06T11:17:33 2002-204T15:56:23Z	Yes
ORIGINATOR	Creating agency or operator (value should be drawn from the SANA "Organizations" registry). The country of origin should also be provided where the originator is not a national space agency.	CNES, ESOC, GSFC, GSOC, JPL, JAXA, INTELSAT/USA, USAF, INMARSAT/UK	Yes
MESSAGE_ID	ID that uniquely identifies a message from a given originator. The format and content of the message identifier value are at the discretion of the originator.	201113719185 ABC-12_34	No

## 6.2.3 OCM METADATA

6.2.3.1 Table 6-3 specifies the metadata keywords. Only those keywords shown in table 6-3 shall be used in OCM metadata.

6.2.3.2 The "OCM Metadata" section is mandatory; "mandatory" in the context of Table 6-3 denotes those keywords which must be included in this section.

6.2.3.3 Only one OCM Metadata section (data block) shall appear in an OCM.

6.2.3.4 OCM Metadata data block

6.2.3.5 The order of occurrence of these OCM metadata keywords shall be fixed as shown in table 6-3, with the exception that comments may be interspersed throughout the metadata section as required.

6.2.3.6 The TIME\_SYSTEM value must remain fixed within an OCM.

**6.2.3.7** Any spacecraft physical characteristics, maneuver, orbit states, covariance and STM values in the OCM data which require time-tagging shall be time-tagged by a relative time value measured with respect to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.3.8** The OCM shall only contain a single metadata section in the entire scope of the message.

NOTE – For some keywords (OBJECT\_NAME, OBJECT\_ID) there are no definitive lists of authorized values maintained by a control authority; the references listed in 0 are the best known sources for authorized values to date.

NOTE 2 – While specification of OBJECT\_DESIGNATOR, OBJECT\_NAME and INTL\_DESIGNATOR are each in and of themselves optional, one of these keywords must be supplied.

NOTE 3 – The only metadata fields which are relied upon by the subsequent optional OCM message subtypes (e.g. maneuver data,

**Table 6-3: OCM Metadata**

Keyword	Description	Examples of Values	Mandatory	Any OCM sections relying upon this field ?
COMMENT	Comments (allowed at any point(s) throughout the OCM Metadata section). (See 7.7 for formatting rules.)	COMMENT This is a comment	No	no
ORIGINATOR_POC	Free text field containing Programmatic or Technical Point-of-Contact (PoC) for OCM	Mr. Rodgers	No	no
ORIGINATOR_PHONE	Free text field containing PoC phone number	+49615130312	No	no
ORIGINATOR_POSITION	Free text field containing contact position of the PoC	Flight Dynamicist Mission Design Lead	No	no
ORIGINATOR_ADDRESS	Free text field containing Technical PoC information for OCM creator (suggest email, website, or physical address, etc.)	JOHN.DOE@ SOMEWHERE.NET	No	no
OBJECT_DESIGNATOR	The satellite catalog designator for the object.	22444	No	no
DESIGNATOR_ID_SOURCE	Specification of the catalog (or source organization) from which the OBJECT_DESIGNATOR was drawn. This is a free-text field.	JSPOC, ISON, ESA, COMSPOC, etc.	No	no
OBJECT_NAME	Spacecraft name for the object.	SPOT, ENVISAT, IRIDIUM, INTELSAT	No	no

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Keyword	Description	Examples of Values	Mandatory	Any OCM sections relying upon this field ?
INTL_DESIGNATOR	The full international designator for the object. Values shall have the format: YYYY-NNNP{PP}, where: YYYY = Year of launch. NNN = Three digit serial number of launch in year YYYY (with leading zeros). P{PP} = At least one capital letter for the identification of the part brought into space by the launch. In cases where the object has no international designator, the value UNKNOWN may be used.	2000-052A 1996-068A 2000-053A 1996-008A	No	no
OBJECT_TYPE	The object type.	PAYLOAD ROCKET BODY UPPER STAGE DEBRIS UINKNOWN OTHER	No	no
EPOCH_TZERO	Epoch from which all OCM relative times are referenced. (For format specification, see 7.5.9). The time scale EPOCH_TZERO is the one specified by "TIME_SYSTEM" keyword in the metadata section.	2001-11-06T00:00:00	Yes	MNVR, STATES, COVAR STM, EC
INCL_DATA_BLOCKS	Comma-delimited list of data blocks included in this message.	MNVR, ORB, COV, OD, PHYSCHAR, PERTS, STM, EC, ATT, USER	No	no
START_TIME	Relative time of the earliest of all time tags corresponding to maneuver, orbital state, covariance, and/or STM data. The epoch is specified in timing system "TIME_SYSTEM"  (For format specification, see 7.5.9 for absolute time format; relative time is measured in seconds from EPOCH_TZERO)	100.0	No	no
STOP_TIME	Relative time of the end of TOTAL time span covered by ALL maneuver, orbital state, covariance and/or STM data contained in this message.  (For format specification, see 7.5.9 for absolute time format; relative time is measured in seconds from EPOCH_TZERO)	1500.0	No	no
TAIMUTC_TZERO	Difference (TAI – UTC) in seconds (i.e. total # leap seconds elapsed since 1958) as modeled by the message originator at epoch "EPOCH_TZERO".	36 [s]	No	no
TIME_SYSTEM_ABS	Timing system used for the absolute time contained in EPOCH_TZERO. <b>Omission of this non-mandatory field defaults to "UTC"</b>	UTC	No	MNVR, STATES, COVAR STM, EC
TIME_SYSTEM_REL	Timing system used for all relative time specifications relative to EPOCH_TZERO. <b>Omission of this non-mandatory field defaults to "UTC"</b>	UTC TAI	No	MNVR, STATES, COVAR STM, EC
UT1MUTC_TZERO	Difference (UT1 – UTC) in seconds, as modeled by the originator at epoch "EPOCH_TZERO".	0.357 [s]	No	no

Keyword	Description	Examples of Values	Mandatory	Any OCM sections relying upon this field ?
UT1MUTC_RATE_TZERO	Rate-of-change of (UT1 – UTC) in milliseconds per day, as modeled by the originator at epoch “EPOCH_TZERO”	.0001 [ms/day]	No	no

## 6.2.4 OCM DATA: SPACE OBJECT PHYSICAL CHARACTERISTICS

**6.2.4.1** Table 6-4 gives an overview of the OCM space object physical characteristics section. Only those keywords shown in table 6-4 shall be used in OCM space object physical characteristics data.

**6.2.4.2** Keyword values shall be provided in the units specified in column three of Table 6-4.

**6.2.4.3** The order of occurrence of these OCM Space Objects Physical Characteristics keywords shall be fixed as shown in table 6-4, with the exception that comments may be interspersed throughout the this section as required.

**6.2.4.4** The “OCM Data: Space Object Physical Characteristics” section is optional; “mandatory” in the context of Table 6-4 denotes those keywords which must be included in this section if this section is included.

**6.2.4.5** Only one space object physical characteristics section shall appear in an OCM.

**6.2.4.6** The space object physical characteristics data section in the OCM shall be indicated by two keywords: PHYS\_START and PHYS\_STOP.

**6.2.4.7** Further definition of Space Object Physical Characteristics parameters is provided in ANNEX C.

**Table 6-4: OCM Data: Space Object Physical Characteristics**

Keyword	Description	Units	Examples of Values	Mandatory
COMMENT	Comments (allowed at any point(s) throughout the OCM Space Object Physical Characteristics). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
PHYS_START	Start of a Space Object Physical Characteristics specification	n/a		Yes
DRAG_AREA	Additional Drag Area ( $A_D$ ) facing the relative wind vector, not already incorporated into the attitude-dependent “AREA_ALONG_OEB” parameters	m**2	2.5	No
DRAG_COEFF	Drag Coefficient ( $C_D$ )	n/a	2.2	No
DRAG_SCALE	Drag scale factor (1.0 represents no scaling). This factor is intended to allow operators to supply the nominal ballistic coefficient components while accommodating ballistic coefficient uncertainties (i.e. 1.06 represents a +6 percent error)  <b>Omission of this non-mandatory field defaults to 1.0.</b>	n/a	1.0	No



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Keyword	Description	Units	Examples of Values	Mandatory
MASS	S/C Mass at the reference epoch "EPOCH_TZERO"	kg	500	No
OEB_FRAME	Name of the reference frame which maps to the Optimally-Encompassing Box (OEB) frame via the Euler sequence OEB_PITCH, OEB_ROLL and OEB_YAW. <b>Allowable values include all entries contained in ANNEX B, subsections B2 and B3, as well as SC_BODY or a unique ID as documented and conveyed in an ICD.</b>  <b>Omission of this non-mandatory field defaults to RIC.</b>	n/a	ITRF-97	No
OEB_FRAME_EPOCH	Epoch of the OEB reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  <b>Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to the time stored in EPOCH_TZERO.</b>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
AREA_ALONG_OEB_MAX	Cross-sectional area of space object when viewed along max OEB ( $\hat{X}_{OEB}$ ) direction as defined in ANNEX C	m**2	0.15	No
AREA_ALONG_OEB_MED	Cross-sectional area of space object when viewed along medium OEB ( $\hat{Y}_{OEB}$ ) direction as defined in ANNEX C	m**2	0.3	No
AREA_ALONG_OEB_MIN	Cross-sectional area of space object when viewed along minimum OEB ( $\hat{Z}_{OEB}$ ) direction as defined in ANNEX C	m**2	0.5	No
OEB_MAX	Maximum physical dimension (along $\hat{X}_{OEB}$ ) of the Optimally-Encompassing Box (OEB) in meters,	m	1	No
OEB_MED	Medium physical dimension (along $\hat{Y}_{OEB}$ ) of Optimally-Encompassing Box (OEB) normal to OEB_MAX direction	m	0.5	No
OEB_MIN	Minimum physical dimension (along $\hat{Z}_{OEB}$ ) of Optimally-Encompassing Box (OEB) in direction normal to both OEB_MAX and OEB_MED directions	m	0.3	No
OEB_YAW	Yaw angle of the (Yaw/Pitch/Roll ordered sequence) that maps from the OEB_FRAME (defined above) to the frame aligned with the optimally-Encompassing Box (defined in ANNEX C).. A value of "-999" denotes a tumbling space object.	deg	30	No
OEB_PITCH	Pitch angle of the (Yaw/Pitch/Roll ordered sequence) that maps from the OEB_FRAME (defined above) to the frame aligned with the optimally-Encompassing Box (defined in ANNEX C).. A value of "-999" denotes a tumbling space object.	deg	1.7	No
OEB_ROLL	Roll angle of the (Yaw/Pitch/Roll ordered sequence) that maps from the OEB_FRAME (defined above) to the frame aligned with the optimally-Encompassing Box (defined in ANNEX C).. A value of "-999" denotes a tumbling space object.	deg	-10	No
RCS	Effective Radar Cross Section of the object	m**2	1.0	No

Keyword	Description	Units	Examples of Values	Mandatory
SOLAR_RAD_AREA	Additional total Solar Radiation Pressure Area ( $A_R$ ) facing the Sun, not already incorporated into the attitude-dependent “AREA_ALONG_OEB” parameters (computed from  $\{ \text{AREA\_ALONG\_OEB\_MAX} \cos(\theta_1) + \text{AREA\_ALONG\_OEB\_MED} \cos(\theta_2) + \text{AREA\_ALONG\_OEB\_MIN} \cos(\theta_3) \}$  Where $\theta_i$ represents the angle between the normal to each MAX/MED/MIN face and the direction to the Sun.	m**2	1.0	No
SOLAR_RAD_COEFF	Solar Radiation Pressure Coefficient ( $C_R$ )	n/a	1.7	No
SOLAR_RAD_SCALE	Solar Radiation Pressure scale factor (1.0 represents no scaling)	n/a	1.0	No
VM_ABS	Absolute Visual Magnitude “normalized” as discussed in ANNEX D to a 1 AU Sun-to-target distance, a phase angle of 0° and a 40,000 km target-to-sensor distance (equivalent of GEO satellite tracked at 15.6° above local horizon)	n/a	15.0	No
IXX	Moment of Inertia about the X-axis of the spacecraft’s primary body frame (e.g. SC_Body_1)	kg*m**2	1000.0	No
IYY	Moment of Inertia about the Y-axis	kg*m**2	800.0	No
I33IZZ	Moment of Inertia about the Z-axis	kg*m**2	400.0	No
I12IXY	Inertia Cross Product of the X & Y axes	kg*m**2	20.0	No
I13IXZ	Inertia Cross Product of the X & Z axes	kg*m**2	40.0	No
I23IYZ	Inertia Cross Product of the Y & Z axes	kg*m**2	60.0	No
PHYS_STOP	End of a Space Object Physical Characteristics specification	n/a		Yes

**6.2.5 OCM DATA: PERTURBATIONS SPECIFICATION**

**6.2.5.1** Table 6-5 provides an overview of the OCM Perturbations Specification section. Only those keywords shown in table 6-5 shall be used in OCM perturbations specification.

**6.2.5.2** Keyword values shall be provided in the units specified in column three of Table 6-5.

**6.2.5.3** The order of occurrence of these OCM Perturbations Specification keywords shall be fixed as shown in table 6-5, with the exception that comments may be interspersed throughout the this section as required.

**6.2.5.4** The OCM Perturbations Specification section is optional; “mandatory” in the context of Table 6-5 denotes those keywords which must be included in this section if this section is included.

**6.2.5.5** Only one OCM Perturbations Specification section shall appear in an OCM.

**6.2.5.6** The OCM Perturbations Specification section shall be delineated by two keywords: PERT\_START and PERT\_STOP.

**6.2.5.7** If the solar radiation coefficient,  $C_R$ , is set to zero, no solar radiation pressure shall be taken into account (see description of SOLAR\_RAD\_COEFF in Table 6-4).

**6.2.5.8** If the atmospheric drag coefficient,  $C_D$ , is set to zero, no atmospheric drag shall be taken into account (see description of DRAG\_COEFF in Table 6-4).

**Table 6-5: OCM Data: Perturbations Specification**

Keyword	Description	Units	Examples of Values	Mandatory
COMMENT	Comments (allowed at any point(s) throughout the OCM Perturbations Specification Data). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
PERT_START	Start of the perturbations specification	n/a		Yes
ALBEDO	Name of the albedo model	n/a		No
ALBEDO_GRID_SIZE	# of grid points used in the albedo model	n/a		No
ATMOSPHERIC_MODEL	Name of atmosphere model	n/a	MSISE90 NRLMSIS00 J70 J71 JRob DTM ...	No
CENTRAL_BODY_ROTATION	Central body angular rotation rate, measured about the major principal axis of the inertia tensor of the central body, relating the True-of-Date and pseudo Earth-fixed frames (not accounting for polar motion).	deg/s	4.17807421629e-3	No
D_NUTATION_DEPS	Correction to Nutation in obliquity $\delta_{deps}$ to maintain compatibility with the ICRS.	arcsec	-0.003 875	
D_NUTATION_DPSI	Correction to Nutation in longitude $\delta_{dpsi}$ to maintain compatibility with the ICRS.	arcsec	-0.052 195	
DX	Free core nutation and time dependent corrections for the X coordinate of the CIP in the ICRS, at EPOCH_TZERO	arcsec	-0.000 205	No
DY	Free core nutation and time dependent corrections for the Y coordinate of the CIP in the ICRS, at EPOCH_TZERO	arcsec	-0.000 136	No
EOP_SOURCE	Source of originator's Earth orientation parameters. This is a free text field, so if the examples on the right are insufficient, others may be used.	n/a	IERS USNO NGA ...	No
EQUATORIAL_RADIUS	Oblate spheroid equatorial radius	km	6378.137	No
GM	Gravitational Coefficient of attracting body (Gravitational Constant x Central Mass)	km**3/s**2	398600.4	No
GRAVITY_MODEL	The name of the geopotential model for central body, followed by the degree and order of the spherical harmonic coefficients applied. Note that specifying a zero value for "order" (i.e. 2 0) denotes zonals ( $J_2 \dots J_D$ ) only.	n/a	EGM-96: 36 36 WGS-84: 8 8 GGM-01: 12 12 TEG-4: 8 2	No
INTERP_METHOD_EOP	Used for EOP data	n/a	LINEAR	No
INTERP_METHOD_SPWX	Used for Space Weather data (SOLAR_F10P7, SOLAR_F10P7_MEAN and KSUBP)	n/a	NONE LINEAR	No
KSUBP	Planetary 3-hour-range Geomagnetic index Kp at EPOCH_TZERO	Kp units	3.2	No
N_BODY_PERTURBATIONS	N-body gravitational perturbations used, each separated by a comma	n/a	MOON, SUN, JUPITER	No

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Keyword	Description	Units	Examples of Values	Mandatory
NUTATION_DEPS	Nutation in obliquity $d\epsilon$ for 1980 IAU Theory of Nutation model, at EPOCH_TZERO	deg	0.002 031 6	No
NUTATION_DPSI	Nutation in longitude $d\psi$ for 1980 IAU Theory of Nutation model, at EPOCH_TZERO	deg	-0.003 410 8	No
OBLATE_FLATTENING	Oblate spheroid oblateness for the polar-symmetric oblate central body model	n/a	1/298.257223563	No
OCEAN_TIDES_MODEL	Name of ocean tides model (optionally specify order or constituent effects (diurnal, semi-diurnal, etc.))	n/a	diurnal	No
POLAR_MOTION_XP	Polar motion coordinate Xp of the Celestial Intermediate Pole at EPOCH_TZERO	arcsec		No
POLAR_MOTION_YP	Polar motion coordinate Yp of the Celestial Intermediate Pole at EPOCH_TZERO	arcsec		No
REDUCTION_THEORY	Specification of the reduction theory used for precession and nutation modeling. This is a free text field, so if the examples on the right are insufficient, others may be used.	n/a	IAU1976/FK5 IAU2010 IERS1996	No
S_PRECNUT	The S parameter provides the position of the Celestial Intermediate Origin (CIO) on the equator of the Celestial Intermediate Pole (CIP) corresponding to the kinematical definition of the [non-rotating origin] in the GCRS when the CIP is moving with respect to the GCRS between the reference epoch and the epoch due to precession and nutation (McCarthy and Petit 2003). (Vallado 2013:214)	arcsec	-0.003 021	No
SHADOW_MODEL	Shadow modeling for Solar Radiation Pressure; dual cone uses both umbra/penumbra regions	n/a	NONE CYLINDRICAL DUAL CONE	No
SOLAR_F10P7	Solar flux proxy F10.7 at EPOCH_TZERO	Solar Flux Units = $10^4$ Jansky= $10^{-22}$ W/(m <sup>2</sup> *Hz)	120.0	No
SOLAR_F10P7_MEAN	Centered-average solar flux proxy $\overline{F10.7}$ at EPOCH_TZERO	Solar Flux Units = $10^4$ Jansky= $10^{-22}$ W/(m <sup>2</sup> *Hz)	132.0	No
SOLID_TIDES_MODEL	Name of solid tides model (optionally specify order or constituent effects (diurnal, semi-diurnal, etc.))	n/a	diurnal	No
SRP_MODEL	Name of SRP model. This is a free text field, so if the examples on the right are insufficient, others may be used.	n/a	GPS_ROCK BOX_WING CANNONBALL COD ...	No
X_PRECNUT	The X-coordinate of the CIP in the ICRS frame, used to locate the GCRF position and velocity vectors [L9]	arcsec	80.531 880	No
Y_PRECNUT	The Y-coordinate of the CIP in the ICRS frame, used to locate the GCRF position and velocity vectors [L9]	arcsec	7.273 921	No
PERT_STOP	End of the perturbations specification	n/a		Yes

## 6.2.6 OCM DATA: MANEUVER SPECIFICATION

**6.2.6.1** Table 6-6 provides an overview of the OCM maneuver specification section. Only those keywords shown in table 6-6 shall be used in the OCM maneuver specification.

**6.2.6.2** Keyword values shall be provided in the units specified in column three of Table 6-6.

**6.2.6.3** The order of occurrence of these OCM Maneuver Specification keywords shall be fixed as shown in table 6-6, with the exception that comments may be interspersed throughout the this section as required.

**6.2.6.4** The “OCM Data: Maneuver Specification” section is optional; “mandatory” in the context of Table 6-6 denotes those keywords which must be included in this section if this section is included.

**6.2.6.5** One or more OCM Maneuver Specification sections may appear in an OCM.

**6.2.6.6** Maneuver data in the OCM shall be indicated by two keywords: MAN\_START and MAN\_STOP.

**6.2.6.7** The ‘MAN\_TYPE’ keyword must appear before the first line of any maneuver time history data.

**6.2.6.8** Impulsive, thrusting, acceleration and attitude maneuver data in the OCM data shall be time-tagged by a relative time value measured with respect to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.6.9** The thrusting and attitude maneuver specifications include the ability to specify duty cycles based on either a reference direction or reference time. As such, specification of the reference direction or reference time is mandatory in the event that the “Phase angle start,” “Phase angle stop” and “Duty cycle ratio” are provided and invoked. Optionally, “Minimum number of repeats” and “Maximum number of repeats” may be specified.

**6.2.6.10** Within each maneuver specification section, the MAN\_TYPE keyword/value specification is mandatory and shall immediately precede maneuver time history lines.

**6.2.6.11** For impulsive maneuvers (MAN\_TYPE=IMPULSE), each  $\Delta V$  maneuver within the  $\Delta V$  time series shall be specified on a single line that contains nine parameters:

- 1) Time “T\_Relative” in **seconds**
- 2) Velocity increment  $\Delta V_x$  in the selected maneuver reference frame
- 3) Velocity increment  $\Delta V_y$  in the selected maneuver reference frame
- 4) Velocity increment  $\Delta V_z$  in the selected maneuver reference frame
- 5) The maneuver duration (assumed to be centered about the specified maneuver time, in seconds)
- 6) One-sigma percentage error on  $\Delta V$  magnitude

- 7) The **Maneuver Object Number (MON)** that this maneuver is to be applied to (nominally “0” for the primary or host vehicle)
- 8) The mass change (where a NEGATIVE VALUE denotes a mass decrement/loss) associated with a  $\Delta V$  imparted to the host (i.e., MON = 0) or the mass (defined as a POSITIVE VALUE) of the deployed object (if MON  $\neq$  0)

**6.2.6.12** A non-zero MON invokes a parent/child deployment scenario, whereby the parent “host” object (MON=0) deploys one or more child space objects by imparting an impulsive  $\Delta V$  to the deployed object as specified by ( $\Delta V_x$ ,  $\Delta V_y$ ,  $\Delta V_z$  in km/s). The MON shall be a positive number starting at “1” and incrementing through all deployed objects until “N” objects have separated. Where appropriate (e.g. with spring deployment mechanisms) and as directed by the OCM creator, recipients of OCMs using the parent/child deployment capability may need to model/incorporate both the child’s deployment  $\Delta V$  as well as the retrograde  $\Delta V$  imparted to the host (as a ratio of the host and deployed object relative masses such that momentum is conserved).

**6.2.6.13** For thrusting (finite burns, MAN\_TYPE=THRUST), each thrusting maneuver (or, in the case of low-thrust, long-duration burns, each low-thrust interval) within the maneuver series shall be specified on a single line that contains seventeen parameters. The message creator may indicate a change in thrust conditions over which interpolation should not be performed by providing exactly two adjacent lines containing the same time stamp. The seventeen parameters shall be:

- 1) Thruster ID (non-negative integer number)
- 2) Time “T\_Relative” at the start of this thrust interval in **seconds**
- 3) Thrust component Tx measured in the selected maneuver reference frame in **Newtons**
- 4) Thrust component Ty measured in the selected maneuver reference frame in **Newtons**
- 5) Thrust component Tz measured in the selected maneuver reference frame in **Newtons**
- 6) One-sigma percentage error on thrust magnitude
- 7) Maneuver duration in **seconds** (measured with respect to the START of the specified thrust interval)
- 8) Thrust vector Euler axis/angle interpolation mode between current and next thrust line (0=OFF and 1=ON)
- 9) Specific impulse in **seconds**
- 10) Burn efficiency “ $\eta$ ” (e.g. 0.95)
- 11) Additional mass change (where a negative number denotes a mass decrement/loss) associated with this thrust interval (**kg**)
- 12) Burn phase angle start (**deg**)
- 13) Burn phase angle stop (**deg**)
- 14) Minimum number of “ON” cycle repeats
- 15) Maximum number of “ON” cycle repeats
- 16) Duty cycle “ON” duration, initiated at first satisfaction of the burn “on” phase angle range constraint or at completion of a Duty Cycle “OFF” duration (in **seconds**)
- 17) Duty cycle “OFF” duration, initiated at the completion of a burn “ON” phase angle range constraint (in **seconds**)

Note that a maneuver can be specified as a time history sequence of lines, each containing these seventeen parameters. Note also that thrust interval data may be abutting, overlapping, alternating or even intermingled in the maneuver time history block, to accommodate multiple thrusters simultaneously in operation or to accommodate changes in thrust direction, efficiency, mass change, etc.

**6.2.6.14** An acceleration profile (MAN\_TYPE=ACCEL) specification allows aggregate modeling of both maneuvers and additional non-conservative perturbations that are not already specified in the “OCM Perturbations Specification” section above. This allows the OCM originator to model and share such maneuver and perturbations information without the OCM recipient needing to. The message creator may indicate a change in acceleration over which interpolation should not be performed by providing exactly two adjacent lines containing the same time stamp. The acceleration time series shall be specified on a single line that contains eight parameters:

- 1) Time “T\_Relative” at the start of this acceleration interval in **seconds**
- 2) Acceleration component  $A_x$  in the selected maneuver frame in **m/s\*\*2**
- 3) Acceleration component  $A_y$  in the selected maneuver frame in **m/s\*\*2**
- 4) Acceleration component  $A_z$  in the selected maneuver frame in **m/s\*\*2**
- 5) One-sigma percentage error on acceleration magnitude
- 6) Maneuver duration in **seconds** (measured with respect to the START of the specified acceleration interval time)
- 7) Mass change (where a negative number denotes a mass decrement/loss) associated with this acceleration interval
- 8) Acceleration vector Euler axis/angle interpolation mode between current and next acceleration line (0=OFF and 1=ON)

**6.2.6.15** An attitude maneuver (MAN\_TYPE=ATTITUDE) specification allows the OCM originator to model and provide maneuver specifics associated with a propulsion system-based attitude change. The attitude maneuver time series shall be specified on a single line that contains eight parameters:

- 9) Time “T\_Relative” at the start of this acceleration interval in **seconds**
- 10) Torque X-component in the selected maneuver frame in **N\*m**
- 11) Torque Y-component in the selected maneuver frame in **N\*m**
- 12) Torque Z-component in the selected maneuver frame in **N\*m**
- 13) One-sigma percentage error on torque magnitude
- 14) Maneuver duration in **seconds** (measured with respect to the START of the specified attitude maneuver interval)
- 15) Mass change (where a negative number denotes a mass decrement/loss) associated with this attitude maneuver interval
- 16) Acceleration vector Euler axis/angle interpolation mode between current and next acceleration line (0=OFF and 1=ON)
- 17) Burn phase angle start (**deg**)
- 18) Burn phase angle stop (**deg**)
- 19) Minimum number of “ON” cycle repeats
- 20) Maximum number of “ON” cycle repeats

- 21) Duty cycle “ON” duration, initiated at first satisfaction of the burn “on” phase angle range constraint or at completion of a Duty Cycle “OFF” duration (in **seconds**)
- 22) Duty cycle “OFF” duration, initiated at the completion of a burn “ON” phase angle range constraint (in **seconds**)

**6.2.6.16** One or more maneuver time histories may be represented in this section. However, multiple representations shall appear only if each maneuver specification is unique from all other maneuver specifications in at least one of the following respects:

- 1) the data basis (PREDICTED, DETERMINED\_TLM or DETERMINED\_OD);
- 2) the timespans have no overlap.
- 3) The thruster ID is different.



**Table 6-6: OCM Data: Maneuver Specification**

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Keyword	Description	Units	Examples of Values	Mandatory
COMMENT	Comments (allowed at any point(s) throughout the OCM Maneuver Specification section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
MAN_START	Start of a maneuver data interval specification	n/a		Yes
MAN_ID	Optional identification number for this maneuver	n/a	E/W 20160305B	No
MAN_PREV_ID	Optional identification number for the previous maneuver. Note: if the message is not part of a sequence of maneuver messages or if this maneuver is the first in a sequence of maneuvers, then MAN_PREV_ID should be excluded from this message.	n/a	E/W 20160305A	No
MAN_NEXT_ID	Optional identification number for the next maneuver. Note: if the message is not part of a sequence of maneuver messages or if this maneuver is the last in a sequence of maneuvers, then MAN_NEXT_ID should be excluded from this message.	n/a	E/W 20160305C	No
MAN_PURPOSE	The user can specify the intention(s) of the maneuver. Multiple maneuver purposes can be provided as a comma-delimited list. While there is no CCSDS-based restriction on the value for this keyword, it is suggested to use: Aerobraking (AEROBRAKE), Attitude adjust (ATT_ADJUST) Collision avoidance (COLA) Disposal (DISPOSAL) Flyby targeting (FLYBY_TARG) Launch & Early Orbit (LEOP) Maneuver cleanup (MNVR_CLEANUP) Mass adjust (MASS_ADJUST) Momentum desaturation (MOM_DESAT) Orbit adjust (ORBIT_ADJUST) Orbit trim (TRIM) Other (OTHER) Period reduction (PER_RED) Pointing Request Message (PRM_ID_xxxx) Relocation (RELOCATION) Science objective (SCI_OBJ) Spin rate adjust (SPIN_RATE_ADJUST) Station-keeping (SK) Trajectory correction (TRAJ_CORR)	n/a	DISPOSAL	No
MAN_CHAR		n/a	TORQUE_FREE ORBIT_FREE BOTH	No
MAN_WIN_START	Identifies the start of maneuver window that may be different than the maneuver execution start time. This may identify the time at which the satellite is placed into a special maneuver attitude control mode, for example (See 7.5.9 for formatting rules.)	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
MAN_WIN_STOP	Identifies the end of the maneuver window that may be different than the maneuver execution end time. This may identify the end time of any special maneuver attitude control mode, for example (See 7.5.9 for formatting rules.)	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
MAN_BASIS	Basis of this maneuver time history data: PREDICTED or DETERMINED_TLM for telemetry-based reconstruction or DETERMINED_OD for orbit determination-based reconstruction. <b>Omission of this non-mandatory field defaults to PREDICTED.</b>	n/a	DETERMINED_OD PREDICTED	No

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MAN_PRED_SOURCE	For predicted maneuvers, specifies the source of the orbit and/or attitude state(s) upon which the maneuver is based. While there is no CCSDS-based restriction on the value for this free-text keyword, it is suggested to use ORB_ID, OD_ID, ATT_ID, or a combination thereof.	n/a	OD_5	No
MAN_CENTER_NAME	Origin of reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, other defined positional references (e.g. Lagrange points) or another spacecraft (in this case the value for 'MAN_CENTER_NAME' is subject to the same rules as for 'OBJECT_NAME'). There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).  <b>Omission of this non-mandatory field defaults to "EARTH"</b>	n/a	EARTH MOON SOLAR SYSTEM BARYCENTER SUN ISS EROS EARTH_SUN_L2	No
MAN_REF_FRAME	Name of the reference frame in which the maneuver vector direction data is provided, if not intrinsic to the definition of the maneuver data. Use of values other than those in ANNEX B, subsections B2 and B3 must be documented and conveyed in an ICD. The reference frame must be the same for all data elements within a given Maneuver Time History interval.  <b>Omission of this non-mandatory field defaults to TNW.</b>	n/a	EME2000	No

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MAN_FRAME_EPOCH	Epoch of the maneuver reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  <b>Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.</b>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
MAN_TYPE	Specifies type of maneuver being specified. Select impulsive $\Delta V$ (MAN_TYPE = IMPULSE) or finite burn thrust (MAN_TYPE = THRUST) or acceleration profile (MAN_TYPE = ACCEL) time history (see 6.2.6.7, 6.2.6.9 and 6.2.6.10 for details). The maneuver data follows this MAN_TYPE specifier line.	n/a	IMPULSE THRUST ACCEL	Yes
MAN_DUTY_CYCLE_TYPE	Specifies the type of duty cycle type to use for these maneuver time history section: NONE denotes full/continuous thrust; TIME denotes a reference time-based duty cycle; and DIR denotes a reference direction-based duty cycle  <b>Omission of this non-mandatory field defaults to NONE</b>	n/a	NONE TIME DIR	No
DC_REF_TIME	Specifies the duty cycle reference time measured in seconds from EPOCH_TZERO. <b>Omission of this non-mandatory field defaults to zero (i.e., at EPOCH_TZERO)</b>	n/a	8000.0	No
DC_REF_DIR	Specifies the duty cycle reference unit vector direction in the "MAN_REF_FRAME" reference frame	n/a	1.0 0.0 0.0	No
				No
				No
				No
				No
				No
... < Insert maneuver lines here >				Yes
MAN_STOP	End maneuver data interval specification	n/a		Yes

## 6.2.7 OCM DATA: ORBIT STATE TIME HISTORY

**6.2.7.1** Table 6-7 provides an overview of the OCM orbit state time history (“ephemeris”) section. Only those keywords shown in table 6-7 shall be used in OCM orbit state time history data specification.

**6.2.7.2** Keyword values shall be provided in the units specified in column three of Table 6-7.

**6.2.7.3** The order of occurrence of these OCM Orbit State Time History keywords shall be fixed as shown in table 6-7, with the exception that comments may be interspersed throughout the this section as required.

**6.2.7.4** The “OCM Data: Orbit State Time History” section is optional; “mandatory” in the context of Table 6-7 denotes those keywords which must be included in this section if this section is included.

**6.2.7.5** One or more OCM Orbit State Time History sections may appear in an OCM.

**6.2.7.6** Orbit state time history data intervals in the OCM shall be indicated by two keywords: ORB\_START and ORB\_STOP.

**6.2.7.7** The ‘ORB\_TYPE’ keyword must appear before the first line of any orbit state time history metadata or state vector data.

**6.2.7.8** The ORB\_TYPE keyword value shall be selected from Table B4.

**6.2.7.9** One or more orbit state time histories may be represented in this section (spanning ORB\_START to ORB\_STOP). However, multiple representations shall appear only if each orbit state time history is unique from all other orbit state time histories in at least one of the following respects:

- 1) the specified orbit state element sets;
- 2) the data basis (PREDICTED, DETERMINED\_TLM or DETERMINED\_OD);
- 3) the reference frame;
- 4) the orbit center;
- 5) the timespan.

**6.2.7.10** The ORB\_STOP keyword must appear after the last line of orbit state data and metadata. Each of these keywords shall appear on a line by itself.

**6.2.7.11** All orbit state values in the OCM data shall be time-tagged by a relative time value measured with respect to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.7.12** Each orbit state time history shall be time-ordered to be monotonically increasing, with the exception that the message creator may indicate a change in state over which interpolation should not be performed by providing exactly two consecutive lines containing a duplicate timestamp (e.g. following application of an impulsive maneuver or spacecraft or orbit event). In the case of such a duplicate timestamp, interpolation prior to the duplicate timestamp shall use the first of the two duplicate timestamp orbit states, and interpolation after the duplicate timestamp shall use the second of the two.

**6.2.7.13** If the user includes orbit states at key mission event times, it is recommended that those mission event states be annotated as such by a preceding descriptive comment line.

**6.2.7.14** Time tags of consecutive orbit states within the ordered sequence may be separated by uniform or non-uniform step size(s).

**6.2.7.15** Orbit state time tags may or may not match those of maneuver, covariance and/or state transition matrix time histories.

**6.2.7.16** Each set of orbit ephemeris data, including the time tag, must be provided on a single line. The order in which data items are given shall be fixed: **T\_Relative** followed by the orbit state as defined in ANNEX B, subsection B4.

**6.2.7.17** At least one space character must be used to separate the items in each orbit ephemeris data line.

**6.2.7.18** The digits of precision and time steps suitable for interpolation of an orbit ephemeris time history should be chosen according to best practice to avoid positional and interpolation loss of precision [L8].

**Table 6-7: OCM Data: Orbit State Time History**

Keyword	Description	Units	Examples of Values	Mandatory
COMMENT	Comments (allowed at any point(s) throughout the OCM Orbit State Time History section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
ORB_ID	Optional identification number for this orbit state time history block	n/a	ORB_20160402_XYZ	No
ORB_START	Start of an orbit state vector or time history section	n/a	n/a	Yes
ORB_N	Number of elements contained in the element set.  <b>This keyword may be used to override the number of elements implied by the selected ORB_TYPE, in which case ORB_TYPE shall be set to ICD.</b>  <b>Omission of this non-mandatory field defaults to the number of elements implied by ORB_TYPE (Table B4).</b>	n/a	8	No
ORB_BASIS	Basis of this Orbit State time history data: PREDICTED or DETERMINED_TLM for telemetry-based or DETERMINED_OD for orbit determination-based. <b>Omission of this non-mandatory field defaults to PREDICTED.</b>	n/a	PREDICTED	No
ORB_AVERAGING	Free-text keyword specifying whether provided orbit state/elements are either osculating or mean element definitions. The type of averaging (singly- or doubly-averaged, Kozai or Brouwer, etc.) may be specified via ICD. <b>Omission of this non-mandatory field defaults to OSCULATING.</b>	n/a	OSCULATING MEAN_BROWER MEAN_KOZAI (other...)	No
ORB_CENTER_NAME	Origin of reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, other defined positional references (e.g. Lagrange points) or another spacecraft (in this case the value for 'ORB_CENTER_NAME' is subject to the same rules as for 'OBJECT_NAME'). There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).  <b>Omission of this non-mandatory field defaults to "EARTH"</b>	n/a	EARTH MOON SOLAR SYSTEM BARYCENTER SUN ISS EROS EARTH_SUN_L2	No
ORB_REF_FRAME	Name of the reference frame in which the orbit data is provided, if not intrinsic to the definition of the orbit data. Use of values other than those in ANNEX B, subsection B2 must be documented and conveyed in an ICD. The reference frame must be the same for all data elements within a given Orbit State Time History interval.  <b>Where the reference frame is not intrinsic to the selected orbit set, omission of this non-mandatory field defaults to ITRF-97.</b>	n/a	EME2000	No

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ORB_FRAME_EPOCH	Epoch of the orbit data reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  <b>Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.</b>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
ORB_TYPE	Specifies the orbit element set type via “ORB_TYPE = YYY” where YYY is selected from annex B, Table B4.	n/a	n/a	Yes
... < Insert orbit lines here >				Yes
ORB_STOP	End of an orbit state vector or time history section	n/a	n/a	Yes



## **6.2.8 OCM DATA: EPHEMERIS COMPRESSION REPRESENTATION(S)**

**6.2.8.1** Ephemeris Compression (EC) techniques are described in Annex K.

**6.2.8.2** Table 6-8 provides an overview of the OCM ephemeris compression section. Only those keywords shown in table 6-8 shall be used in OCM ephemeris compression data specification.

**6.2.8.3** Keyword values shall be provided in the units specified in column three of Table 6-8.

**6.2.8.4** The order of occurrence of these OCM Ephemeris Compression Representation keywords shall be fixed as shown in table 6-8, with the exception that comments may be interspersed throughout the this section as required.

**6.2.8.5** The “OCM ephemeris compression” section is optional; “mandatory” in the context of Table 6-8 denotes those keywords which must be included in this section if this section is included.

**6.2.8.6** One or more OCM Ephemeris Compression sections may appear in an OCM.

**6.2.8.7** Ephemeris compression sections in the OCM shall be indicated by means of two keywords: EC\_START and EC\_STOP. The EC\_START keyword must appear before the first line of any ephemeris compression metadata or coefficient data. The EC\_STOP keyword must appear after the last line of coefficient data and metadata. Each of these keywords shall appear on a line by itself.

**6.2.8.8** One or more ephemeris compression segments may be represented in this section (spanning EC\_START to EC\_STOP). However, multiple representations shall appear only if they are clearly differentiated from each other by one or more preceding comment(s) or by ICD agreement, and each ephemeris compression segment is unique from all other ephemeris compression segments in at least one of the following respects:

- 1) the specified orbit state element set;
- 2) the ephemeris compression segment is based upon a unique orbit determination or navigation solution;
- 3) the orbit state (when the EC represents different mission events as a function of time from launch window open);
- 4) the reference frame;
- 5) the orbit center;
- 6) the ephemeris compression timespan.

**6.2.8.9** The EC\_STATE\_TYPE keyword value shall be selected from Table B4 or B5 (not including “COV\_NNXNN”).

**6.2.8.10** The OCM EC implementation may also be used to accommodate secular orbit perturbations via polynomials governing time. This means that a seventh set of polynomial coefficients may be supplied to yield either an “adjusted” state event time as a function of time, e.g.,  $t_{event} = EC_{representation}(time\ of\ interest)$ . Such an implementation facilitates the use of more simple baseline orbit propagators for “Hybrid” EC application. For this reason, orbit state types (EC\_STATE\_TYPE) may be also selected from ANNEX B, subsections B4 or B5, where subsection B5 (excepting COV\_NNXNN) allows orbit states to be comprised of time in addition to the standard six-element orbit state specification.

**6.2.8.11** The OCM EC implementation may also be used to specify orbit states at a specific event which vary as a function of time within a specified launch window (as opposed to sequential orbit states as a function of correspondingly-sequential time). As such, each polynomial segment may be used to provide the time and orbit state corresponding to a specific launch event (e.g. Stage 2 ignition). Such an application would therefore typically draw its orbit state types (EC\_STATE\_TYPE) from ANNEX B, subsection B5 (rather than subsection B4), allowing event time to accompany the standard six-element orbit state specification.

**6.2.8.12** The user may choose to use either polynomial and/or series representations to provide orbit state (or orbit state residual) information at key mission events by having EC\_TSTART and EC\_TSTOP set to the same value, whereupon those key mission events should be annotated by a preceding descriptive comment line.

**6.2.8.13** At least one space character must be used to separate the items in each coefficient data line.

**6.2.8.14** The digits of precision suitable for ephemeris representation specification should be chosen according to best practice to avoid EC loss of precision for the recipient’s intended use case.

**Table 6-8: OCM Data: Ephemeris Compression Time History**

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Keyword	Description	Units	Examples of Values	Mandatory
EC_START	Start of a Ephemeris Compression Time History section	n/a	n/a	Yes
COMMENT	Comments (allowed at any point(s) throughout the OCM Ephemeris Compression Time History section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
EC_TSTART	Start time relative to <b>EPOCH_TZERO</b> of this Ephemeris Compression time interval of applicability	n/a	0.0	Yes
EC_BASIS_PROP	Specifies the orbit propagator which is to serve as the basis, upon which the EC representation additively corrects. Specifying EC_BASIS_PROP = NONE indicates that the EC representation is not a hybrid method and the returned functional values obtained from the EC representation correspond directly with desired orbit state information, whereas specification (in free text, including specification of any non-standardized force model or geodetic system implementations and where relevant integrator type i.e. RK 4/5 or RK 8/9; Cowell 9) of a message creator/recipient-shared orbit propagator selects a “Hybrid EC representation” approach in the orbit element definition specified by “EC_STATE_TYPE = YYY” above. Note that this orbit propagator and underlying force model are <u>not</u> required to match the force model specified in the “OCM Force Model” section above to facilitate rapid EC evaluation in field operational use.  <b>Omission of this non-mandatory field defaults to NONE.</b>	n/a	SGP4	No
EC_ORB_STATE	Specifies the orbit state elements for employment of a “Hybrid EC representation” approach in the orbit element definition specified by “EC_STATE_TYPE = YYY” above. The number of state vector rows <u>following</u> the EC_ORB_STATE keyword containing the initial state correspond to the “EC_BASIS_PROP” required state vector format (shared either by ICD agreement or common industry understanding). Note the specification of the initial condition state vector epoch may be included in this common understanding. <b>This keyword and state vector data are mandatory if and only if EC_BASIS_PROP is not set to “NONE”</b>	n/a	6700.0 0.0 0.0 0.0 0.0 0.839099633	No
EC_REPRESENT	Specifies the type of EC representation used in the coefficients which immediately follow. Valid options are: CHEBYSHEV or FOURIER (normative values, or others are permitted as a free text field). Specific implementation details of the basis functions and algorithms used shall be clarified by accompanying ICD where necessary.  <b>Where the EC representation is not specified, omission of this non-mandatory field defaults to CHEBYSHEV.</b>	n/a	CHEBYSHEV	No
EC_STATE_TYPE	Indicates EC representation via “EC_STATE_TYPE = YYY” where YYY is selected from ANNEX B, subsection B4 or B5 (not including “COV_NNXNN”). <b>Omission of this non-mandatory field defaults to EQUIN.</b>	n/a	EQUIN	No

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

Keyword	Description	Units	Examples of Values	Mandatory
EC_CENTER_NAME	<p>Origin of reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, other defined positional references (e.g. Lagrange points) or another spacecraft (in this case the value for 'EC_CENTER_NAME' is subject to the same rules as for 'OBJECT_NAME'). There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).</p> <p><b>Omission of this non-mandatory field defaults to "EARTH"</b></p>	n/a	EARTH MOON SOLAR SYSTEM BARYCENTER SUN ISS EROS EARTH_SUN_L2	No
EC_REF_FRAME	<p>Name of the reference frame in which the Ephemeris Compression data is computed, if not intrinsic to the EC orbit element definition. Use of values other than those in ANNEX B, subsections B2 must be documented and conveyed via an ICD. The reference frame must be the same for all data elements within a given Ephemeris Compression Time History interval.</p> <p><b>Where the reference frame is not intrinsic to the selected EC set, omission of this non-mandatory field defaults to EME2000.</b></p>	n/a	EME2000	No
EC_FRAME_EPOCH	<p>Epoch of the Ephemeris Compression reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)</p> <p><b>Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.</b></p>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
EC_REPR_N	<p>Number of terms (coefficients) in the selected EC representation for this segment. Coefficients shall be supplied in columnar fashion, with the "ith" row corresponding to the "jth" column orbital element.</p> <p>In the FOURIER representation, the cosine coefficients are supplied for each orbit element, followed (on the same line) by the sine coefficients for each element.</p> <p>As such, EC_REPR_N always denotes the number of coefficient data rows to follow prior to the next EC_REPRESENT or EC_TSTOP specifier.</p>	n/a	10	Yes
EC_TSTOP	End time relative to <b>EPOCH_TZERO</b> of this Ephemeris Compression time interval of applicability	n/a	86400.0	Yes
... < Insert EC data here >				Yes
EC_STOP	End of a Ephemeris Compression section	n/a	n/a	Yes

## **6.2.9 OCM DATA: ORBIT DETERMINATION DATA**

**6.2.9.1** Table 6-9 provides an overview of the OCM orbit determination data section. Only those keywords shown in table 6-9 shall be used in OCM orbit determination data specification.

**6.2.9.2** Keyword values shall be provided in the units specified in column three of Table 6-9.

**6.2.9.3** The order of occurrence of these OCM Orbit Determination Data keywords shall be fixed as shown in table 6-9, with the exception that comments may be interspersed throughout the this section as required.

**6.2.9.4** The “OCM Data: Orbit Determination Data” section is optional; “mandatory” in the context of Table 6-9 denotes those keywords which must be included in this section if this section is included.

**6.2.9.5** Only one Orbit Determination Data section shall appear in any OCM.

**6.2.9.6** Orbit determination data in the OCM shall be indicated by two keywords: OD\_START and OD\_STOP.

**6.2.9.7** All orbit determination event times shall be specified in DAYS relative to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.9.8** Definition: an 'observation' is a unique measurement set of a satellite's state from a single sensor configuration at a single time (e.g. azimuth from a single sensor at a single time).

**6.2.9.9** Definition: a 'sensor track' is a set of at least three observations for the same object, observed by the same sensor configuration, where each observation is within a specified number of minutes (which is dependent on the orbit regime of the object) of the other observations in the track (e.g. a set of 10 two-way transponder range measurements from the same sensor using the same transponder on the satellite).

**Table 6-9: OCM Data: Orbit Determination Data**

Keyword	Description	Units	Examples of Values	Mandatory
COMMENT	Comments (allowed at any point(s) throughout the OCM Orbit Determination Data section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
OD_ID	Optional identification number for this orbit determination	n/a	OD_20160402	No
OD_PREV_ID	Optional identification number for the previous orbit determination. Note: if this orbit determination is the first one, then OD_PREV_ID should be excluded from this message.	n/a	OD_20160401	No
OD_START	Start of an orbit determination data section	n/a	n/a	Yes
OD_METHOD	Type of orbit determination method used to produce the orbit estimate. Commonly used methods include Batch Weighted Least Squares (BWLS), the Extended Kalman Filter (EKF).	n/a	BWLS, EKF	Yes
OD_EPOCH	Epoch of the orbit determination solved-for state (See 7.5.9 for formatting rules.)  <b>Where the orbit determination epoch is not supplied, omission of this non-mandatory field defaults to ZERO (i.e. occurs at EPOCH_TZERO).</b>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
DAYS_SINCE_FIRST_OBS	Days elapsed between first accepted observation and OD_EPOCH	d	3.5	No
DAYS_SINCE_LAST_OBS	Days elapsed between last accepted observation and OD_EPOCH	d	1.2	No
RECOMMENDED_OD_SPAN	Number of days of observations recommended for the OD of the object ( <i>useful only for Batch OD systems</i> )	d	5.2	No
ACTUAL_OD_SPAN	Actual time span used for the OD of the object (NOTE: should equal (DAYS_SINCE_FIRST_OBS - DAYS_SINCE_LAST_OBS))	d	2.3	No
OBS_AVAILABLE	The number of observations available within the actual OD time span	n/a	100	No
OBS_USED	The number of observations accepted within the actual OD time span	n/a	90	No
TRACKS_AVAILABLE	The number of sensor tracks, for the actual time span, that were available for the OD	n/a	33	No
TRACKS_USED	The number of sensor tracks, for the actual time span, that were accepted for the OD	n/a	30	No
MAXIMUM_OBS_GAP	The maximum time between observations in the OD of the object	d	1.0	No
OD_EPOCH_EIGMAJ	Positional error ellipsoid $1\sigma$ major eigenvalue at the epoch of the OD	km	.05873	No
OD_EPOCH_EIGMED	Positional error ellipsoid $1\sigma$ medium eigenvalue at the epoch of the OD	km	.0357	No
OD_EPOCH_EIGMIN	Positional error ellipsoid $1\sigma$ minor eigenvalue at the epoch of the OD	km	.0215	No
OD_CONFIDENCE	OD confidence metric, which by definition spans 0 to 100%. (useful only for Filter-based OD systems). The OD Confidence metric shall be defined by ICD.	Percent	95.3	No

WEIGHTED_RMS	<p><i>(Useful / valid only for Batch OD systems).</i></p> <p>The weighted RMS residual ratio, defined as:</p> $\text{Weighted RMS} = \sqrt{\frac{\sum_{i=1}^N w_i (y_i - \hat{y}_i)^2}{N}}$ <p>Where <math>y_i</math> is the observation measurement at the <math>i</math>th time</p> <p><math>\hat{y}_i</math> is the current estimate of <math>y_i</math>,</p> <p><math>w_i = \frac{1}{\sigma_i^2}</math> is the weight (sigma) associated with the measurement at the <math>i</math>th time and <math>N</math> is the number of observations.</p> <p>This is a value that can generally identify the quality of the most recent vector update, and is used by the analyst in evaluating the OD process. A value of 1.00 is ideal.</p>	(measurement units)	1.3	No
TRK_MESSAGE_IDS	A free-text list of the file name(s) and/or associated identification number(s) of tracking Data Message (TDM) observations upon which this OD is based.	n/a	TDM_0005.txt	No
DATA_TYPES	Comma-separated list of observation data types utilized in this orbit determination. Although this is a free-text field, it is recommended to use data type descriptor(s) as provided in Table 3-5 of the Tracking Data Message standard, with the exception of the DATA_START, DATA_STOP, and COMMENT keywords.	n/a	n/a	No
OD_STOP	End of an orbit determination data section	n/a	n/a	Yes



## 6.2.10 OCM DATA: ORBIT STATE COVARIANCE TIME HISTORY

**6.2.10.1** Table 6-10 provides an overview of the OCM covariance time history section. Only those keywords shown in table 6-9 shall be used in OCM covariance time history data specification.

**6.2.10.2** Keyword values shall be provided in the units specified in column three of Table 6-10.

**6.2.10.3** The order of occurrence of these OCM Orbit State Covariance Time History keywords shall be fixed as shown in table 6-10, with the exception that comments may be interspersed throughout the this section as required.

**6.2.10.4** The “OCM Data: Orbit State Covariance Time History” section is optional; “mandatory” in the context of Table 6-10 denotes those keywords which must be included in this section if this section is included.

**6.2.10.5** One or more OCM covariance time history sections may appear in an OCM.

**6.2.10.6** Covariance time history data intervals in the OCM shall be indicated by means of two keywords: COV\_START and COV\_STOP. The COV\_START keyword must appear before the first line of any covariance metadata or matrix data. The COV\_STOP keyword must appear after the last line of covariance data and metadata. Each of these keywords shall appear on a line by itself.

**6.2.10.7** One or more covariance time histories may be represented in this section (spanning COV\_START to COV\_STOP). However, multiple representations shall appear only if they are clearly differentiated from each other by one or more precluding comment(s) or by ICD agreement, and each covariance time history is unique from all other covariance time histories in at least one of the following respects:

- 1) the covariance’s specified orbit state element sets;
- 2) the covariance data basis (e.g. PREDICTED, ACTUAL, etc.);
- 3) the reference frame;
- 4) the covariance timespan.

**6.2.10.8** The COV\_TYPE keyword value shall be selected from Table B4 or B5.

**6.2.10.9** All covariance matrices in the OCM data shall be time-tagged by a relative time value measured with respect to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.10.10** Each covariance time history shall be time-ordered to be monotonically increasing, with the exception that the message creator may indicate a change in state over which interpolation should not be performed by providing exactly two consecutive covariance data blocks containing a duplicate timestamp (e.g. following application of an impulsive maneuver

or spacecraft or orbit event). In the case of such a duplicate timestamp, interpolation prior to the duplicate timestamp shall use the first of the two duplicate timestamp covariance matrices, and interpolation after the duplicate timestamp shall use the second of the two.

**6.2.10.11**If the user includes covariances at key mission event times, it is recommended that those mission event covariances be annotated as such by a preceding descriptive comment line.

**6.2.10.12**Within each covariance time history, one or more covariance matrices may appear at any desired frequency (for example, multiple covariances when based upon Monte Carlo simulations spanning multiple events or when propagated to multiple time points).

**6.2.10.13**Time tags of consecutive covariance information within the ordered sequence may be separated by uniform or non-uniform step size(s).

**6.2.10.14**Covariance time tags may or may not match those of maneuver, orbit state and/or state transition matrix time histories.

NOTE: Interpolation of covariance matrices at neighboring relative time points shall be done by (1) eigenvalue/vector decomposition; (2) linear (or higher-order) interpolation of neighboring eigenvalues; (3) Euler axis/angle rotation of eigenvectors at intermediate time(s) of interest; and (4) Re-composition of attained eigenvalues and eigenvectors into covariances at time(s) of interest [L16]. Direct interpolation of covariance matrix components can produce invalid (non-positive-semidefinite) covariances.

**6.2.10.15**The time of the event associated with provided covariance matrices must be provided via the “T = ” keyword. The reference frame of the covariance matrix, if different from that of the states in the ephemeris, must be provided via the ‘COV\_REF\_FRAME’ keyword.

**6.2.10.16**Values in the covariance matrix shall be expressed in the applicable reference frame and shall be presented sequentially from upper left [1,1] to lower right in lower triangular form, row-by-row from left to right. Variance and covariance values shall be expressed in standard double precision as related in 7.5.

**6.2.10.17**Each row of the lower triangular covariance matrix must be provided on a single line. The order in which data items are given shall be fixed. The elements in each row of covariates shall be defined by the COV\_TYPE keyword specification (note that only a single line shall be provided for COV\_TYPE = TEIGVAL3EIGVEC3).

**6.2.10.18**If COV\_TYPE ≠ TEIGVAL3EIGVEC3, then each of the “N” rows of the covariance matrix shall contain from one to “N” numbers depending on what row of the matrix is being represented (first row has one element, second row has two, continuing in this fashion until the “Nth” row has “N” elements).

**6.2.10.19**At least one space character must be used to separate the items in each covariance matrix data line.

**6.2.10.20**The digits of precision and time steps suitable for interpolation of a covariance time history should be chosen according to best practice to avoid covariance and interpolation loss of precision [L8].

**Table 6-10: OCM Data: Covariance Time History**

Keyword	Description	Units	Examples of Values	Mandatory
COV_START	Start of a covariance time history section	n/a	n/a	Yes
COMMENT	Comments (allowed at any point(s) throughout the OCM Covariance Time History section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
COV_BASIS	Basis of this covariance time history data:  PREDICTED or DETERMINED_EMPIRICAL (for empirically-derived such as overlap analyses) or DETERMINED_OD for orbit determination-based.. <b>Omission of this non-mandatory field defaults to PREDICTED.</b>	n/a	PREDICTED	No
COV_REF_FRAME	Name of the reference frame in which the covariance data is provided, if not intrinsic to the definition of the covariance data. Use of values other than those in ANNEX B, subsections B2 and B3 must be documented and conveyed via an ICD. The reference frame must be the same for all data elements within a given Covariance Time History interval.  <b>Where the reference frame is not intrinsic to the selected covariance set, omission of this non-mandatory field defaults to TNW.</b>	n/a	EME2000	No
COV_FRAME_EPOCH	Epoch of the covariance data reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  <b>Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.</b>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
COV_NNXNN	Number of diagonal elements contained in the full covariance.  <b>This keyword may be used to override the number of elements implied by the selected COV_TYPE. Omission of this non-mandatory field defaults to the number of elements implied by COV_TYPE (ANNEX B, subsections B2 and B3).</b>	n/a	10	No
COV_TYPE	Indicates covariance composition via “COV_TYPE = YYY” where YYY is selected from ANNEX B, subsections B4 and B5).	n/a	n/a	Yes
T	Time relative to EPOCH_TZERO. <b>Where the time is not provided, omission of this non-mandatory field defaults to 0.0.</b>	s	10	No
...< Insert covariance data here>				Yes
COV_STOP	End of a covariance time history section	n/a	n/a	Yes

## 6.2.11 OCM DATA: STATE TRANSITION MATRIX TIME HISTORY

**6.2.11.1** State Transition Matrices (STMs) can be very useful in mapping both an initial state, and (separately) differences about that state, to other time(s) of interest. Following the terminology and definitions of reference [L9], pp. 82, 778-780 and 809) allows the analyst to map states, or alternatively state differences, at time  $t_0$  to another time  $t_i$ . As noted in reference [L9], these are distinctly different in definition and content from each other. Both types of State Transition Matrices are supported, as specified by the STM\_MAP\_MODE keyword.

**6.2.11.2** Table 6-11 provides an overview of the OCM state transition matrix time history section. Only those keywords shown in table 6-10 shall be used in OCM state transition matrix time history data specification.

**6.2.11.3** Keyword values shall be provided in the units specified in column three of Table 6-11

**6.2.11.4** The order of occurrence of these OCM State Transition Matrix Time History keywords shall be fixed as shown in table 6-11, with the exception that comments may be interspersed throughout the this section as required.

**6.2.11.5** The “OCM Data: State Transition Matrix Time History” section is optional; “mandatory” in the context of Table 6-11 denotes those keywords which must be included in this section if this section is included.

**6.2.11.6** One or more OCM State Transition Matrix Time History sections may appear in an OCM.

**6.2.11.7** State transition matrix time history data intervals in the OCM shall be indicated by means of two keywords: STM\_START and STM\_STOP. The STM\_START keyword must appear before the first line of any state transition matrix metadata or matrix data. The STM\_STOP keyword must appear after the last line of state transition matrix data and metadata. Each of these keywords shall appear on a line by itself.

**6.2.11.8** One or more state transition matrix time histories may be represented in this section (spanning STM\_START to STM\_STOP). However, multiple representations shall appear only if they are clearly differentiated from each other by one or more precluding comment(s) or by ICD agreement, and each state transition matrix time history is unique from all other state transition matrix time histories in at least one of the following respects:

- 7) the specified orbit state element set;
- 8) the state transition matrix time history is based upon a unique orbit determination or navigation solution;
- 9) the reference frame;
- 10) the orbit center;

11) the state transition matrix timespan.

**6.2.11.9** The STM\_TYPE keyword value shall be selected from Table B4.

**6.2.11.10** All state transition matrices in the OCM data shall be time-tagged by a relative time value measured with respect to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.11.11** Each state transition matrix time history shall be time-ordered to be monotonically increasing, with the exception that the message creator may indicate a change in vehicle state by providing exactly two consecutive state transition matrix data blocks containing a duplicate timestamp (e.g. following application of an impulsive maneuver or spacecraft or orbit event).

**6.2.11.12** No interpolation of the state transition matrix time history shall be undertaken, since the state transition matrix pre- and post-multiplies the state (or covariance) in the mapping process to yield states and covariances that may then be properly interpolated.

**6.2.11.13** If the user includes state transition matrices at key mission event times, it is recommended that those mission event state transition matrices be annotated as such by a preceding descriptive comment line.

**6.2.11.14** Time tags of consecutive state transition matrices within the ordered sequence may be separated by uniform or non-uniform step size(s).

**6.2.11.15** State transition matrix time tags may or may not match those of maneuver, orbit state and/or covariance time histories.

**6.2.11.16** The time of the event associated with provided state transition matrices must be provided via the "T = " keyword. The reference frame of the state transition matrices, if different from that of the states in the ephemeris, must be provided via the 'STM\_FRAME' keyword.

**6.2.11.17** Values in each state transition matrix shall be expressed in the applicable reference frame and shall be presented sequentially from upper to lower and row-by-row from left to right. State transition matrix values shall be expressed in standard double precision as discussed in 7.5.

**6.2.11.18** Each row of each state transition matrix must be provided on a single line. The order in which data items are given shall be fixed. The elements in each row shall be defined by the STM\_TYPE keyword specification. The "N" rows of the state transition matrix shall each contain "N" numbers.

**6.2.11.19** At least one space character must be used to separate the items in each state transition matrix data line.

**6.2.11.20** The digits of precision and time steps suitable for state transition matrix time history should be chosen according to best practice to avoid STM propagation loss of precision.

**Table 6-11: OCM Data: State Transition Matrix Time History**

<b>Keyword</b>	<b>Description</b>	<b>Units</b>	<b>Examples of Values</b>	<b>Mandatory</b>
STM_START	Start of a state transition matrix time history section	n/a	n/a	Yes
COMMENT	Comments (allowed at any point(s) throughout the OCM State Transition Matrix Time History section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
STM_MAP_MODE	Indicates whether state transition matrix maps: <ul style="list-style-type: none"> <li>- An initial state to later states (STATE) or</li> <li>- Initial state differences to later differences (DIFFERENCES)</li> </ul>	n/a	STATE DIFFERENCES	Yes
STM_REF_TIME	Epoch time of the initial state or initial state differences relative to EPOCH_TZERO, to which the state transition matrix is referenced and at which time the STM $\equiv$ the identity matrix	n/a	0.0	Yes
STM_CENTER_NAME	Origin of reference frame, which may be a natural solar system body (planets, asteroids, comets, and natural satellites), including any planet barycenter or the solar system barycenter, other defined positional references (e.g. Lagrange points) or another spacecraft (in this case the value for 'STM_CENTER_NAME' is subject to the same rules as for 'OBJECT_NAME'). There is no CCSDS-based restriction on the value for this keyword, but for natural bodies it is recommended to use names from the NASA/JPL Solar System Dynamics Group at <a href="http://ssd.jpl.nasa.gov">http://ssd.jpl.nasa.gov</a> (reference [5]).  <b>Omission of this non-mandatory field defaults to "EARTH"</b>	n/a	EARTH MOON SOLAR SYSTEM BARYCENTER SUN ISS EROS EARTH_SUN_L2	No
STM_REF_FRAME	Name of the reference frame in which the state transition matrix data is computed, if not intrinsic to the definition of the state transition matrix data. Use of values other than those in ANNEX B, subsections B2 and B3 must be documented and conveyed via an ICD. The reference frame must be the same for all data elements within a given State Transition Matrix Time History interval.  <b>Where the reference frame is not intrinsic to the selected STM set, omission of this non-mandatory field defaults to ICRF2000.</b>	n/a	EME2000	No

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Keyword	Description	Units	Examples of Values	Mandatory
STM_FRAME_EPOCH	Epoch of the state transition matrix data reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  <b>Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.</b>	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
STM_N	Dimensionality of the “N x N” state transition matrix.  <b>This keyword may be used to override the number of elements implied by the selected STM_TYPE. Omission of this non-mandatory field defaults to the number of elements implied by STM_TYPE (Table B4).</b>	n/a	6	No
STM_TYPE	Indicates state transition matrix composition via “STM_TYPE = YYY” where YYY is selected from ANNEX B, subsection B4 or B5. <b>Omission of this non-mandatory field defaults to CARTPV.</b>	n/a	n/a	Yes
T	Time relative to EPOCH_TZERO. <b>Where the time is not provided, omission of this non-mandatory field defaults to 0.0.</b>	s	10	No
... < Insert STM data here >				Yes
STM_STOP	End of a state transition matrix time history section	n/a	n/a	Yes



## 6.2.1 OCM DATA: ATTITUDE TIME HISTORY

**6.2.1.1** Table 6-12 provides an overview of the OCM Attitude Time History section. Only those keywords shown in table 6-12 shall be used in OCM Attitude Time History data specification.

**6.2.1.2** Keyword values shall be provided in the units specified in column three of Table 6-12.

**6.2.1.3** The order of occurrence of these OCM Attitude Time History keywords shall be fixed as shown in table 6-12, with the exception that comments may be interspersed throughout the this section as required.

**6.2.1.4** The “OCM Data: Attitude Time History” section is optional; “mandatory” in the context of Table 6-12 denotes those keywords which must be included in this section if this section is included.

**6.2.1.5** One or more OCM Maneuver Specification sections may appear in an OCM.

**6.2.1.6** Attitude Time History data intervals in the OCM shall be indicated by means of two keywords: ATT\_START and ATT\_STOP. The ATT\_START keyword must appear before the first line of any attitude metadata or data. The ATT\_STOP keyword must appear after the last line of attitude data and metadata. Each of these keywords shall appear on a line by itself.

**6.2.1.7** One or more attitude time histories may be represented in this section (spanning ATT\_START to ATT\_STOP). However, multiple representations shall appear only if they are clearly differentiated from each other by one or more precluding comment(s) or by ICD agreement, and each attitude time history is unique from all other attitude time histories in at least one of the following respects:

- 1) the attitude time history’s specified representation;
- 2) the attitude type (e.g., quaternion or Euler angle)
- 3) the attitude data basis (e.g. PREDICTED, ACTUAL, etc.);
- 4) the reference frame;
- 5) the attitude timespan.

**6.2.1.8** This attitude time history shall map the transformation from one reference frame to another (i.e., either transforms coordinates in reference frame “A” into coordinates in frame “B,” or frame “B” to “A”), with the quaternion-based frame transformation (matrix “M”) of the mapping  $X = M * X_0$  defined as:

$$M = \begin{bmatrix} q_1^2 - q_2^2 - q_3^2 - q_c^2 & 2(q_1q_2 - q_cq_3) & 2(q_1q_3 - q_cq_2) \\ 2(q_1q_2 - q_cq_3) & -q_1^2 + q_2^2 - q_3^2 + q_c^2 & 2(q_2q_3 - q_cq_1) \\ 2(q_1q_3 + q_cq_2) & 2(q_2q_3 - q_cq_1) & -q_1^2 - q_2^2 + q_3^2 + q_c^2 \end{bmatrix}$$

And the Euler angle rotations defined via a sequence of three direction cosine matrices, e.g. for an “XYZ” rotational transformation,

$$M = \begin{bmatrix} \cos(\theta_3) & \sin(\theta_3) & 0 \\ -\sin(\theta_3) & \cos(\theta_3) & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} \cos(\theta_2) & 0 & -\sin(\theta_2) \\ 0 & 1 & 0 \\ \sin(\theta_2) & 0 & \cos(\theta_2) \end{bmatrix} \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos(\theta_1) & \sin(\theta_1) \\ 0 & -\sin(\theta_1) & \cos(\theta_1) \end{bmatrix}$$

**6.2.1.9** All attitude states in the OCM data shall be time-tagged by a relative time value measured with respect to the epoch time specified via the EPOCH\_TZERO keyword.

**6.2.1.10** The attitude state itself may be specified (using the “ATT\_TYPE” keyword) as either being a quaternion (w/associated rates-of-change) or as an Euler rotation sequence (w/associated rates-of-change).

**6.2.1.11** For ATT\_TYPE=QUATERNION, each attitude state at a single time shall be provided on a single line (with  $\phi$  denoting the rotation angle) consisting of:

- Time (seconds from EPOCH\_TZERO)
- Q1 =  $e_1 * \sin(\phi/2)$
- Q2 =  $e_2 * \sin(\phi/2)$
- Q3 =  $e_3 * \sin(\phi/2)$
- QC =  $\cos(\phi/2)$
- Q1\_DOT [1/s]
- Q2\_DOT [1/s]
- Q3\_DOT [1/s]
- Q4\_DOT [1/s]

**6.2.1.12** For ATT\_TYPE=EULER, each attitude state at a single time shall be provided on a single line (with  $\phi$  denoting the rotation angle) consisting of:

- Time (seconds from EPOCH\_TZERO)
- EULER\_ROT\_SEQ, the three-letter Euler rotation sequence moniker, with each letter selected from one of (X=X-axis, Y = Y-axis, or Z = Z-axis)
- ANGLE\_1, Angle of the first rotation in the sequence (= leftmost letter)
- ANGLE\_2, Angle of the second rotation in the sequence (= middle letter)
- ANGLE\_3, Angle of the third rotation in the sequence (= rightmost letter)
- ANGLE\_1\_DOT, the time derivative of ANGLE\_1 [deg/s]
- ANGLE\_2\_DOT, the time derivative of ANGLE\_2 [deg/s]
- ANGLE\_3\_DOT, the time derivative of ANGLE\_3 [deg/s]

**6.2.1.13** Each attitude time history shall be time-ordered to be monotonically increasing, with the exception that the message creator may indicate a change in attitude state over which

interpolation should not be performed by providing exactly two consecutive attitude data blocks containing a duplicate timestamp (e.g. following application of an impulsive maneuver or spacecraft or orbit event). In the case of such a duplicate timestamp, interpolation prior to the duplicate timestamp shall use the first of the two duplicate timestamp attitude orientations, and interpolation after the duplicate timestamp shall use the second of the two.

**6.2.1.14** If the user includes attitudes at key mission event times, it is recommended that those mission event attitudes be annotated as such by a preceding descriptive comment line.

**6.2.1.15** Within each attitude time history, one or more attitudes may appear at any desired frequency (for example, multiple attitudes when propagated to multiple time points).

**6.2.1.16** Time tags of consecutive attitude information within the ordered sequence may be separated by uniform or non-uniform step size(s).

**6.2.1.17** Attitude time tags may or may not match those of maneuver, orbit state and/or state transition matrix time histories.

NOTE: Interpolation of attitude orientations shall be done by an Euler axis/angle rotation methodology [L16]. Direct interpolation of a series of attitude data can produce invalid intermediate attitudes.

**Table 6-12: OCM Data: Attitude Time History**

Keyword	Description	Units	Examples of Values	Mandatory
ATT_START	Start of the Spacecraft Attitude Time History section	n/a	n/a	Yes
ATT_ID	Optional identification number for this attitude state time history block	n/a	ORB_20160402_XY Z	No
COMMENT	Comments (allowed at any point(s) throughout the OCM Attitude Time History section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
ATT_TYPE	Indicates attitude representation via “ATT_TYPE = YYY” where YYY is either “QUATERNION” (requiring Q1, Q2, Q3, QC and their associated rates-of-change to be provided) or “EULER” (requiring EULER_ROT_SEQ, ANGLE_1, ANGLE_2, ANGLE_3 and their associated rates-of-change to be provided). Omission of this non-mandatory field defaults to QUATERNION.	n/a	EULER	No
ATT_REF_FRAME_A	Name of the reference frame to which the attitude data are referenced. Use of values other than those in ANNEX B, subsections B2 must be documented and conveyed via an ICD. The reference frame must be the same for all data elements within a given Attitude Time History interval.  <b>Omission of this non-mandatory field defaults to EME2000.</b>	n/a	EME2000	No

Keyword	Description	Units	Examples of Values	Mandatory
ATT_FRAME_A_EPOCH	Epoch of the Attitude “A” reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
ATT_REF_FRAME_B	Name of the reference frame to which the attitude data are referenced. Use of values other than those in ANNEX B, subsections B2 or must be documented and conveyed via an ICD. The reference frame must be the same for all data elements within a given Attitude Time History interval.  <b>Omission of this non-mandatory field defaults to SC_BODY_1.</b>	n/a	SC_BODY_1	Yes
ATT_FRAME_B_EPOCH	Epoch of the Attitude “B” reference frame, if not intrinsic to the definition of the reference frame. (See 7.5.9 for formatting rules.)  Where the reference frame epoch is required and not intrinsic to the selected reference frame, omission of this non-mandatory field defaults to EPOCH_TZERO.	n/a	2001-11-06T11:17:33 2002-204T15:56:23Z	No
ATT_DIR	Specification of direction of the attitude mapping (i.e. either from frame “A” to “B”, or frame “B” to frame “A”). <b>Omission of this non-mandatory field defaults to A2B.</b>	n/a	A2B B2A	No
SPIN_ALPHA	Right ascension of spin axis vector, measured in “ATT_REF_FRAME” reference frame	deg	270.0	No
SPIN_DELTA	Declination of the spin axis vector, measured in “ATT_REF_FRAME” reference frame	deg	80.0	No
SPIN_ANGLE	Phase of the satellite about the spin axis	deg	230.0	No
SPIN_ANGLE_VEL	Angular velocity of satellite around spin axis	deg/s	0.03	No
PRECESSION_ANGLE	Nutation angle of spin axis	deg	1.2	No
PRECESSION_PERIOD	Precession period of the spin axis	s	1000.0	No
PRECESSION_PHASE	precession phase	deg	20.0	No
ANGVEL_X	X component of the angular velocity vector of the object frame measured with respect to the “ATT_REF_FRAME” reference frame	deg/s	2.0	No
ANGVEL_Y	Y component of the angular velocity vector of the object frame measured with respect to the “ATT_REF_FRAME” reference frame	deg/s	3.0	No
ANGVEL_Z	Z component of the angular velocity vector of the object frame measured with respect to the “ATT_REF_FRAME” reference frame	deg/s	0.0	No
... < Insert attitude lines here >				No
ATT_STOP	End of an Attitude Time History section	n/a	n/a	Yes

## 6.2.2 OCM DATA: USER-DEFINED PARAMETERS

**6.2.2.1** A section of User Defined Parameters may be provided if necessary. In principle, this provides flexibility, but also introduces complexity, non-standardization, potential ambiguity, and potential processing errors. Accordingly, if used, the keywords and their

meanings must be described in an ICD. User Defined Parameters, if included in an OCM, should be used as sparingly as possible; their use is not encouraged.

**6.2.2.2** The “OCM Data: User-Defined Parameters” section is optional; “mandatory” in the context of Table 6-12 denotes those keywords which must be included in this section if this section is included.

**6.2.2.3** Table 6-12 provides an overview of the OCM user-defined data section. Only those keywords shown in Table 6-12 shall be used in OCM user-defined data specification.

**Table 6-12: OCM Data: User-Defined Parameters**

<b>Keyword</b>	<b>Description</b>	<b>Units</b>	<b>Examples of Values</b>	<b>Mandatory</b>
COMMENT	Comments (allowed at any point(s) throughout the OCM User-Defined Data section). (See 7.7 for formatting rules.)	n/a	COMMENT This is a comment	No
USER_DEFINED_x	User defined parameter, where ‘x’ is replaced by a variable length user specified character string. Any number of user defined parameters may be included, if necessary to provide essential information that cannot be conveyed in COMMENT statements.	n/a	USER_DEFINED_EARTH_MODEL = WGS-84	No

### 6.3 OCM EXAMPLES

Figure 6-1 through figure 6-4 are examples of Orbit Comprehensive Messages. The first has only a time history of orbital states and constitutes a minimal content OCM. The second includes space object characteristics and perturbations specifications; the third includes a time series of maneuvers, a time history of Cartesian position and velocity orbit states, followed by a time history of Keplerian elements; and the fourth includes a time series of covariance matrices.

```

CCSDS_OCM_VERS = 3.0
CREATION_DATE  = 1998-11-06T09:23:57
ORIGINATOR     = JAXA
OBJECT_NAME    = GODZILLA 5
EPOCH_TZERO   = 1998-12-18T14:28:15.1172

ORB_START
ORB_REF_FRAME  = EME2000
ORB_TYPE       = CARTPV
  0.0 2789.6 -280.0 -1746.8 4.73 -2.50 -1.04
 10.0 2783.4 -308.1 -1877.1 5.19 -2.42 -2.00
 20.0 2776.0 -336.9 -2008.7 5.64 -2.34 -1.95
    < intervening data records omitted here >
 500.0 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535
ORB_STOP

```

**Figure 6-1: Simple/Succinct OCM File example with only Cartesian ephemeris**

**NEED TO ADD OD EXAMPLE**

```

CCSDS_OCM_VERS = 3.0

CREATION_DATE      = 1998-11-06T09:23:57
ORIGINATOR         = JAXA
TECH_POC           = Mr. Rodgers, (719)555-5555, email@email.XXX
EPOCH_TZERO        = 1998-12-18T14:28:15.1172

OBJECT_NAME        = GODZILLA 5
OBJECT_ID          = 1998-999ZZZ
TIME_SYSTEM        = UT1
TAIMUTC_TZERO     = 36 [s]
UT1MUTC_RATE_TZERO = 0.0001 [ms/day]
UT2MUTC_TZERO     = .357 [s]

COMMENT S/C Physical Characteristics:

PHYS_START
MASS                = 100.0 [kg]
OEB_MAX             = 2.0 [m]
OEB_MED             = 1.0 [m]
OEB_MIN             = 0.5 [m]
OEB_YAW             = 30.0 [deg]
OEB_PITCH           = 1.7 [deg]
OEB_ROLL            = -10.0 [deg]
AREA_ALONG_OEB_MAX = 0.15 [m**2]
AREA_ALONG_OEB_MED = 0.3 [m**2]
AREA_ALONG_OEB_MIN = 0.5 [m**2]
PHYS_STOP

COMMENT Perturbations Specification:

PERT_START
GM                  = 398600.4415 [km**3/s**2]
GRAVITY_MODEL       = EGM-96: 36D 360
ATMOSPHERIC_MODEL   = NRLMSIS00
N_BODY_PERTURBATIONS= MOON, SUN
SOLAR_F10P7         = 105.0
SOLAR_F10P7_MEAN   = 120.0
KSUBP               = 12.0
PERT_STOP

COMMENT             GEOCENTRIC, CARTESIAN, EARTH FIXED

ORB_START
ORB_REF_FRAME       = EFG
ORB_TYPE            = CARTPVA
0.000000 2789.6 -280.0 -1746.8 4.73 -2.50 -1.04 0.008 0.001 -0.159
ORB_STOP
    
```

**Figure 6-2: OCM example with space object characteristics and perturbations**

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS      = 3.0

CREATION_DATE       = 1998-11-06T09:23:57
ORIGINATOR          = JAXA

OBJECT_NAME         = GODZILLA 5
OBJECT_ID           = 1998-057A
EPOCH_TZERO        = 1998-12-18T14:28:15.1172
TIME_SYSTEM        = UTC

COMMENT S/C Physical Characteristics:
PHYS_START
MASS                = 100.0 [kg]
SOLAR_RAD_COEFF    = 1.300
DRAG_AREA          = 10.00 [m**2]
DRAG_COEFF         = 2.300
PHYS_STOP

COMMENT Perturbations specification
PERT_START
GM                  = 398600.4415          [km**3/s**2]
PERT_STOP

COMMENT              = 100-second in-track burn w/effic η=0.95, Isp=300s, 5% 1-sigma error
MAN_START
MAN_BASIS           = PREDICTED
MAN_PURPOSE         = SK
MAN_REF_FRAME       = RTN
MAN_TYPE            = THRUST
1 500.0 0.0 10.0 0.0 5.0 100.0 1 330.0 0.95 0.0
1 502.0 0.0 10.1 0.0 5.0 100.0 1 330.0 0.95 0.0
1 505.0 0.0 10.2 0.0 5.0 100.0 1 330.0 0.95 0.0
2 503.0 0.0 5.0 0.0 5.0 100.0 1 330.0 0.95 0.0
2 505.0 0.0 5.1 0.0 5.0 100.0 1 330.0 0.95 0.0
2 510.0 0.0 5.2 0.0 5.0 100.0 1 330.0 0.95 0.0
MAN_STOP

ORB_START
ORB_REF_FRAME       = TOD
ORB_FRAME_EPOCH    = 1998-12-18T14:28:15.1172
ORB_TYPE           = CARTPVA
0.000000 2789.6 -280.0 -1746.8 4.73 -2.50 -1.04 0.008 0.001 -0.159
10.000000 2783.4 -308.1 -1877.1 5.19 -2.42 -2.00 0.008 0.001 0.001
20.000000 2776.0 -336.9 -2008.7 5.64 -2.34 -1.95 0.008 0.001 0.159
< intervening data records omitted here >
500.000000 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535
ORB_STOP

ORB_START
ORB_REF_FRAME       = EME2000
ORB_TYPE           = KPLR
0.000000 6600.0 .03 28.5 50.0 30.0 10.0
10.000000 6600.0 .03 28.5 50.0 30.0 10.1
20.000000 6600.0 .03 28.5 50.0 30.0 10.2
< intervening data records omitted here >
500.000000 6600.0 .03 28.5 50.0 30.0 35.0
ORB_STOP

```

**Figure 6-3: OCM example with maneuvers, Cartesian and Keplerian ephemeris**



CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```

CCSDS_OCM_VERS      = 3.0

CREATION_DATE       = 1998-11-06T09:23:57
ORIGINATOR          = JAXA

OBJECT_NAME         = GODZILLA 5
OBJECT_ID           = 1998-057A
EPOCH_TZERO        = 1998-12-18T14:28:15.1172
TIME_SYSTEM         = UTC

COMMENT Perturbations specification
PERT_START
GM                  = 398600.4415          [km**3/s**2]
PERT_STOP

COMMENT S/C Physical Characteristics:
PHYS_START
MASS                = 1913.000          [kg]
SOLAR_RAD_AREA     = 10.000            [m**2]
SOLAR_RAD_COEFF    = 1.300
DRAG_AREA          = 10.000            [m**2]
DRAG_COEFF         = 2.300
PHYS_STOP

ORB_START
COMMENT             GEOCENTRIC, CARTESIAN, EARTH FIXED
ORB_CENTER_NAME    = EARTH
ORB_REF_FRAME      = ITRF-97
ORB_FRAME_EPOCH    = 1998-12-18T14:28:15.1172
ORB_TYPE           = CARTPVA

    0.000000 2789.6 -280.0 -1746.8 4.73 -2.50 -1.04 0.008 0.001 -0.159
    10.000000 2783.4 -308.1 -1877.1 5.19 -2.42 -2.00 0.008 0.001 0.001
    20.000000 2776.0 -336.9 -2008.7 5.64 -2.34 -1.95 0.008 0.001 0.159
    < intervening data records omitted here >
    500.000000 2164.375 1115.811 -688.131 -3.53328 -2.88452 0.88535

ORB_STOP

COV_START
COV_REF_FRAME      = EME2000
COV_TYPE           = ADBARV

T = 10.00
  3.331349e-04
  4.618927e-04  6.782421e-04
 -3.070007e-04 -4.221234e-04  3.231931e-04
 -3.349365e-07 -4.686084e-07  2.484949e-07  4.296022e-10
 -2.211832e-07 -2.864186e-07  1.798098e-07  2.608899e-10  1.767514e-10
 -3.041346e-07 -4.989496e-07  3.540310e-07  1.869263e-10  1.008862e-10  6.224444e-10
    < intervening data records omitted here >
T = 500.00
  3.442450e-04
  4.507816e-04  6.893532e-04
 -3.060006e-04 -4.110123e-04  3.342042e-04
 -3.238254e-07 -4.575073e-07  2.373838e-07  4.307133e-10
 -2.100721e-07 -2.753075e-07  1.687087e-07  2.507788e-10  1.878625e-10
 -3.030235e-07 -4.878385e-07  3.430200e-07  1.758152e-10  1.007751e-10  6.224444e-10
COV_STOP

COV_START
COV_TYPE           = EFG
T = 10.00
  3.331349e-04
  4.618927e-04  6.782421e-04
 -3.070007e-04 -4.221234e-04  3.231931e-04
COV_STOP

```

Figure 6-4: OCM example with Covariance Matrix

CCSDS PROPOSED STANDARD FOR ORBIT DATA MESSAGES

```
CCSDS_OCM_VERS = 3.0

CREATION_DATE = 1998-11-06T09:23:57
ORIGINATOR = JAXA

OBJECT_NAME = GODZILLA 5
OBJECT_ID = 1998-057A
EPOCH_TZERO = 1998-12-18T14:28:15.1172

COMMENT HERE IS A STATE TRANSITION MATRIX DATA BLOCK:
STM_START
STM_MAP_MODE = STATE
STM_TSTART = 0.0
STM_TYPE = CARTPV
STM_ORB_STATE = 2789.6 -280.0 -1746.8 4.73 -2.50 -1.04
STM_REF_FRAME = ICRF2000

T = 0.00
1.0 0.0 0.0 0.0 0.0 0.0
0.0 1.0 0.0 0.0 0.0 0.0
0.0 0.0 1.0 0.0 0.0 0.0
0.0 0.0 0.0 1.0 0.0 0.0
0.0 0.0 0.0 0.0 1.0 0.0
0.0 0.0 0.0 0.0 0.0 1.0

< intervening data records omitted here >

T = 500.00
1.23456 7.89012 3.45678 9.01234 5.67890 1.23456
7.89012 3.45678 9.01234 5.67890 1.23456 7.89012
3.45678 9.01234 5.67890 1.23456 7.89012 3.45678
9.01234 5.67890 1.23456 7.89012 3.45678 9.01234
5.67890 1.23456 7.89012 3.45678 9.01234 5.67890
1.23456 7.89012 3.45678 9.01234 5.67890 1.23456
STM_STOP

⋮ (continued on next page)
```

```

! (continued from previous page)

COMMENT  HERE IS AN EPHEMERIS COMPRESSION DATA BLOCK:
EC_START
_

COMMENT          HERE ARE EPHEM COMPR COEFS FOR DAY 1 (CHEBY W/FOURIER 2ND STAGE):
EC_TSTART        = 0.0
COMMENT          HERE IS THE BASELINE ORBIT, TO BE AUGMENTED BY CHEBY/FOURIER
EC_BASIS_PROP    = KEPLERIAN
EC_ORB_STATE     =
6700.0 0.0 0.0 0.0 0.0 0.839099633

EC_REPRESENT     = CHEBYSHEV
EC_STATE_TYPE    = EQUIN
EC_REF_FRAME     = ICRF2000
EC_REPR_N        = 10
1.23456 7.89012 3.45678 9.01234 5.67890 1.23456
  < intervening data records omitted here >
5.67890 1.23456 7.89012 3.45678 9.01234 5.67890

EC_REPRESENT     = FOURIER
EC_REPR_N        = 20
EC_STATE_TYPE    = EQUIN
EC_REF_FRAME     = ICRF2000

COMMENT  *** NOTE: Following are COSINE (col 1-6) and SINE (7-12) coeffs ***
1.234 7.890 3.456 9.012 5.678 1.234 1.234 7.890 3.456 9.012 5.678 1.234
  < intervening data records omitted here >
1.234 7.890 3.456 9.012 5.678 1.234 1.234 7.890 3.456 9.012 5.678 1.234
EC_TSTOP         = 86400.0

COMMENT          CHEBYSHEV POLYS FOR ORBIT STATES AT DAY 2 (SANS FOURIER CLEANUP):
EC_TSTART        = 86400.0
EC_BASIS_PROP    = NONE
EC_REF_FRAME     = ICRF2000

EC_REPRESENT     = CHEBYSHEV
EC_STATE_TYPE    = EQUIN
EC_REPR_N        = 30
3.45678 9.01234 5.67890 1.23456 7.89012 3.45678
  < intervening data records omitted here >
9.01234 5.67890 1.23456 7.89012 3.45678 9.01234
EC_TSTOP         = 172800.0

EC_STOP
    
```

**Figure 6-5: OCM example with STM (Cartesian position and velocity elements) and Ephemeris Compression (Equinoctial elements)**

## **7 ORBIT DATA MESSAGE SYNTAX**

### **7.1 OVERVIEW**

This section details the syntax requirements for each of the Orbit Data Messages.

### **7.2 GENERAL**

The Orbit Data Messages (OPM, OMM, OEM or OCM) shall observe the syntax described in 7.3 through 7.7.

### **7.3 ODM LINES**

**7.3.1** Each OPM, OMM, OEM or OCM line shall be one of the following:

- Header line;
- Metadata line;
- Comment line;
- Data line; or
- Blank line.

**7.3.2** Each OPM, OMM, or OEM line must not exceed 254 ASCII characters and spaces (excluding line termination character[s]).

**7.3.3** OCM lines may be of arbitrary length. If exchange between the two parties requires a maximum line length, that limit should be negotiated and specified in an ICD.

**7.3.4** Only printable ASCII characters and blanks shall be used. Control characters (such as TAB, etc.) shall not be used, with the exception of the line termination characters specified below.

**7.3.5** Blank lines may be used at any position within the file. Blank lines shall have no assignable meaning, and may be ignored.

**7.3.6** The first header line must be the first non-blank line in the file.

**7.3.7** All lines shall be terminated by a single Carriage Return or a single Line Feed, or a Carriage Return/Line Feed pair or a Line Feed/Carriage Return pair.

## **7.4 KEYWORD = VALUE NOTATION AND ORDER OF ASSIGNMENT STATEMENTS**

**7.4.1** For the OPM and OMM, all header, metadata, and data lines shall use ‘keyword = value’ notation, abbreviated as KVN.

**7.4.1.1** For the OEM, all header and metadata elements shall use KVN notation.

**7.4.1.2** OEM ephemeris data lines shall not use KVN format; rather, the OEM ephemeris data line has a fixed structure containing seven required fields (epoch time, three position components, three velocity components), and three optional acceleration components. (See 5.2.4.)

**7.4.1.3** OEM covariance matrix epoch and covariance reference frame (if used) shall use KVN format. The OEM covariance data lines shall not use KVN format; rather, the OEM covariance data line has a fixed structure containing from one to six required fields (a row from the 6x6 lower triangular form covariance matrix). (See 5.2.5.)

**7.4.1.4** For the OCM, all header and metadata elements shall use KVN notation.

**7.4.1.5** OCM orbit state time history data lines shall not use KVN format; rather, the structure of such OCM orbit state time history data lines is comprised of time relative to EPOCH\_TZERO followed by the parameters corresponding to the selected orbit set (See 6.2.7).

**7.4.1.6** OCM covariance matrix epoch and covariance reference frame (if used) shall use KVN format. The OCM covariance data lines shall not use KVN format; rather, the OCM covariance data line has a fixed structure containing from one to “N” required fields (a row from the N x N lower triangular form of a square covariance matrix). (See 0.)

**7.4.2** The keywords ‘COMMENT’, [wild card]‘\_START’ and [wild card]‘\_STOP’ are exceptions to the KVN syntax assignment.

**7.4.3** Only a single ‘keyword = value’ assignment shall be made on a line.

**7.4.4** Keywords must be uppercase and must not contain blanks.

**7.4.5** Any white space immediately preceding or following the keyword shall not be significant.

**7.4.6** Any white space immediately preceding or following the ‘equals’ sign shall not be significant.

**7.4.7** Any white space immediately preceding the end of line shall not be significant.

**7.4.8** The order of occurrence of mandatory and optional KVN assignments shall be fixed as shown in the tables in sections 3, 4, 5 and 6 that describe the OPM, OMM, OEM and OCM keywords.

## 7.5 VALUES

**7.5.1** A non-empty value field must be specified for each mandatory keyword.

**7.5.2** Integer values shall consist of a sequence of decimal digits with an optional leading sign ('+' or '-'). If the sign is omitted, '+' shall be assumed. Leading zeroes may be used. The range of values that may be expressed as an integer is:

$$-2,147,483,648 \leq x \leq +2,147,483,647 \quad (\text{i.e., } -2^{31} \leq x \leq 2^{31}-1).$$

**7.5.3** Non-integer numeric values may be expressed in either fixed-point or floating-point notation. Both representations may be used within an OPM, OMM, OEM or OCM .

**7.5.4** Non-integer numeric values expressed in fixed-point notation shall consist of a sequence of decimal digits separated by a period as a decimal point indicator, with an optional leading sign ('+' or '-'). If the sign is omitted, '+' shall be assumed. Leading and trailing zeroes may be used. At least one digit shall appear before and after a decimal point.

**7.5.5** Non-integer numeric values expressed in floating point notation shall consist of a sign, a mantissa, an alphabetic character indicating the division between the mantissa and exponent, and an exponent, constructed according to the following rules:

- a) The sign may be '+' or '-'. If the sign is omitted, '+' shall be assumed.
- b) The mantissa must be a string of no more than 16 decimal digits with a decimal point ('.') in the second position of the ASCII string, separating the integer portion of the mantissa from the fractional part of the mantissa.
- c) The character used to denote exponentiation shall be 'E' or 'e'. If the character indicating the exponent and the following exponent are omitted, an exponent value of zero shall be assumed (essentially yielding a fixed point value).
- d) The exponent must be an integer, and may have either a '+' or '-' sign (if the sign is omitted, then '+' shall be assumed).
- e) The maximum positive floating point value is approximately 1.798E+308, with 16 significant decimal digits precision. The minimum positive floating point value is approximately 4.94E-324, with 16 significant decimal digits precision.

NOTE – These specifications for integer, fixed point and floating point values conform to the XML specifications for the data types four-byte integer 'xsd:int', 'decimal', and 'double', respectively (reference [6]). The specifications for floating point values conform to the IEEE double precision type (references [6] and [7]). Floating point numbers in IEEE extended-single or IEEE extended-double precision may be represented, but do require an ICD between exchange partners because of their implementation-specific attributes (reference [7]). The special values 'NaN', '-Inf', '+Inf', and '-0' are not supported in the ODM.

**7.5.6** Text value fields must be constructed using only all uppercase or all lowercase.

**7.5.7** Blanks shall not be permitted within numeric values and time strings.

**7.5.8** In value fields that are text, an underscore shall be equivalent to a single blank. Individual blanks shall be retained (shall be significant), but multiple contiguous blanks shall be equivalent to a single blank.

**7.5.9** In value fields that represent an absolute time tag or epoch, times shall be given in one of the following two formats:

YYYY-MM-DDThh:mm:ss[.d→d][Z]

or

YYYY-DDDThh:mm:ss[.d→d][Z]

where ‘YYYY’ is the year, ‘MM’ is the two-digit month, ‘DD’ is the two-digit day, ‘DDD’ is the three-digit day of year, ‘T’ is constant, ‘hh:mm:ss[.d→d]’ is the time in hours, minutes seconds, and optional fractional seconds; ‘Z’ is an optional time code terminator (the only permitted value is ‘Z’ for Zulu, i.e., UTC). As many ‘d’ characters to the right of the period as required may be used to obtain the required precision, up to the maximum allowed for a fixed point number. All fields shall have leading zeros. (See reference [1], ASCII Time Code A or B.). Where such epochs occur within one second after leap second introduction, the hh:mm:ss portion of the above time specification shall use the convention XX:XX:60.XXXX.

**7.5.10** The time system for CREATION\_DATE is UTC; for all other keywords representing times or epochs, the time system is determined by the TIME\_SYSTEM metadata keyword.

## 7.6 UNITS IN THE ORBIT DATA MESSAGES

### 7.6.1 OPM/OMM/OCM UNITS

**7.6.1.1** For documentation purposes and clarity, units may be included as ASCII text after a value in the OPM, OMM and OCM. If units are displayed, they must exactly match the units (including lower/upper case) as specified in tables 3-3, 4-3, 5-3 and 6-4 through 6-12. If units are displayed, then:

- a) there must be at least one blank character between the value and the units text;
- b) the units must be enclosed within square brackets (e.g., ‘[km]’);
- c) exponents of units shall be denoted with a double asterisk (i.e., ‘\*\*’, for example,  $m/s^2=m/s^{**2}$ ).

**7.6.1.2** Some of the items in the applicable tables are dimensionless. The table shows a unit value of ‘n/a’, which in this case means that there is no applicable units designator for these items (e.g., for ECCENTRICITY). The notation ‘[n/a]’ should not appear in an OPM, OCM or OMM.

## 7.6.2 OEM UNITS

**7.6.2.1** In an OEM ephemeris data line, units shall be km, km/s, and  $\text{km/s}^2$  for position, velocity, and acceleration components, respectively, but the units shall not be displayed.

**7.6.2.2** In an OEM covariance matrix line, units shall be  $\text{km}^2$ ,  $\text{km}^2/\text{s}$ , or  $\text{km}^2/\text{s}^2$  depending on whether the element is computed from two position components, one position component and one velocity component, or two velocity components. The units shall not be displayed.

## 7.6.3 OCM UNITS

**7.6.3.1** In an OCM orbit state data line, units shall be degrees for angular quantities, kilometers for distance quantities and seconds for time quantities. The units shall not be displayed.

**7.6.3.2** In an OCM covariance matrix line, units shall be comprised of the requisite combination of degrees for angular quantities, kilometers for distance quantities and seconds for time quantities. The units shall not be displayed.

**7.6.3.3** In an OCM state transition matrix line, units shall be compatible with the corresponding orbit type's requisite combination of degrees for angular quantities, kilometers for distance quantities and seconds for time quantities. The units shall not be displayed.

**7.6.3.4** Units for all other OCM quantities are as specified in section 6.2.

## 7.7 COMMENTS IN THE ORBIT DATA MESSAGES

**7.7.1** There are certain pieces of information that provide clarity and remove ambiguity about the interpretation of the information in a file, yet are not standardized so as to fit cleanly into the 'keyword = value' paradigm. Rather than force the information to fit into a space limited to one line, the ODM producer should put certain information into comments and use the ICD to provide further specifications.

**7.7.2** Comments may be used to provide provenance information or to help describe dynamical events or other pertinent information associated with the data. This additional information is intended to aid in consistency checks and elaboration where needed, but shall not be required for successful processing of a file.

**7.7.3** For the OPM, OMM, OEM and OCM, comment lines shall be optional.

**7.7.4** All comment lines shall begin with the 'COMMENT' keyword followed by at least one space. This keyword must appear on every comment line, not just the first such line. The remainder of the line shall be the comment value. White space shall be retained (shall be significant) in comment values.



**7.7.5** Placement of comments shall be as specified in the tables in sections 3, 4, 5 and 6 that describe the OPM, OMM, OEM and OCM keywords.

**7.7.6** Comments in the OPM may appear in the OPM Header immediately after the ‘CCSDS\_OPM\_VERS’ keyword, at the very beginning of the OPM Metadata section, and at the beginning of a logical block in the OPM Data section. Comments must not appear between the components of any logical block in the OPM Data section.

NOTE – The logical blocks in the OPM Data section are indicated in table 3-3.

**7.7.7** Comments in the OMM may appear in the OMM Header immediately after the ‘CCSDS\_OMM\_VERS’ keyword, at the very beginning of the OMM Metadata section, and at the beginning of a logical block in the OMM Data section. Comments must not appear between the components of any logical block in the OMM Data section.

NOTE – The logical blocks in the OMM Data section are indicated in table 4-3.

**7.7.8** Comments in the OEM may appear in the OEM Header immediately after the ‘CCSDS\_OEM\_VERS’ keyword, at the very beginning of the OEM Metadata section (after the ‘META\_START’ keyword), at the beginning of the OEM Ephemeris Data Section, and at the beginning of the OEM Covariance Data section (after the ‘COV\_START’ keyword). Comment lines must not appear within any block of ephemeris lines or covariance matrix lines.

**7.7.9** Comments in the OCM may appear anywhere within the OCM Header, Metadata, Space Object Physical Characteristics, Force Model, Maneuver, Orbit State Time History, Covariance Time History, and State Transition Matrix Time History data sections.

**7.7.10** Extensive comments in an ODM are recommended in cases where there is insufficient time to negotiate an ICD.

**7.7.11** The following comments should be provided:

- a) Information regarding the genesis, history, interpretation, intended use, etc., of the state vector, spacecraft, maneuver, or ephemeris that may be of use to the receiver of the OPM, OMM, or OEM:

COMMENT Source: File created by JPL Multi-Mission Navigation Team as part  
COMMENT of Launch Operations Readiness Test held on 20 April 2001.

- b) Natural body ephemeris information: When the Earth is not the center of motion, the ephemerides of the planets, satellites, asteroids, and/or comets (including associated constants) consistent with the ODM should be identified so that the recipient can, in a consistent manner, make computations involving other centers:

COMMENT Based on latest orbit solution which includes observations  
COMMENT through 2000-May-15 relative to planetary ephemeris DE-0405.

- c) OEM accuracy vs. efficiency: If the covariance data section of the OEM is not utilized, the producer of an OEM should report in comment lines what the expected accuracy of the ephemeris is, so the user can smooth or otherwise compress the data without affecting the accuracy of the trajectory. The OEM producer also should strive to achieve not only the best accuracy possible, taking into account prediction errors, but also consider the efficiency of the trajectory representation (e.g., step sizes of fractional seconds between ephemeris lines may be necessary for precision scientific reconstruction of an orbit, but are excessive from the standpoint of antenna pointing predicts generation).

## **7.8 ORBIT DATA MESSAGE KEYWORDS**

### **7.8.1 VERSION KEYWORDS**

The Header of the OPM, OMM, OEM and OCM shall provide a CCSDS Orbit Data Message version number that identifies the format version; this is included to anticipate future changes. The version keywords for the OPM, OMM, OEM and OCM shall be `CCSDS_OPM_VERS`, `CCSDS_OMM_VERS`, `CCSDS_OEM_VERS` and `CCSDS_OCM_VERS`, respectively. The value shall have the form of 'x.y', where 'y' shall be incremented for corrections and minor changes, and 'x' shall be incremented for major changes. Version x.0 shall be reserved for versions accepted by the CCSDS as an official Recommended Standard ('Blue Book'). Testing shall be conducted using OPM, OMM, OEM and OCM version numbers less than 1.0 (e.g., 0.x). Exchange participants should specify in the ICD the specific OPM, OMM, OEM and OCM version numbers they will support. The following version numbers are supported:

<b>Version Keyword</b>	<b>Version Number</b>	<b>Applicable Recommendation</b>
CCSDS_OPM_VERS	1.0	Silver Book 1.0, 09/2004
CCSDS_OPM_VERS	2.0	Silver Book 2.0, 09/2009
CCSDS_OMM_VERS	2.0	Silver Book 2.0, 09/2009
CCSDS_OEM_VERS	1.0	Silver Book 1.0, 09/2004
CCSDS_OEM_VERS	2.0	Silver Book 2.0, 09/2009
CCSDS_OPM_VERS	3.0	Blue Book 3.0 (this document)
CCSDS_OMM_VERS	3.0	Blue Book 3.0 (this document)
CCSDS_OEM_VERS	3.0	Blue Book 3.0 (this document)
CCSDS_OCM_VERS	3.0	Blue Book 3.0 (this document)

## **7.8.2 GENERAL KEYWORDS**

**7.8.2.1** Only those keywords shown in table 3-1, table 3-2, and table 3-3 shall be used in an OPM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

**7.8.2.2** Only those keywords shown in table 4-1, table 4-2, and table 4-3 shall be used in an OMM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

**7.8.2.3** Only those keywords shown in table 5-2 and table 5-3 shall be used in an OEM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

**7.8.2.4** Only those keywords shown in tables 6-2, table 6-3, table 6-4, table 6-5, table 6-6, table 6-7, table 6-8, table 6-9, and table 6-10 shall be used in an OCM. Some keywords represent mandatory items and some are optional. KVN assignments representing optional items may be omitted.

## ANNEX A

**IMPLEMENTATION CONFORMANCE****STATEMENT PRO FORMA****(NORMATIVE)****A1 INTRODUCTION****A1.1 OVERVIEW**

This annex provides the Implementation Conformance Statement (ICS) Requirements List (RL) for an implementation of the Orbit Data Message (CCSDS 502.0). The ICS for an implementation is generated by completing the RL in accordance with the instructions below. An implementation shall satisfy the mandatory conformance requirements referenced in the RL.

- The RL in this annex is blank. An implementation's completed RL is called the ICS. The ICS states which capabilities and options have been implemented. The following can use the ICS:
  - the implementer, as a checklist to reduce the risk of failure to conform to the standard through oversight;
  - a supplier or potential acquirer of the implementation, as a detailed indication of the capabilities of the implementation, stated relative to the common basis for understanding provided by the standard ICS proforma;
  - a user or potential user of the implementation, as a basis for initially checking the possibility of interworking with another implementation (it should be noted that, while interworking can never be guaranteed, failure to interwork can often be predicted from incompatible ICS lists);
  - a tester, as the basis for selecting appropriate tests against which to assess the claim for conformance of the implementation.

**A1.2 ABBREVIATIONS AND CONVENTIONS**

The RL consists of information in tabular form. The status of features is indicated using the abbreviations and conventions described below.

Item Column

The item column contains sequential numbers for items in the table.

Feature Column

The feature column contains a brief descriptive name for a feature. It implicitly means “Is this feature supported by the implementation?”

Status Column

The status column uses the following notations:

- M            mandatory;
- O            optional;
- C            conditional;
- X            prohibited;
- I            out of scope;
- N/A         not applicable.

Support Column Symbols

The support column is to be used by the implementer to state whether a feature is supported by entering Y, N, or N/A, indicating:

- Y        Yes, supported by the implementation.
- N        No, not supported by the implementation.
- N/A     Not applicable.

The support column should also be used, when appropriate, to enter values supported for a given capability.

**A1.3 INSTRUCTIONS FOR COMPLETING THE RL**

An implementer shows the extent of compliance to the Recommended Standard by completing the RL; that is, the state of compliance with all mandatory requirements and the options supported are shown. The resulting completed RL is called an ICS. The implementer shall complete the RL by entering appropriate responses in the support or values supported column, using the notation described in A1.2. If a conditional requirement is inapplicable, N/A should be used. If a mandatory requirement is not satisfied, exception information must be supplied by entering a reference  $X_i$ , where  $i$  is a unique identifier, to an accompanying rationale for the noncompliance.

**A2 ICS PROFORMA FOR ORBIT DATA MESSAGE****A2.1 IDENTIFICATION OF ICS**

Date of Statement (DD/MM/YYYY)	
ICS serial number	
System Conformance statement cross-reference	

**A2.2 IDENTIFICATION OF IMPLEMENTATION UNDER TEST (IUT )**

Implementation name	
Implementation version	
Special Configuration	
Other Information	

**A2.3 IDENTIFICATION OF SUPPLIES**

Supplier	
Contact Point for Queries	
Implementation Name(s) and Versions	
Other information necessary for full identification, e.g., name(s) and version(s) for machines and/or operating systems; System Name(s)	

**A2.4 DOCUMENT VERSIONS**

CCSDS 503.0 Document Version	
<p>Have any exceptions been required?</p> <p>(Note: A YES answer means that the implementation does not conform to the Recommended Standard. Non-supported mandatory capabilities are to be identified in the ICS, with an explanation of why the implementation is non-conforming.)</p>	<p>Yes _____ No_____</p>

## A2.5 REQUIREMENTS LISTS

[See CCSDS A20.1-Y-1, *CCSDS Implementation Conformance Statements* (Yellow Book, Issue 1, April 2014).]

Item	Feature	Keyword	Reference (Blue book)	Status (M/O/...)	Support



## ANNEX B

**VALUES FOR TIME\_SYSTEM AND FRAME RELATED KEYWORDS****(NORMATIVE)**

The values in this annex represent the set of acceptable values for the TIME\_SYSTEM, REF\_FRAME, OEB\_FRAME, MAN\_REF\_FRAME, ORB\_REF\_FRAME, COV\_REF\_FRAME and STM\_REF\_FRAME keywords in the OPM, OMM, OEM and OCM. (For details and description of these time systems, see reference [L1]) If exchange partners wish to use different settings, the settings should be documented in the ICD.

**B1 TIME\_SYSTEM METADATA KEYWORD**

<b>Time System Value</b>	<b>Meaning</b>
BEIDOU	Beidou
GALILEO	Galileo
GLONASS	Glonass
GMST	Greenwich Mean Sidereal Time
GPS	Global Positioning System
MET	Mission Elapsed Time (note)
MRT	Mission Relative Time (note)
NAVIC	Navic
SCLK	Spacecraft Clock (receiver) (requires rules for interpretation in ICD)
TAI	International Atomic Time
TCB	Barycentric Coordinate Time
TDB	Barycentric Dynamical Time
TCG	Geocentric Coordinate Time
TT	Terrestrial Time
UT1	Universal Time
UTC	Coordinated Universal Time
ICD	Other timing system, as defined in ICD

If MET or MRT is chosen as the TIME\_SYSTEM, then the epoch of either the start of the mission for MRT, or of the event for MET, should either be given in a comment in the message or provided in an ICD. The time system for the start of the mission or the event should also be provided in the comment or the ICD. If these values are used for the TIME\_SYSTEM, then the times given in the file denote a duration from the mission start or event. However, for clarity, an ICD should be used to fully specify the interpretation of the times if these values are to be used. The time format should only utilize three digit days from the MET or MRT epoch, not months and days of the months.

## **B2 REFERENCE FRAME KEYWORDS**

Fixed Reference Frame Value	Meaning
DTRFyyyy	The DTRFyyyy (e.g. DTRF2014) is the ITRS realization considering corrections for non-tidal atmospheric and hydrological loading, as of year “yyyy” (e.g. 2000)
EFG	Earth-Fixed Greenwich (E, F, G)
EME2000	Earth Mean Equator and Equinox of J2000
GCRF	Geocentric Celestial Reference Frame
GRC	Greenwich Rotating Coordinates
ICRFyyyy	International Celestial Reference Frame (Barycentric) solution as of year “yyyy” (e.g. 2000)
ITRFyyyy	International Terrestrial Reference Frame solution as of year “yyyy” (e.g. 2000)
ITRF-93	International Terrestrial Reference Frame 1993
ITRF-97	International Terrestrial Reference Frame 1997
MCI	Mars Centered Inertial
MEME	Mean Equator Mean Equinox
MOON_ME	Moon Mean Earth (ME) frame, which has its X axis pointed along the mean direction to the center of the Earth and the Z axis pointing to the mean direction of rotation. The ME frame is typically used to specify the location of objects on the Moon.
MOON_MEIAUE	Moon-Centered, Moon Mean Equator and IAU-Node of Epoch frame as specified in [L11, Fig. 6-2].
MOON_PA	Moon Principal Axis (PA) frame which is defined by the inertial tensor of the Moon. The PA frame is used as the basis for Lunar gravity models, in the numerical integration of the planetary ephemerides, and as the reference for modern moon gravity solutions. Euler angles supplied as part of the JPL DE planetary ephemerides relate the MOON_PA frame to ICRF.
TDR	True of Date, Rotating (Realized as ITRF Fixed)
TEME	True Equator Mean Equinox (see below NORAD comment)
TOD	True of Date (True Equator True Equinox)
UVW	Launch go-inertial reference frame, with U in local horizon plane along inertial launch azimuth (downrange), W along the geodetic vertical and V completing the set (cross-range). In typical use the go-inertial epoch should be specified in an accompanying comment field.
ICD	Other reference frame, as defined in ICD

NORAD Two Line Element Sets are implicitly in a True Equator Mean Equinox (TEME) reference frame, which is ill defined in international standard or convention. TEME may be used only for OMMs based on NORAD Two Line Element sets, and in no other circumstances. There are subtle differences between TEME of Epoch and TEME of Date (see reference [L3] or [L4]). The effect is very small relative to TLE accuracy, and there is uncertainty regarding which of these is used by NORAD. The preferred option is TEME of Date. Users should specify in the ICD if their assumption is TEME of Epoch.

### B3 RELATIVE REFERENCE FRAME KEYWORDS

In addition to the above reference frames, maneuver and covariance data can be specified in the following relative frames:

Relative Reference Frame Value	Meaning
ACTUATOR_x	Actuator reference frame ('x' = 0→9): could denote reaction wheels, solar arrays, thrusters, etc.
CSS_xy	Coarse Sun Sensor ('x' = 0→9, 'y' = 0→9)
DSS_x	Digital Sun Sensor ('x' = 0→9)
GYRO_x	Gyroscope Reference Frame ('x' = 0→9)
INSTRUMENT_y	Instrument 'y' reference frame ('y' = A→Z, 0→9)
NSW	"NADIR, Sun, Normal" – This frame aligns the x-axis in the NADIR direction, the y-axis as much as possible toward the Sun while still being normal to the x-axis, and the z-axis completing the right-hand set
RIC	'Radial, In-track, Cross-track'
RSW	Another name for 'Radial, Transverse, Normal'
RTN	Radial, Transverse, Normal (QSW)
SC_BODY_x	Spacecraft Body Frame ('x' = 0→9); requires clear specification via ICD
SC_BODY_y	Spacecraft Body Frame of another object ('y' = A→Z); requires clear specification via ICD
SENSOR_x	Sensor 'x' reference frame ('x' = A→Z, 0→9)
STARTRACKER_x	Star Tracker Reference Frame ('x' = 0→9)
TAM_x	Three Axis Magnetometer Reference Frame ('x' = 0→9)
TNW	A local orbital coordinate frame that has the x-axis along the Tangential (or velocity) vector, z-axis ("W") along the orbital angular momentum vector ( $\vec{\omega} = \vec{r} \times \vec{v}$ ), and N completing the

	right handed system (i.e., for a circular orbit “N” points in the Nadir direction and for an eccentric orbit, “N” points as close to Nadir as possible while still being normal to the T-W plane).
VNC	A local orbital coordinate frame that has the x-axis along the Velocity (or tangential) vector, y-axis Normal to the orbit along the orbital angular momentum vector ( $\vec{\omega} = \vec{r} \times \vec{v}$ ), and z-axis is the “Co-normal” direction completing the right handed system (i.e., for a circular orbit “C” points in the radius vector direction whereas for an eccentric orbit, “C” points as close to radial as possible while still being normal to the V-N plane).
ICD	Other relative reference frame, as defined in ICD

## B4 ELEMENT SET KEYWORDS

Orbit element states and/or time histories may be specified in the following element sets.

Orbit elements shall be interpreted as osculating elements unless pre-coordinated between the message originator and recipient to contain mean elements (e.g. singly- or doubly-averaged elements based upon Kozai, Brouwer or other theories).

It is not allowed to specify non-inertial reference frames when employing inertial element sets, or to specify inertial reference frames when employing non-inertial element sets.

Orbit Element Set Value	Meaning
ADBARV	Spherical 6-element set ( $\alpha\delta\beta Arv$ : right ascension +E°, declination +N°, inertial flight path angle measured from the radial direction to inertial velocity direction (e.g. 90° for circular orbit), inertial azimuth angle, measured from local North to projection of inertial velocity in local horizontal plane, radius magnitude and velocity magnitude)
CARTP	Cartesian 3-element position (only) orbit state (X, Y, Z)
CARTPV	Cartesian 6-element position and velocity orbit state (X, Y, Z, XD, YD, ZD)
CARTPVA	Cartesian 9-element position, velocity and acceleration orbit state (X, Y, Z, XD, YD, ZD, XDD, YDD, ZDD)
EQUIN	Equinoctial 7-element set ( $[ahk\lambda pqf_r] = [a, a_g, a_f, L=(\Omega + \omega + f_r M), \chi, \psi, f_r = \pm 1]$ as defined in Vallado [L9])
EQUINMOD	Equinoctial 7-element modified set ( $[pfghkLf_r] = [a(1-e^2), a_f, a_g, \chi, \psi, L = (\Omega + \omega + f_r v), f_r = \pm 1]$ as defined in Vallado [L9])
KPLR	Keplerian 6-element classical set ( $aei\Omega\omega v$ : semi-major axis, eccentricity, inclination, right ascension of the ascending node, argument of perigee and true anomaly)
KPLRM	Keplerian 6-element classical set ( $aei\Omega\omega M$ : semi-major axis, eccentricity, inclination, right ascension of the ascending node, argument of perigee and mean anomaly)
LDBARV	Modified spherical 6-element set ( $\lambda\delta\beta Arv$ : Earth longitude +E°, declination +N°, inertial flight path angle measured from the radial direction to inertial velocity direction (e.g. 90° for circular orbit), inertial azimuth angle, measured from local North to projection of inertial velocity in local horizontal plane, radius magnitude and velocity magnitude)
ICD	Other element set definition, as defined in ICD

**B5 ADDITIONAL COVARIANCE SET KEYWORDS**

In addition to the above orbit element sets, covariance data can be specified in the following orbit sets:

<b>Orbit Element Set Value</b>	<b>Meaning</b>
TCARTP	4x4: Time & Cartesian 3-element position (only) errors (X, Y, Z)
TCARTPV	7x7: Time & Cartesian 6-element position and velocity errors (X, Y, Z, XD, YD, ZD)
TCARTPVA	10x10: Time & Cartesian 9-element position, velocity and acceleration errors (X, Y, Z, XD, YD, ZD, XDD, YDD, ZDD)
TEIGVAL3EIGVEC3	13-element eigenvalue/eigenvector representation time history corresponding to the 3x3 position covariance time history, with each line containing Time, the three (major, medium and minor) eigenvalues IN DESCENDING ORDER, and the corresponding three eigenvectors matching the major, medium and minor eigenvalues
TKPLR	7x7: Time & Keplerian 6-element classical set ( $aei\Omega\omega\nu$ : semi-major axis, eccentricity, inclination, right ascension of the ascending node, argument of perigee and true anomaly) errors
TKPLRM	7x7: Time & Keplerian 6-element classical set ( $aei\Omega\omega M$ : semi-major axis, eccentricity, inclination, right ascension of the ascending node, argument of perigee and mean anomaly) errors
TADBARV	7x7: Time & Spherical 6-element set ( $\alpha\delta\beta Arv$ : right ascension +E°, declination +N°, inertial flight path angle measured from the radial direction to inertial velocity direction (e.g. 90° for circular orbit), inertial azimuth angle, measured from local North to projection of inertial velocity in local horizontal plane, radius magnitude and velocity magnitude) errors
TEQUIN	7x7: Time & Equinoctial 6-element set ( $[ahk\lambda pq] = [a, a_g, a_f, L=(\Omega + \omega + f_r M), \chi, \psi]$ as defined in Vallado [L9]) errors
TEQUINMOD	7x7: Time & Equinoctial 6-element modified set ( $[pfghkL] = [a(1-e^2), a_f, a_g, \chi, \psi, L = (\Omega + \omega + f_r \nu)]$ per Vallado [L9])
TLDBARV	7x7: Time & Modified spherical 6-element set ( $\lambda\delta\beta Arv$ : Earth longitude +E°, declination +N°, inertial flight path angle measured from the radial direction to inertial velocity direction (e.g. 90° for circular orbit), inertial azimuth angle, measured from local North to projection of inertial velocity in local horizontal plane, radius magnitude and velocity magnitude) errors



COV_NNXNN	Generic NN x NN covariance containing “NN” rows and columns, with “NN” containing a TWO-DIGIT (including leading zero) representation) of the covariance size. Valid examples are: COV_02X02, COV_12X12. The contents of the covariance matrix are as defined in adjacent COMMENTS or ICD
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## ANNEX C

## SATELLITE PHYSICAL CHARACTERISTICS SPECIFICATION

## (INFORMATIVE)

## C1 OVERVIEW

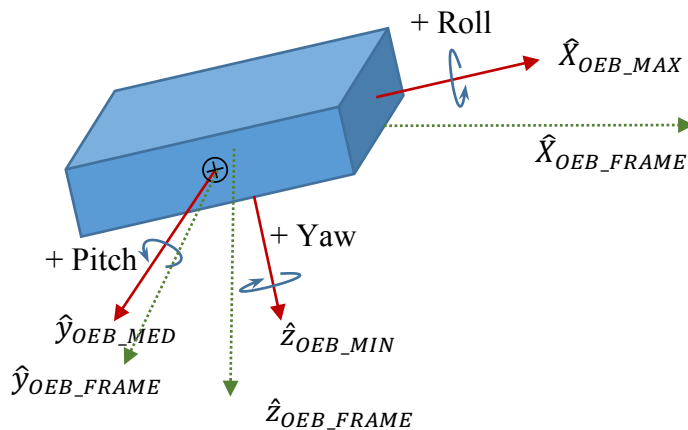
This annex defines satellite dimensional and orientational parameters of the OCM's satellite physical characteristics specification.

To facilitate improved modeling of space object attitude, hard body collision, and drag and SRP acceleration forces, the OCM allows the specification of an “**Optimally-Encompassing Box**” (OEB).

For a box-shaped satellite (e.g., a CubeSat) without appendages, the satellite and its corresponding OEB are one and the same. For a satellite having solar arrays that extend from the spacecraft body structure, the OEB would extend from the main satellite body to encompass the deployed solar arrays as well.

As shown in the figure below, the OEB reference frame axes (depicted in RED) are defined by convention as follows:

- The OEB x-axis is along the **longest** dimension of the OEB ( $\hat{X}_{OEB\_MAX}$ )
- The OEB y-axis is along the **intermediate** dimension ( $\hat{Y}_{OEB\_MED}$ )
- The OEB z-axis is along the **short** dimension ( $\hat{Z}_{OEB\_MIN}$ ).



A fixed orientation of the OEB with respect to the user-specified “OEB\_FRAME” is defined using an ordered sequence of Euler rotations that map from the user-specified OEB\_FRAME to the Optimally-Encompassing Box vector directions. The above figure shows the proper definitions and adopted sign conventions for Yaw, Pitch and Roll angles. The resulting transformation sequence is:

$$\begin{bmatrix} x \\ y \\ z \end{bmatrix}_{\text{OEB}} = \begin{bmatrix} \cos(Yaw) & -\sin(Yaw) & 0 \\ \sin(Yaw) & \cos(Yaw) & 0 \\ 0 & 0 & 1 \end{bmatrix} \times \begin{bmatrix} \cos(Pitch) & 0 & \sin(Pitch) \\ 0 & 1 & 0 \\ -\sin(Pitch) & 0 & \cos(Pitch) \end{bmatrix} \\ \times \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos(Roll) & -\sin(Roll) \\ 0 & \sin(Roll) & \cos(Roll) \end{bmatrix} \begin{bmatrix} x \\ y \\ z \end{bmatrix}_{\text{OEB\_FRAME}}$$

The physical dimensions of the OEB (long, intermediate and short dimensions) are specified via OEB\_MAX, OEB\_MED and OEB\_MIN respectively.

The cross-sectional area as viewed along the OEB x, y and z axes (long, intermediate and short dimension directions) are specified via AREA\_ALONG\_OEB\_MAX, AREA\_ALONG\_OEB\_MED and AREA\_ALONG\_OEB\_MIN, respectively.

## ANNEX D

## APPARENT-TO-ABSOLUTE VISUAL MAGNITUDE RELATIONSHIP (INFORMATIVE)

### D1 OVERVIEW

This annex presents the relationships to be used to map apparent to absolute visual magnitude for inclusion in an OCM. These equations based on reference [L12] examine signal magnitude for reflected illumination by an exoatmospheric Resident Space Object (RSO). The equations do not account for spatial distribution across multiple detectors, which involves characterizing the Point Spread Function of the system.

Definitions:

$E_{EntranceAperture}$	Target's specific entrance aperture radiance [ $W/m^2$ ]
$I_{Sun}$	Solar Intensity $\approx 3.088374161 \times 10^{25}$ [W]
$d_{SunToTarget}$	Distance from the sun to the target (e.g. 1 AU = $1.4959787066 \times 10^{11}$ m)
$E_{Sun}$	Exoatmospheric solar irradiance (nominally 1380 [ $W/m^2$ ] at 1 AU)
$\varphi$	Phase or CATS angle from sun to the sensor, relative to target [rad]
$Phase(\varphi)$	Geometric reflectance function [between 0 and 1]
$F$	General shadowing term accounting for the penumbra region's influence [unitless ratio between 0 = umbra and 1 = full Sun illumination]
$A_{Target}$	Effective area of the target [ $m^2$ ]
$\pi$	Pi constant
$\rho$	Reflectance of the target [between 0 (none) and 1 (perfect reflectance)]
$I_{Target}$	Intensity of reflected energy from target treated as a point source [W]
$E_{Target}$	Target Irradiance at Sensor w/o atmos loss [ $W/m^2$ ]
$r_{Target}$	Effective radius of the target [ $m^2$ ]
$d_{TargetToSensor}$	Distance from target to sensor [m]
$\tau_{Atmosphere}$	Atmospheric transmission [unitless between 0 and 1]
$E_0$	Ref. Visual Magnitude (Vega) Irradiance [ $2.77894 \times 10^{-8} W/m^2$ ]

Given an optical sensor's measured target entrance aperture radiance:

$$VM_{apparent} = -2.5 \log_{10} \frac{E_{target}}{E_0} \text{ [vmag]}$$

$$E_{target} = \frac{E_{EntranceAperture}}{\tau_{Atmosphere}(\theta)} \text{ [W/m}^2\text{]}$$

or if  $VM_{apparent}$  known:  $E_{target} = E_0 10^{\left[-\frac{VM_{apparent}}{2.5}\right]}$

$$I_{target} = E_{target} d_{TargetToSensor}^2 \text{ [W]}$$

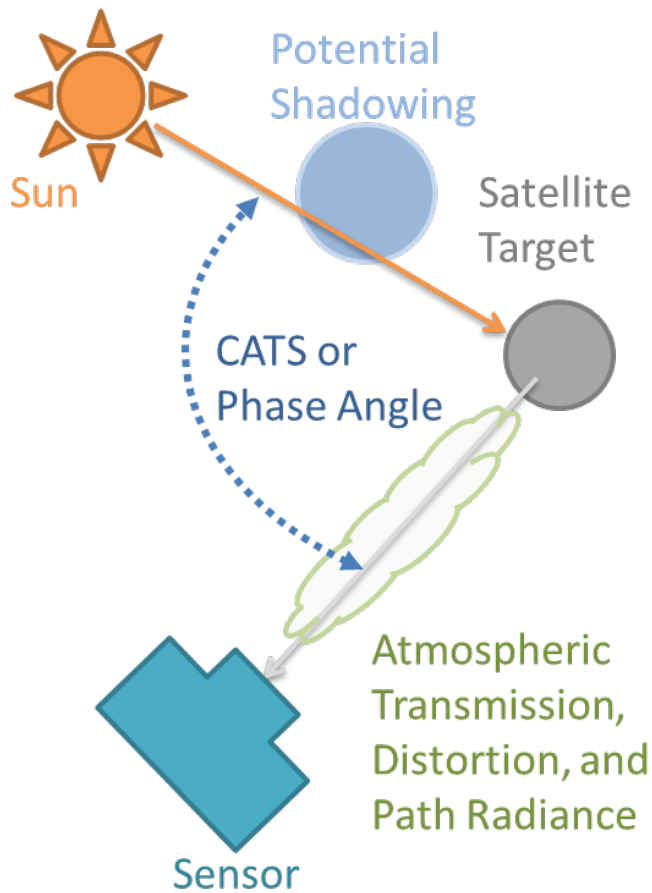
$$E_{Sun} = \frac{I_{Sun}}{d_{SunToTarget}^2} \text{ [W/m}^2\text{]}$$

$$Phase(\varphi) = \frac{\sin \varphi + (\pi - \varphi) \cos \varphi}{\pi} \text{ [ratio]}$$

$$\rho A_{Target} = \frac{\pi I_{Target}}{E_{Sun} F Phase(\varphi)} \text{ [m}^2\text{]}$$

From the above equations,  $VM_{absolute}$  “normalized” to a 1 AU Sun-to-target distance, a phase angle of 0° and a 40,000 km target-to-sensor distance (equivalent to a GEO satellite tracked at 15.6° elevation above the optical site’s local horizon), is obtained as:

$$VM_{absolute} = -2.5 \log_{10} \left\{ \frac{[E_{Sun1AU} = 1380 \text{ W/m}^2] [Phase(0 \text{ rad}) = 1.0] [\rho A_{Target} \text{ from above, in m}^2]}{\pi [E_0 = 2.77894 \times 10^{-8} \text{ W/m}^2] [1.6 \times 10^{15} \text{ m}^2]} \right\}$$



## ANNEX E

**ABBREVIATIONS AND ACRONYMS****(INFORMATIVE)**

ASCII	American Standard Code for Information Interchange
CCSDS	Consultative Committee for Space Data Systems
CIO	Celestial Intermediate Origin
CIP	Celestial Intermediate Pole
DSST	Draper Semi-Analytic Satellite Theory
ECI	Earth Centered Inertial
EGM	Earth Gravitational Model, Earth Geopotential Model
EME2000	Earth Mean Equator and Equinox of J2000 (Julian Date 2000)
EOP	Earth Orientation Parameters
GCRF	Geocentric Celestial Reference Frame
GPS	Global Positioning System
IAU	International Astronomical Union
ICD	Interface Control Document
ICRF	International Celestial Reference Frame
IEC	International Electrotechnical Commission
IERS	International Earth Rotation and Reference Systems Service
IIRV	Improved Inter-Range Vector
ISO	International Standards Organization
ITRF	International Terrestrial Reference Frame
ITRS	International Terrestrial Reference System
GRC	Greenwich Rotating Coordinate Frame
KVN	Keyword = Value Notation

NORAD	North American Aerospace Defense Command
OD	Orbit Determination
ODM	Orbit Data Message
OEB	Optimally-Encompassing Box
OEM	Orbit Ephemeris Message
OCM	Orbit Comprehensive Message
OMM	Orbit Mean-Elements Message
OPM	Orbit Parameter Message
RTN	Radial, Transverse (along-track) and Normal
S/C	Spacecraft
SGP4	US Air Force Simplified General Perturbations No. 4
SPK	Satellite, Planetary Kernel
TAI	International Atomic Time
TCB	Barycentric Coordinate Time
TCG	Geocentric Coordinate Time
TDB	Barycentric Dynamical Time
TDR	True of Date Rotating
TDT	Terrestrial Dynamical Time (see also ‘TT’)
TEME	True Equator Mean Equinox
TLE	Two Line Element
TOD	True Equator and Equinox of Date
TT	Terrestrial Dynamical Time (see also ‘TDT’)
USM	Universal Semianalytical Method
UTC	Coordinated Universal Time
W3C	World Wide Web Consortium
WGS	World Geodetic System

XML      Extensible Markup Language



## ANNEX F

**RATIONALE FOR ORBIT DATA MESSAGES****(INFORMATIVE)****F1 OVERVIEW**

This annex presents the rationale behind the design of each message. It may help the application engineer to select a suitable message.

A specification of requirements agreed to by all parties is essential to focus design and to ensure the product meets the needs of the Member Agencies and satellite operators. There are many ways of organizing requirements, but the categorization of requirements is not as important as the agreement to a sufficiently comprehensive set. In this section the requirements are organized into three categories:

- a) **Primary Requirements:** These are the most elementary and necessary requirements. They would exist no matter the context in which the CCSDS is operating, i.e., regardless of pre-existing conditions within the CCSDS, its Member Agencies, or other independent users.
- b) **Heritage Requirements:** These are additional requirements that derive from pre-existing Member Agency or other independent user requirements, conditions or needs. Ultimately these carry the same weight as the Primary Requirements. This Recommended Standard reflects heritage requirements pertaining to some of the CCSDS Areas' home institutions collected during the preparation of the document; it does not speculate on heritage requirements that could arise from other sources. Corrections and/or additions to these requirements are expected during future updates.
- c) **Desirable Characteristics:** These are not requirements, but they are felt to be important or useful features of the Recommended Standard.

## F2 PRIMARY REQUIREMENTS ACCEPTED BY THE ORBIT DATA MESSAGES

### F2.1 PRIMARY REQUIREMENTS

Requirement	OPM?	OMM?	OEM?	OCM?
Data must be provided in digital form (computer file).	Y	Y	Y	Y
The file specification must not require of the receiving exchange partner the separate application of, or modeling of, spacecraft dynamics or gravitational force models, or integration or propagation.	N	N	Y	Y
The interface must facilitate the receiver of the message to generate a six-component Cartesian state vector (position and velocity) at any required epoch.	Y	Y	Y	Y
State vector information must be provided in a reference frame that is clearly identified and unambiguous.	Y	Y	Y	Y
Identification of the object and the center(s) of motion must be clearly identified and unambiguous.	Y	Y	Y	Y
Time measurements (time stamps, or epochs) must be provided in a commonly used, clearly specified system.	Y	Y	Y	Y
The time bounds of the ephemeris must be unambiguously specified.	N/A	N/A	Y	Y
The Recommended Standard must provide for clear specification of units of measure.	Y	Y	Y	Y
Files must be readily ported between, and useable within, 'all' computational environments in use by Member Agencies.	Y	Y	Y	Y
Files must have means of being uniquely identified and clearly annotated. The file name alone is considered insufficient for this purpose.	Y	Y	Y	Y
File name syntax and length must not violate computer constraints for those computing environments in use by Member Agencies.	Y	Y	Y	Y
A means to convey information about the uncertainty of the state shall be provided.	Y	Y	Y	Y

**F2.2 HERITAGE REQUIREMENTS**

<b>Requirement</b>	<b>OPM?</b>	<b>OMM?</b>	<b>OEM?</b>	<b>OCM?</b>
Ephemeris data is reliably convertible into the SPICE SPK (NASA) format (reference [L6]) and IIRV (NASA) format (reference [L7]) using a standard, multi-mission, unsupervised pipeline process. A complete ephemeris, not subject to integration or propagation by the customer, must be provided.	N	N	Y	Y
Ephemeris data provided for Deep Space Network (DSN), Ground Network (GN), and Space Network (SN) scheduling or operations (metric predicts) is to be certified by the providing Agency as correct and complete for the intended purpose. The receiving Agency cannot provide evaluation, trajectory propagation or other usability services.	N	N	Y	Y
The ODM shall provide a mechanism by which messages may be uniquely identified and clearly annotated. Facilitates discussion between the recipient and the message originator, should that be necessary.	N	N	N	Y
The ODM shall provide a mechanism by which maneuvers may be uniquely identified and clearly annotated. Facilitates discussion between the recipient and the message originator, should that be necessary.	N	N	N	Y
The Recommended Standard is, or includes, an ASCII format.	Y	Y	Y	Y
The Recommended Standard does not require software supplied by other Agencies.	Y	N	Y	Y

**F2.3 DESIRABLE CHARACTERISTICS**

<b>Requirement</b>	<b>OPM?</b>	<b>OMM?</b>	<b>OEM?</b>	<b>OCM?</b>
The Recommended Standard applies to non-traditional objects, such as landers, rovers, balloons, and natural bodies (asteroids, comets).	Y	N	Y	Y
The Recommended Standard allows state vectors to be provided in other than the traditional EME2000 inertial reference frame; one example is the International Astronomical Union (IAU) Mars body-fixed frame. (In such a case, provision or ready availability of supplemental information needed to transform data into a standard frame must be arranged.)	Y	Y	Y	Y
The Recommended Standard is extensible with no disruption to existing users/uses.	Y	Y	Y	Y
The Recommended Standard is consistent with, and ideally a part of, ephemeris products and processes used for other space science purposes.	N	Y	N	Y
The Recommended Standard is as consistent as reasonable with any related CCSDS ephemeris Recommended Standards used for earth-to-spacecraft or spacecraft-to-spacecraft applications.	Y	Y	Y	Y

## F2.4 APPLICABILITY OF CRITERIA TO MESSAGE OPTIONS

The selection of one particular message will depend on the optimization criteria in the given application. Section F2.5 compares the three recommended messages in terms of the relevant selection criteria identified by the CCSDS:

## F2.5 APPLICABILITY OF THE CRITERIA TO ORBIT DATA MESSAGES

Criteria	Definition	Applicable to OPM?	Applicable to OMM?	Applicable to OEM?	Applicable to OCM?
Modeling Fidelity	Permits modeling of any dynamic perturbation to the trajectory.	N	N	Y	Y
Human Readability	Provides easily readable message corresponding to widely used orbit representation.	Y	Y	Y	Y
Remote Body Extensibility	Permits use for assets on remote solar system bodies.	Y	N	Y	Y
Lander/Rover Compatibility	Permits exchange of non-orbit trajectories.	N	N	Y	Y

## F3 INCREASING ORBIT PROPAGATION FIDELITY OF AN OPM OR OMM

Some OPM, OMM and/or OCM users may desire/require a higher fidelity propagation of the state vector or Keplerian elements. A higher fidelity technique may be desired/required to minimize inconsistencies in predictions generated by diverse, often operator-unique propagation schemes. Nominally the OPM, OMM and OCM are engineered only for low- to medium-fidelity orbit propagation. However, with the inclusion of additional context information, it is possible for users to provide data that could be used to provide a relatively higher fidelity orbit propagation. For this relatively higher fidelity orbit propagation, a much greater amount of ancillary information regarding spacecraft properties and dynamical models should be provided. Higher fidelity orbit propagations may be useful in special studies such as orbit conjunction studies.

Spacecraft orbit determination is a stochastic estimation problem; observations are inherently uncertain, and not all of the phenomena that influence satellite motion are clearly discernible. State vectors and Keplerian elements with their respective covariances are best propagated with models that include the same forces and phenomena that were used for determining the orbit. Including this information in an OPM/OMM allows exchange partners to compare the results of their respective orbit propagations.

With additional context information, the OPM/OMM/OCM may be used for assessing mutual physical or electromagnetic interference among Earth-orbiting spacecraft, developing collaborative maneuvers, and propagating the orbits of active satellites, inactive man-made objects, and near-Earth debris fragments. The additional information facilitates dynamic modeling of any user’s approach to conservative and non-conservative phenomena.

The primary vehicle for the provision of additional optional ancillary information to be used when propagating an OPM/OMM/OCM is the COMMENT mechanism. A number of potential COMMENT statements are included in annex G. Alternatively, the ‘USER\_DEFINED\_’ keyword prefix may be used, though this usage is not encouraged.

**F4 SERVICES RELATED TO THE DIFFERENT ORBIT DATA MESSAGE FORMATS**

The different orbit data messages have been distinguished by the self-interpretability of the messages. The different services that can be achieved without special arrangements between users of the CCSDS orbit data messages are listed in table **Error! Reference source not found.**

**F4.1 SERVICES AVAILABLE WITH ORBIT DATA MESSAGES**

Service	Definition	Applicable to OPM?	Applicable to OMM?	Applicable to OEM?	Applicable to OCM?
Absolute Orbit Interpretation	State availability at specific times for use in additional computations (geometry, event detection, etc.).	Y	Y	Y	Y
Relative Orbit Interpretation	Trajectory comparison and differencing for events based on the same time source.	Only at time specified at Epoch	Only at time specified at Epoch	Y	Y

## ANNEX G

**ITEMS FOR AN INTERFACE CONTROL DOCUMENT****(INFORMATIVE)****G1 STANDARD ICD ITEMS**

In several places in this document there are references to items which should be specified in an Interface Control Document (ICD) between participants that supplements an exchange of ephemeris data. The ICD should be jointly produced by both participants in a cross-support involving the transfer of ephemeris data. This annex compiles those recommendations into a single section. Although the Orbit Data Messages described in this document may at times be used in situations in which participants have not negotiated interface control documents (ICD), ICDs based on the content specified in this Recommended Standard should be developed and negotiated whenever possible.<sup>1</sup>

<b>Item</b>	<b>Section</b>
1) Definition of orbit accuracy requirements pertaining to any particular ODM.	1.2
2) Method of physically exchanging ODMs (transmission).	1.2, 3.1, 4.1, 5.1, 6.1
3) Whether the ASCII format of the ODM will be KVN or XML.	2.1
4) OPM, OMM, OEM and/or OCM file-naming conventions.	3.1, 4.1, 5.1, 6.1
5) Format on values used for the 'ORIGINATOR' keyword.	3.2.2, 4.2.2, 5.2.2, 6.2.2
6) Situations where the OBJECT_ID is not published in the UN OOSA index (reference [2]).	3.2.3, 4.2.3, 5.2.3, 6.2.3
7) Detailed description of any user defined parameters used.	3.2.4, 4.2.4, 6.2.10
8) Type of TEME reference frame, if applicable (TEME of Epoch or TEME of Date).	4.2.3
9) If floating point numbers in extended-single or extended-double precision are to be used, then discussion of implementation specific attributes is required in an ICD between exchange partners.	7.5
10) Information which must appear in comments for any given ODM exchange.	7.7

<sup>1</sup> EDITOR'S COMMENT: The greater the amount of material which must be specified via ICD, the lesser the utility/benefit of the ODM (custom programming may be required to tailor software for each ICD).

Item	Section
11) Specific OPM, OMM OEM and/or OCM version numbers that will be exchanged.	7.8.1
12) Specific information security interoperability provisions that apply between agencies.	0
13) Exceptions for the REF_FRAME and/or TIME_SYSTEM metadata keywords that are not drawn from annex A2.5.	annex A2.5
14) Interpretation of TIME_SYSTEM specified as MET, MRT or SCLK, if to be exchanged, and how to transform them to a standardized time system. The ICD should specify that elapsed days are to be used for epochs, with year starting at zero.	annex A2.5

## ANNEX H

**CHANGES IN ODM VERSION 2****(INFORMATIVE)**

This annex lists the differences between ODM 1.0 and ODM 2.0. The differences are divided into those which affect the content of one or more of the orbit data messages, and those which only affect the document.

**H1 CHANGES IN THE MESSAGES**

1. The Orbit Mean-Elements Message (OMM) was added to provide better support for ISO Technical Committee 20, Subcommittee 14 objectives (see section 4).
2. The 6x6 covariance matrix (lower triangular form) included in the initial version of the OMM was added to the OPM and OEM to allow producers of these files to provide the uncertainties associated with the state(s).
3. The option to use the Julian Date in formatting of epochs and other time fields is withdrawn, as this format is described in neither the CCSDS Time Code Formats (reference [1]) nor the ISO 8601 standard 'Data elements and interchange formats — Information interchange — Representation of dates and times'.
4. Optional accelerations were added to the state vectors provided in the OEM format (see section 5).
5. Some restrictions were imposed on the placement of COMMENT statements in order to allow easy conversion of ODMs from KVN format to XML format or vice versa.
6. The requirement to put the OBJECT\_ID parameter in UNOOSA index format was changed from a requirement ('shall') to a recommendation ('should') based on current operational uses of the OEM.
7. Maximum line width for all messages changed to 254 to be consistent with the Tracking Data Message (TDM) and Attitude Data Messages (ADM) Recommended Standards.
8. The rules for text value fields were constrained to only all uppercase or all lowercase.
9. The fields in the 'Spacecraft Parameters' block of the OPM were changed from mandatory to optional parameters.
10. The block of optional User Defined Parameters included in the initial version of the OMM is added to the OPM.



11. The REF\_FRAME\_EPOCH is added to accommodate cases when the reference frame epoch is not intrinsic to the definition of the reference frame.
12. The relationship between successive blocks of ephemeris data was clarified such that the repetition of time tags is relative to the USEABLE\_STOP\_TIME and USEABLE\_START\_TIME instead of the STOP\_TIME and START\_TIME.

## **H2 CHANGES IN THE DOCUMENT**

1. A normative annex for primary TIME\_SYSTEM and reference frame related keywords was added, replacing non-normative references to the Navigation Green Book (reference [L1]). The CCSDS documents are not allowed to make normative references to non-normative documents.
2. Annexes were rearranged to conform to CCSDS Guidelines that were inadvertently not followed in the first version of the ODM (specifically, normative annexes are supposed to appear first, prior to the informative annexes).
3. The formats of units allowed in the OPM were changed to make them compliant with the International System (SI) of Units. In the Blue Book version 1, the SI conventions were not observed. In all cases, this was merely a change in case conventions from upper case to lower case.
4. A few changes were made to harmonize the ODM with the other Navigation Data Messages (Attitude Data Messages [ADM] and Tracking Data Message [TDM]). Most of these changes were generated from the CCSDS Agency Review processes of the ADM and TDM.
5. In the original ODM Blue Book, several aspects of the CCSDS 'Style Guide' were not followed when the ODM was originally published. This version corrects these styling errors.
6. The annex that describes information to be included in an ICD was significantly revised to suggest additional information that would be worthwhile to exchange. Also, a checklist was added that will allow exchange partners to exchange ODMs when there is no time to negotiate a formal ICD by inserting COMMENT statements into an ODM.
7. The new Orbit Comprehensive Message (OCM) was added.
8. The syntax rules for the OPM, OMM, OEM and the new OCM were consolidated into a common syntax section (see section 0).
9. The rules for processing COMMENT keywords were consolidated into a single section of the document (see section 0).
10. Improved discussion of information security considerations was provided, per Secretariat request (see section 0).

## ANNEX I

**CREATING A VERSION 1.0 COMPLIANT OPM/OEM****(INFORMATIVE)**

This annex describes the actions that the user should perform in order to create an OPM or OEM that is compatible with implementations of the ODM version 1.0 Blue Book. Note: there is no version 1.0 compatible OMM; that message type first appears in the ODM version 2.0. Examples of version 1.0 compatible OPM and OEM messages are shown in the body of the document.

**I1 ODM VERSION 1.0 COMPATIBLE OPM**

For various reasons, the user may wish to create an OPM that is compatible with the ODM Version 1. In this case, the user must observe the following requirements:

- If the software implementation of the exchange partner will not accept a version 2.0 OPM, the value associated with the `CCSDS_OPM_VERS` keyword must be ‘1.0’.
- If the software implementation of the exchange partner will accept either version 1.0 or 2.0 OPMs, the value associated with the `CCSDS_OPM_VERS` keyword may be either ‘1.0’ or ‘2.0’.
- The OPM line length must not exceed 80 characters (including end of line markers).
- The user must *not* code the `REF_FRAME_EPOCH` metadata keyword.
- The user must code the Spacecraft Parameters logical block.
- The user must *not* code the Covariance Matrix logical block.
- The user must *not* code the User Defined Parameters logical block.
- If units are desired in the version 1.0 compatible OPM, it may be that an exchange partner’s software requires units in upper case characters as they were shown in the Version 1.0 OPM. While it seems unlikely that the case of the units in an OPM would be checked, the user should be aware of this potential issue.

NOTE – OPM Version 2.0 commenting rules are backward compatible to OPM Version 1.0.

**I2 ODM VERSION 1.0 COMPATIBLE OEM**

For various reasons, the user may wish to create an OEM that is compatible with the ODM Version 1. In this case, the user must observe the following requirements:

- If the software implementation of the exchange partner will not accept a version 2.0 OEM, the value associated with the `CCSDS_OEM_VERS` keyword must be ‘1.0’.

- If the software implementation of the exchange partner will accept either version 1.0 or 2.0 OEMs, the value associated with the CCSDS\_OEM\_VERS keyword may be either '1.0' or '2.0'.
- The user must **not** code the REF\_FRAME\_EPOCH metadata keyword.
- The user must **not** code the acceleration components in the OEM Data Lines.
- The user must **not** code the Covariance Matrix logical block.
- The user must ensure that ephemeris data time tags do not overlap except at the STOP\_TIME/START\_TIME boundary.

NOTE – OEM Version 2.0 commenting rules are backward compatible to OEM Version 1.0.

## ANNEX J

### SECURITY, SANA, AND PATENT CONSIDERATIONS

#### J1 SECURITY CONSIDERATIONS

##### J1.1 ANALYSIS OF SECURITY CONSIDERATIONS

This subsection presents the results of an analysis of security considerations applied to the technologies specified in this Recommended Standard.

##### J1.2 CONSEQUENCES OF NOT APPLYING SECURITY TO THE TECHNOLOGY

The consequences of not applying security to the systems and networks on which this Recommended Standard is implemented could include potential loss, corruption, and theft of data. Because these messages are used in pointing request and potential satellite and instrument pointing maneuvers, the consequences of not applying security to the systems and networks on which this Recommended Standard is implemented could include compromise or loss of the mission if malicious tampering of a particularly severe nature occurs.

##### J1.3 POTENTIAL THREATS AND ATTACK SCENARIOS

Potential threats or attack scenarios include, but are not limited to, (a) unauthorized access to the programs/processes that generate and interpret the messages, (b) unauthorized access to the messages during transmission between exchange partners and (c) modification of the messages between partners. Protection from unauthorized access during transmission is especially important if the mission utilizes open ground networks, such as the Internet, to provide ground-station connectivity for the exchange of data formatted in compliance with this Recommended Standard. It is strongly recommended that potential threats or attack scenarios applicable to the systems and networks on which this Recommended Standard is implemented be addressed by the management of those systems and networks.

##### J1.4 DATA PRIVACY

Privacy of data formatted in compliance with the specifications of this Recommended Standard should be assured by the systems and networks on which this Recommended Standard is implemented.

**J1.5 DATA INTEGRITY**

Integrity of data formatted in compliance with the specifications of this Recommended Standard should be assured by the systems and networks on which this Recommended Standard is implemented.

**J1.6 AUTHENTICATION OF COMMUNICATING ENTITIES**

Authentication of communicating entities involved in the transport of data which complies with the specifications of this Recommended Standard should be provided by the systems and networks on which this Recommended Standard is implemented.

**J1.7 DATA TRANSFER BETWEEN COMMUNICATING ENTITIES**

The transfer of data formatted in compliance with this Recommended Standard between communicating entities should be accomplished via secure mechanisms approved by the Information Technology Security functionaries of exchange participants.

**J1.8 CONTROL OF ACCESS TO RESOURCES**

Control of access to resources should be managed by the systems upon which originator formatting and recipient processing are performed.

**J1.9 AUDITING OF RESOURCE USAGE**

Auditing of resource usage should be handled by the management of systems and networks on which this Recommended Standard is implemented.

**J1.10 UNAUTHORIZED ACCESS**

Unauthorized access to the programs/processes that generate and interpret the messages should be prohibited in order to minimize potential threats and attack scenarios.

**J1.11 DATA SECURITY IMPLEMENTATION SPECIFICS**

Specific information-security interoperability provisions that may apply between agencies and other independent users involved in an exchange of data formatted in compliance with this Recommended Standard could be specified in an ICD.

**J2 SANA CONSIDERATIONS**

The following ODM-related items will be registered with the SANA Operator.

- The ODM XML templates (see reference **Error! Reference source not found.**).

The following ODM elements should be from the SANA registry:

- the spacecraft names that appear as origin and target in the ODM (see reference **Error! Reference source not found.**);
- the ODM originators (see reference **Error! Reference source not found.**).

- the ODM originators (see reference **Error! Reference source not found.**).

The use of reference **Error! Reference source not found.** is a convenient solution of the identification of celestial bodies in absence of a corresponding SANA reference. For spacecraft the common identifiers in the SANA registry shall be preferred.

The registration rule for new entries in the registry is the approval of new requests by the CCSDS Area or Working Group responsible for the maintenance of the ODM at the time of the request. New requests for this registry should be sent to SANA (<mailto:info@sanaregistry.org>).

## **J2.1 PATENT CONSIDERATIONS**

The recommendations of this document have no patent issues.

## ANNEX K

## EPHEMERIS COMPRESSION (EC) TECHNIQUES (INFORMATIVE)

Ephemeris Compression (EC) techniques are a “lossy” approach to conveying orbit state time history(ies). EC techniques can dramatically reduce message transmission network bandwidth, CPU, I/O and disk storage requirements (e.g. by three orders of magnitude or more) while still retaining sufficient accuracy for many applications. Initial implementations focused on fitting Cartesian position and velocity [L13] with Chebyshev polynomials. Though successfully used in that manner operationally for many decades, recent research [L14] into application of such a compression technique indicates that best EC performance is obtained by: (1) employing “Hybrid” ephemeris compression by adopting an orbit-based element set definition (such as equinoctial elements) rather than a Cartesian representation; (2) adopting an orbit state and an accompanying sharable and efficient orbit propagator to use as the “basis” for the orbit representation; (3) using either exclusive or combined use of Chebyshev or Fourier representations to best “imitate” the residuals between the “truth” and efficient orbit propagator orbit state time histories captured in any of the reference frame definitions contained in ANNEX B, subsection B2 and orbit element set definitions contained in ANNEX B, subsections B4 or B5; and (4) using one or more consecutively staged ephemeris compression sequences to sequentially absorb the residuals between the adopted/sharable propagator and the precise ephemeris at each stage.

**7.8.2.5** In the OCM implementation of ephemeris compression, each Chebyshev or Fourier representation’s independent time variable shall be “normalized” to a time interval of  $-1 \leq t^* \leq +1$  via the following formula [L15], where  $a$  denotes the actual start time (i.e. EC\_TSTART) of the ephemeris compression representation segment’s time interval of validity,  $b$  denotes the corresponding actual segment stop time (i.e. EC\_TSTOP),  $t$  denotes the actual time of interest measured with respect to EPOCH\_TZERO and  $t^*$  denotes “normalized time”:

$$t^* = \frac{2 \cdot t - (a + b)}{(b - a)}$$

**7.8.2.6** EC users are encouraged to employ a blending function between subsequent EC data segments to ensure a smooth positional transition. Defining  $F_1$  as the desired orbit element functional value at actual time  $t$  obtained from the ephemeris compression representation of segment #1 and  $F_2$  from segment #2, and  $t_1$  and  $t_2$  as the overlap blending actual (non-normalized) start and stop times, respectively,

$$F(t) = \frac{1}{2} F_1(t) \left\{ 1 + \cos \left[ \left( \frac{t-t_1}{t_2-t_1} \right) \pi \right] \right\} + \frac{1}{2} F_2(t) \left\{ 1 + \cos \left[ \left( \frac{t_2-t}{t_2-t_1} \right) \pi \right] \right\}, t_1 \leq t \leq t_2$$

Centering the overlap interval on each segment boundary, retaining the definition of “ $b$ ” (from above) as the actual segment stop time (i.e. EC\_TSTOP) and further defining  $\Delta = t_2 - t_1$ , the blending function becomes:

$$F(t) = \frac{1}{2}F_1(t) \left\{ 1 - \sin \left[ \left( \frac{t-b}{\Delta} \right) \pi \right] \right\} + \frac{1}{2}F_2(t) \left\{ 1 - \sin \left[ \left( \frac{b-t}{\Delta} \right) \pi \right] \right\}, \quad b - \frac{\Delta}{2} \leq t \leq b + \frac{\Delta}{2}$$



## ANNEX L

## INFORMATIVE REFERENCES

## (INFORMATIVE)

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